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GRAPHS OF OBLIQUE-SHOCK RELATIONSHIPS FOR VARIOUS RATIOS OF SPECIFIC HEATS

W. C. Armstrong and J. C. Latimer ARO, Inc.

July 1969

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FOREWORD

The work reported herein was sponsored by Arnold Engineering Development Center (AEDC), Air Force Systems Command (AFSC), under Program Element 65401F.

The results presented were obtained by ARO, Inc. (a subsidiary of Sverdrup & Parcel and Associates, Inc.), contract operator of AEDC, AFSC, Arnold Air Force Station, Tennessee, under Contract F40600-69-C-0001. The work was performed under ARO Project No. BD5924, and the manuscript was submitted for publication on April 18, 1969.

This technical report has been reviewed and is approved.

Forrest B. Smith, Jr. Research Division Directorate of Plans and Technology

Harry L. Maynard Colonel, USAF Director of Plans and Technology

ABSTRACT

This report is a presentation of graphs of the oblique-shock relation-ships for several ratios of specific heats at several Mach numbers. The specific heat ratios vary from 1.10 to 1.66 in increments of 0.02, and a plot is also shown for a specific heat ratio equal to 1.667. The Mach numbers range from 1.5 to 6.0 with increments of 0.5 and from 6.0 to 10.0 with increments of 1.0; Mach number 15.0 is presented separately.

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NOMENCLATURE														
M	Mach number													
P	P Pressure													
q Dynamic pressure (ρV^2)														
T Temperature														
V Velocity														
γ Ratio of specific heats														
δ Flow deflection angle (wedge angle)														
θ	Shock angle													
ρ	Density													
SUBSCRIPTS														
1	Upstream of shock													
2	Downstream of shock													

Stagnation condition

t

SECTION I

Oblique shocks occur in supersonic flow when the flow encounters a compression corner (Fig. 1, Appendix). Often quick qualitative solutions to the oblique-shock equations are needed. These have been available for some time at one ratio of specific heat ($\gamma = 1.4$) in NACA Report 1135 in graphical form. The graphs in Report 1135 are applicable to air, nitrogen, and other diatomic gases.

The need also exists for similar graphs applicable to nondiatomic gases, for example, argon and engine exhaust gases. The specific heat ratios vary from less than 1.12 for exhaust gases to 1.67 for argon. The purpose of this report is to present graphs to cover specific heat ratios from 1.1 to 1.67 and Mach numbers from 1.5 to 15.

SECTION II METHOD OF SOLUTION

To obtain solutions which do not depend upon stagnation conditions, a perfect gas has been assumed. All the pertinent equations were obtained from NACA Report 1135.

For a given Mach number and ratio of specific heat, the maximum flow deflection angle for an attached shock is obtained from two equations. The first computes the shock angle, and the second the flow deflection angle.

$$\sin^2 \theta \delta_{\text{max}} = \frac{1}{4y M_1^2} \left\{ (y+1) M_1^2 - 4 + \sqrt{(y+1) [(y+1) M_1^4 + 8(y-1) M_1^2 + 16]} \right\}$$

$$\tan \delta_{\text{max}} = \frac{2 \cot \theta \delta_{\text{max}} (M_1^2 \sin^2 \theta \delta_{\text{max}} - 1)}{2 + M_1^2 (y+1-2\sin^2 \theta \delta_{\text{max}})}$$

This establishes the upper limit of the flow deflection. For flow deflection angles less than δ_{\max} , the shock angle is obtained by extracting the middle root of the cubic equation in $\sin^2\theta$:

$$\sin^{6}\theta - \left[\frac{M_{1}^{2} + 2}{M_{1}^{2}} - \gamma \sin^{2}\delta\right] \sin^{4}\theta + \left[\frac{2M_{1}^{2} + 1}{M_{1}^{4}} + \left\{\left(\frac{\gamma + 1}{2}\right)^{2} + \frac{\gamma - 1}{M_{1}^{2}}\right\} \sin^{2}\theta\right] \sin^{2}\theta - \frac{\cos^{2}\delta}{M_{1}^{4}} = 0$$

¹Ames Research Staff. "Equations, Tables, and Charts for Compressible Flow." NACA Report 1135, 1953.

The least root corresponds to a decrease in entropy which violates the second law of thermodynamics, and the greatest root corresponds to a strong shock which is not normally obtained.

Once the shock angle is known, the other plotted parameters are obtained from the following equations.

$$\begin{split} \frac{P_2}{P_1} &= \frac{2y \ M_1^2 \sin^2\theta - (y-1)}{y+1} \\ \frac{\rho_2}{\rho_1} &= \frac{(y+1) \ M_1^2 \sin^2\theta}{(y-1) \ M_1^2 \sin^2\theta + 2} \\ \frac{T_2}{T_1} &= \frac{\left[2y \ M_1^2 \sin^2\theta - (y-1)\right] \left[(y-1) \ M_1^2 \sin^2\theta + 2\right]}{(y+1)^2 \ M_1^2 \ Sin^2\theta} \\ M_2^2 &= \frac{(y+1)^2 \ M_1^4 \sin^2\theta - 4(M_1^2 \sin^2\theta - 1) \ (yM_1^2 \sin^2\theta + 1)}{\left[2y \ M_1^2 \sin^2\theta - (y-1)\right] \left[(y-1) \ M_1^2 \sin^2\theta + 2\right]} \\ P_{t_2}/P_{t_1} &= \frac{(y+1) \ M_1^2 \sin^2\theta}{\left[(y-1) \ M_1^2 \sin^2\theta + 2\right]} \frac{y}{y-1} \left[\frac{y+1}{2y \ M_1^2 \sin^2\theta - (y-1)}\right]^{\frac{1}{y-1}} \\ \frac{P_2 - P_1}{q} &= \frac{4 \ (M_1^2 \sin^2\theta - 1)}{(y+1)M_1^2} \end{split}$$

SECTION III RESULTS AND DISCUSSION

The results are presented as graphs (Figs. 2 through 31) in the following manner: The abscissa for all plots is the wedge angle (δ). There are seven graphs for each ratio of specific heat (γ). The ordinates are shock angle (θ). Mach number behind the shock (M2), pressure ratio across the shock (P2/P1), density ratio across shock (ρ 2/ ρ 1), temperature ratio across shock (T2/T1), total pressure ratio across shock (P $_{t2}$ /P $_{t1}$), and coefficient (Δ P/q).

Each graph contains curves for the following Mach numbers: 1.5, 2.0, 2.5, 3.0, 3.5, 4.0, 4.5, 5.0, 5.5, 6.0, 7.0, 8.0, 9.0, 10.0, and 15.0. The ratios of specific heat presented start at 1.10 and go to 1.66 in 0.02 increments. The last plot is for a specific heat ratio of 1.667, corresponding to a monatomic gas.

APPENDIX ILLUSTRATIONS

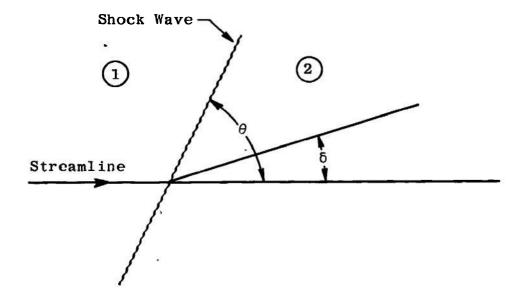


Fig. 1 Notation for Oblique Shock Wave

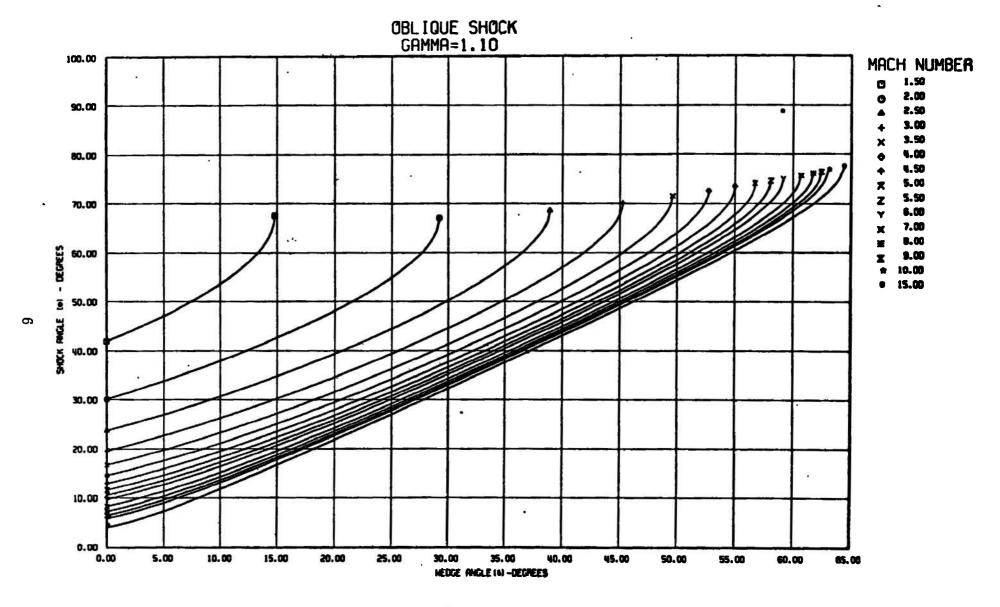


Fig. 2 y = 1.10

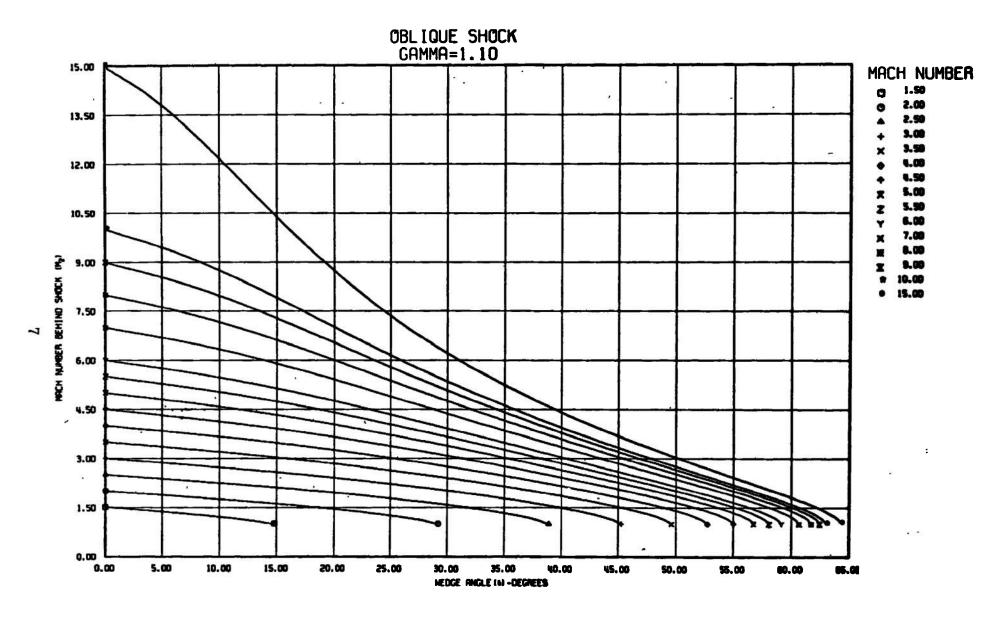


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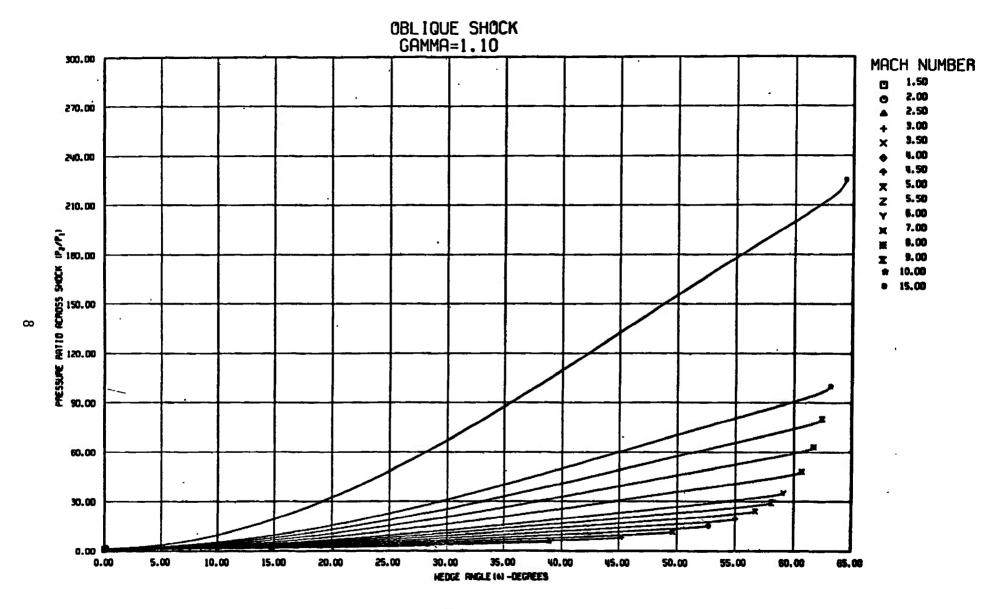


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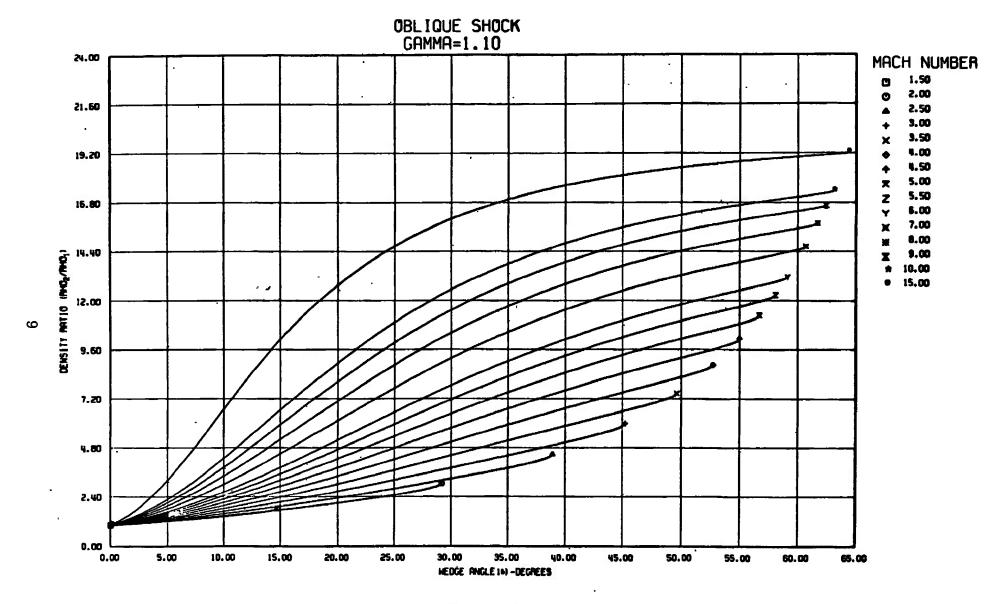


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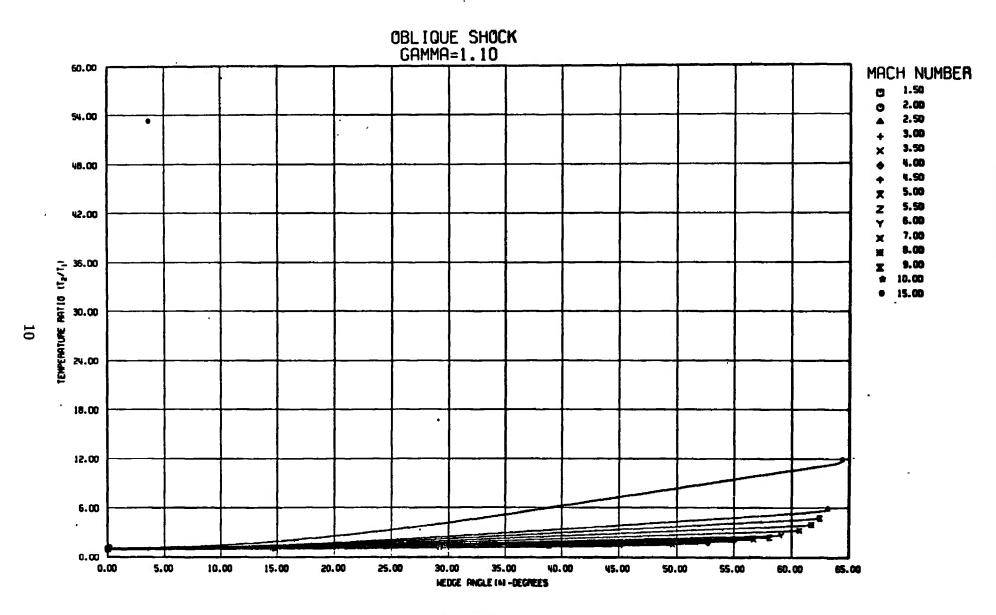


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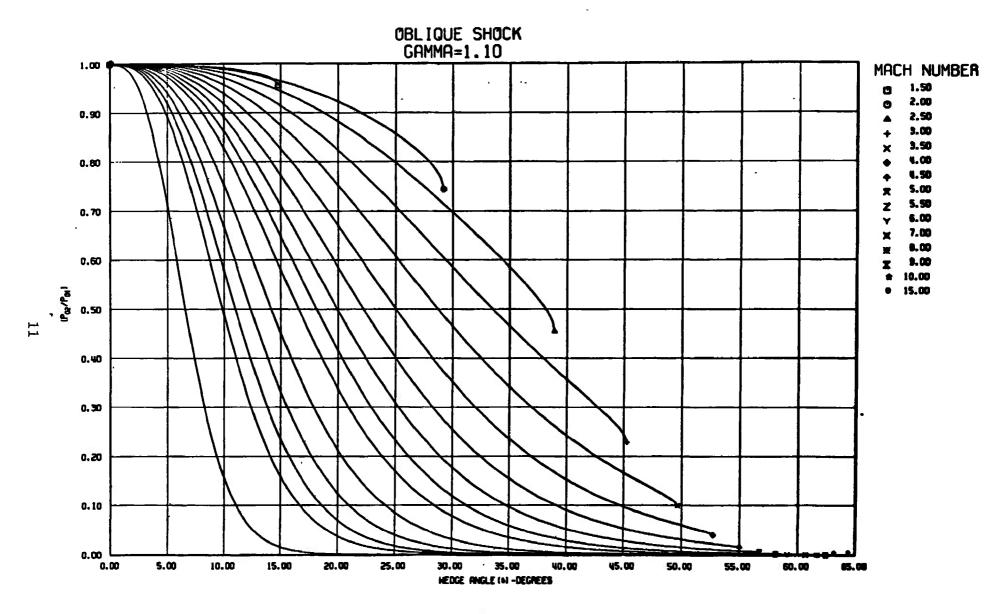


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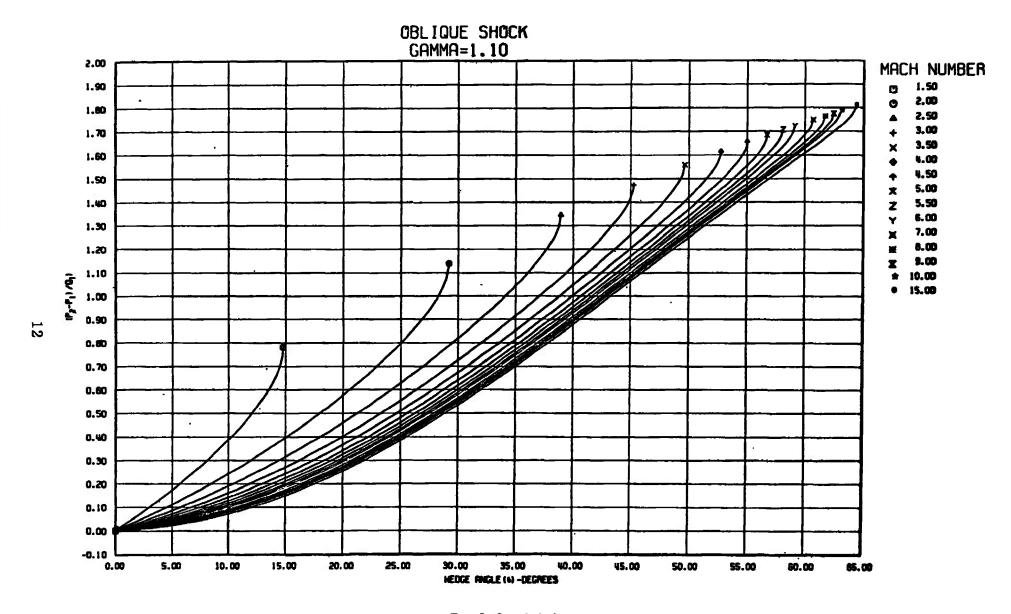


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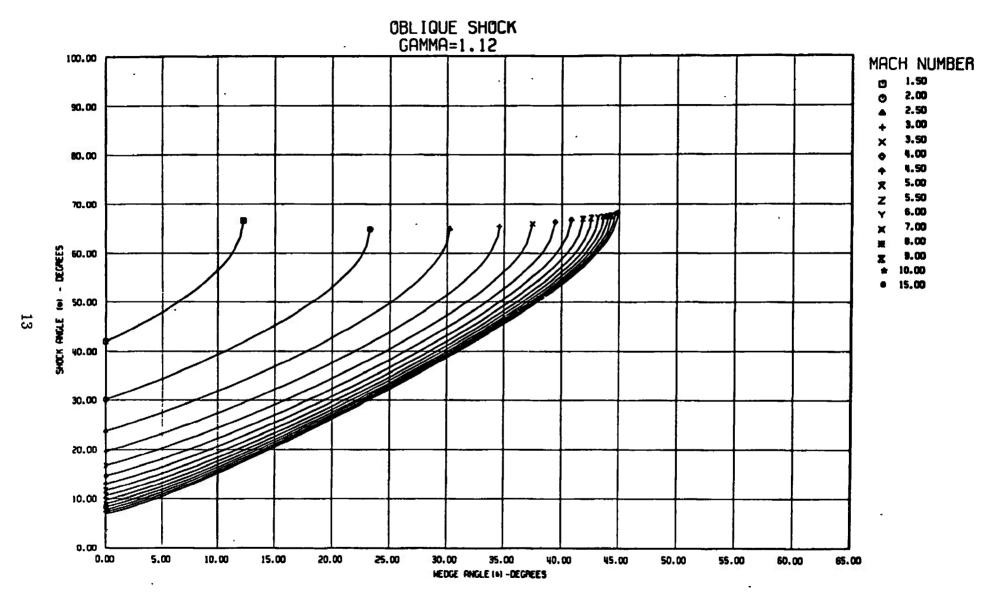


Fig. 3 y = 1.12

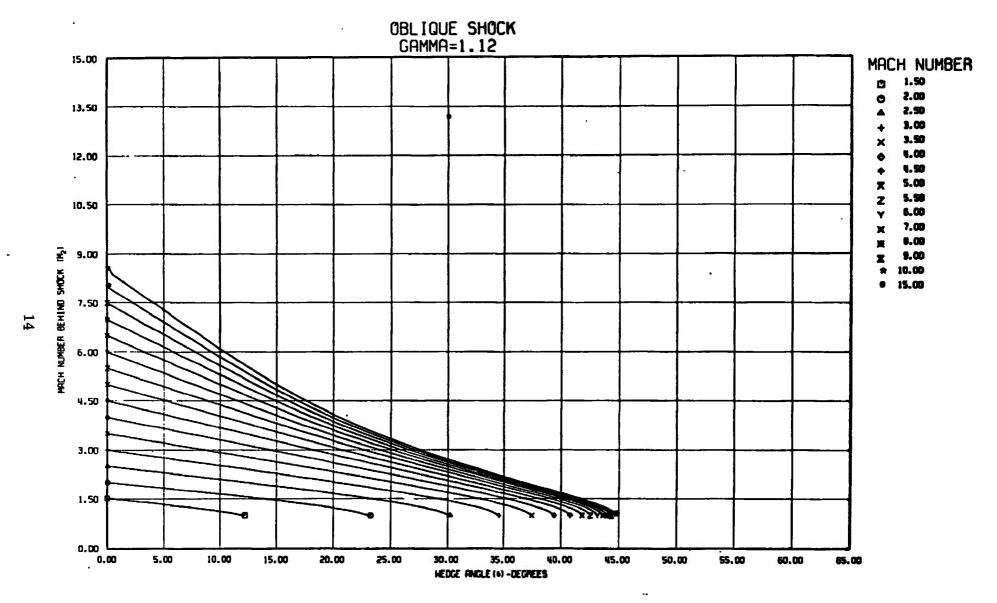


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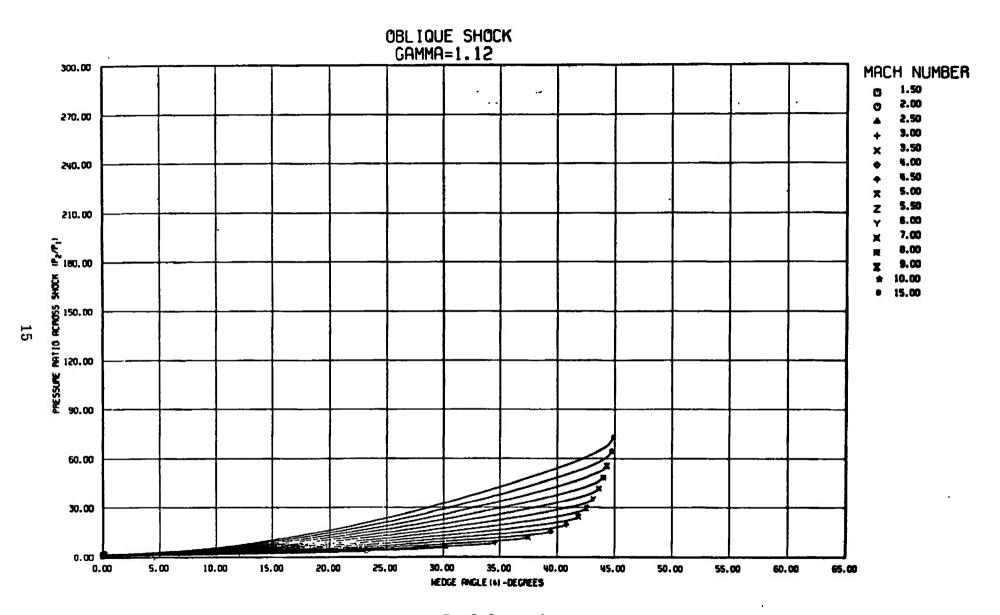


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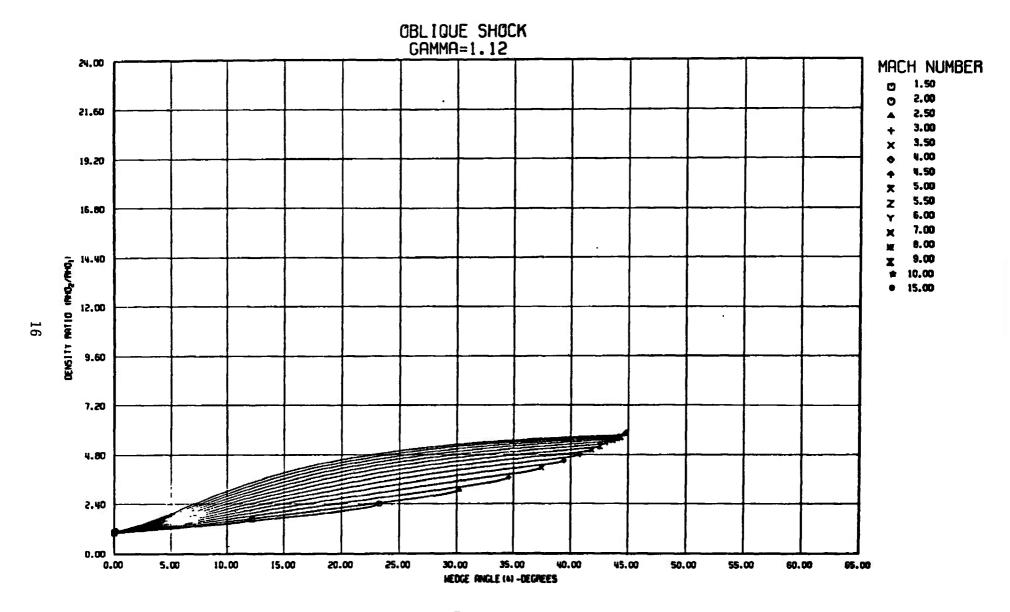


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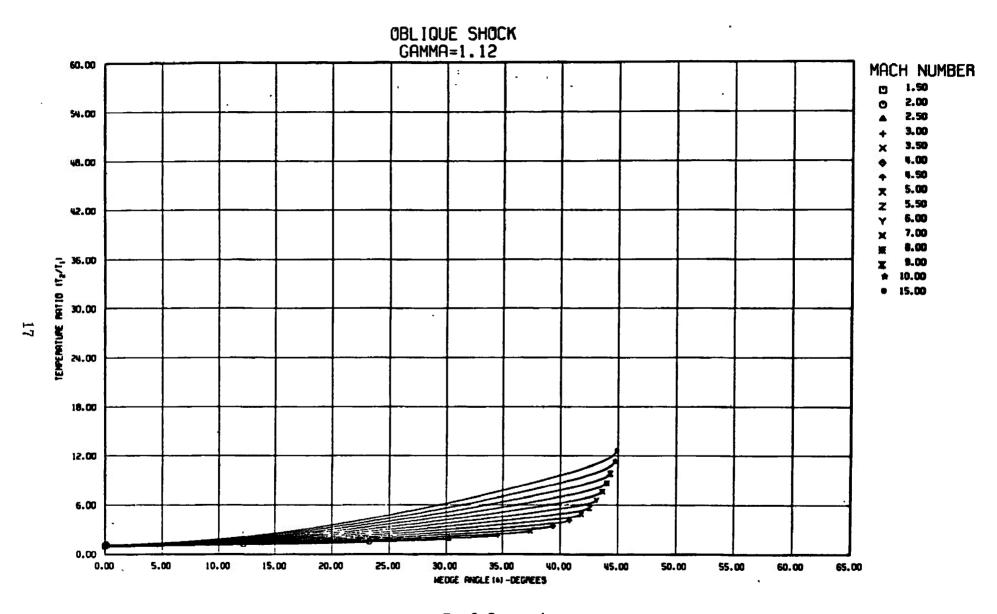


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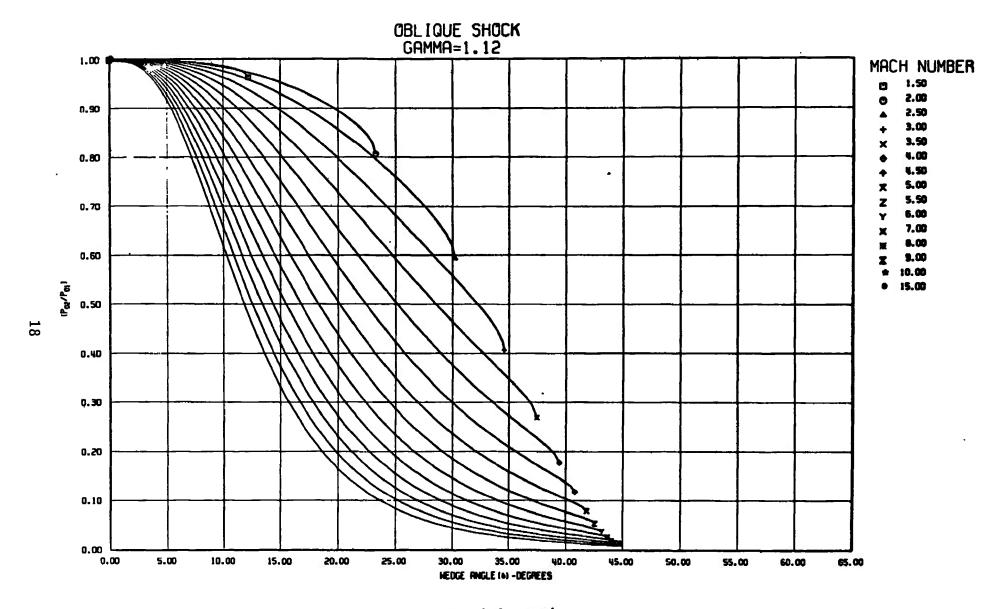


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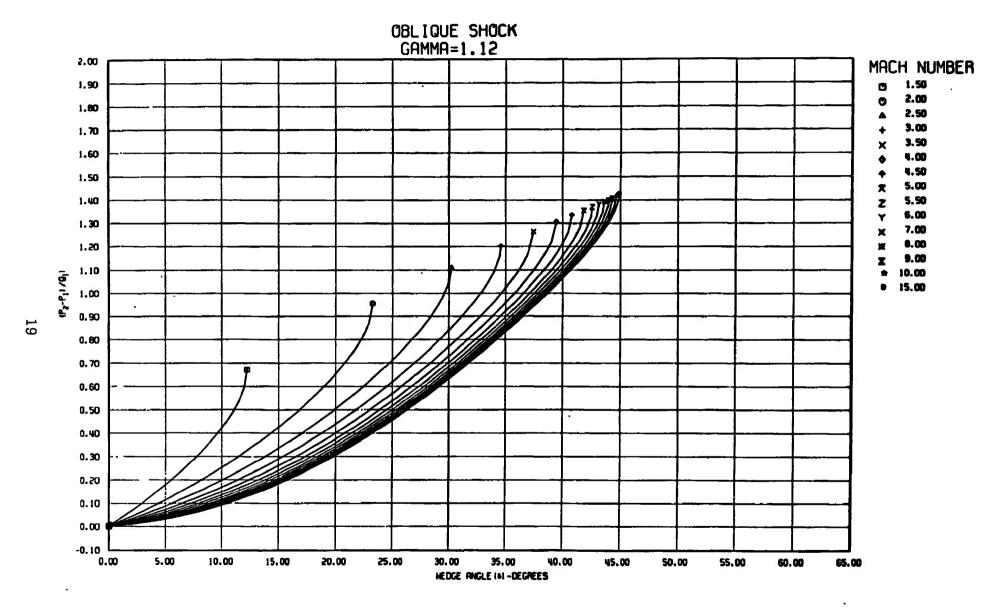


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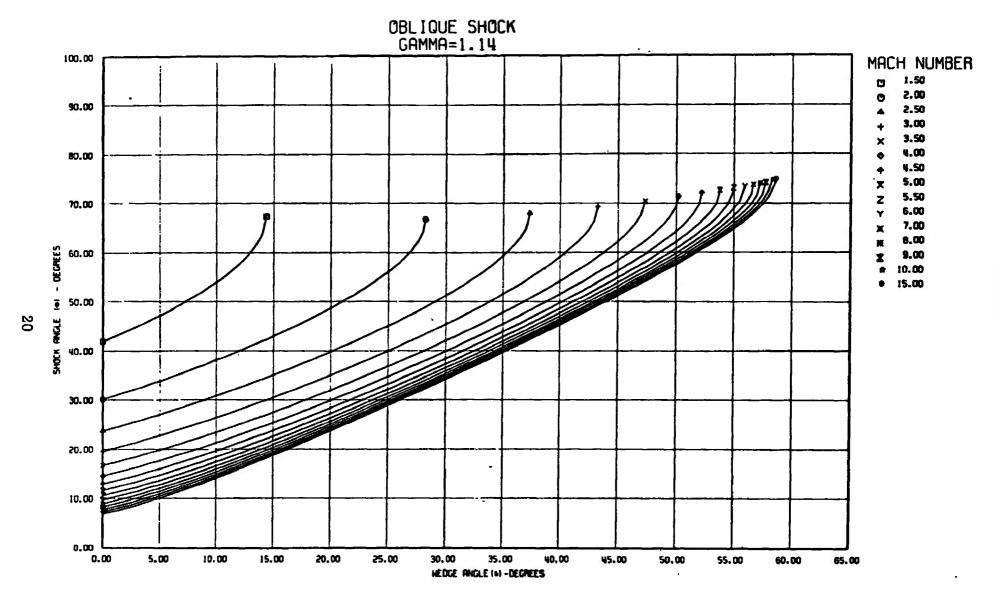


Fig. 4 y = 1.14

Fig. 4 Continued

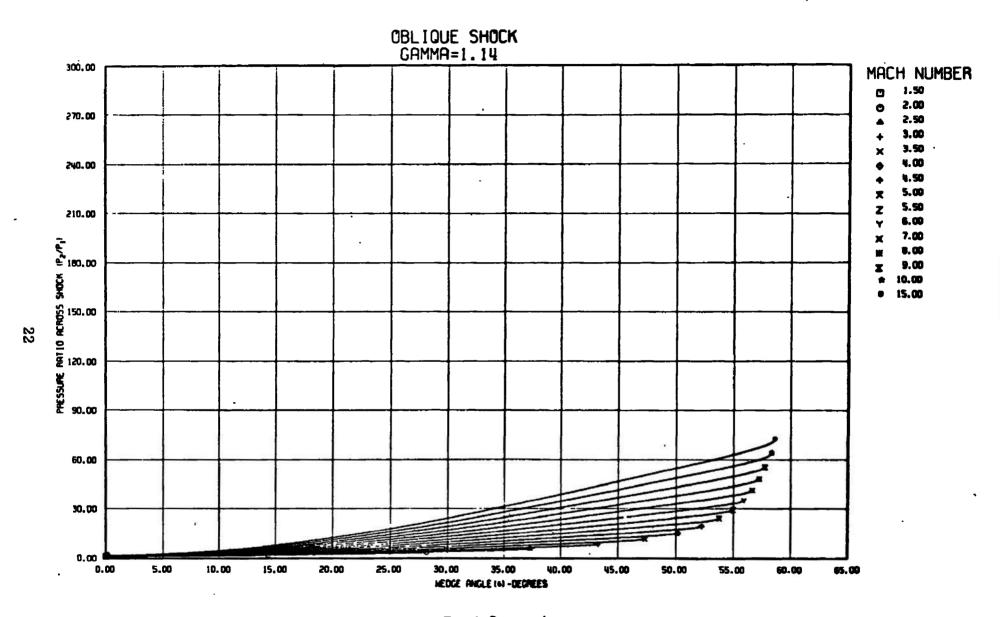


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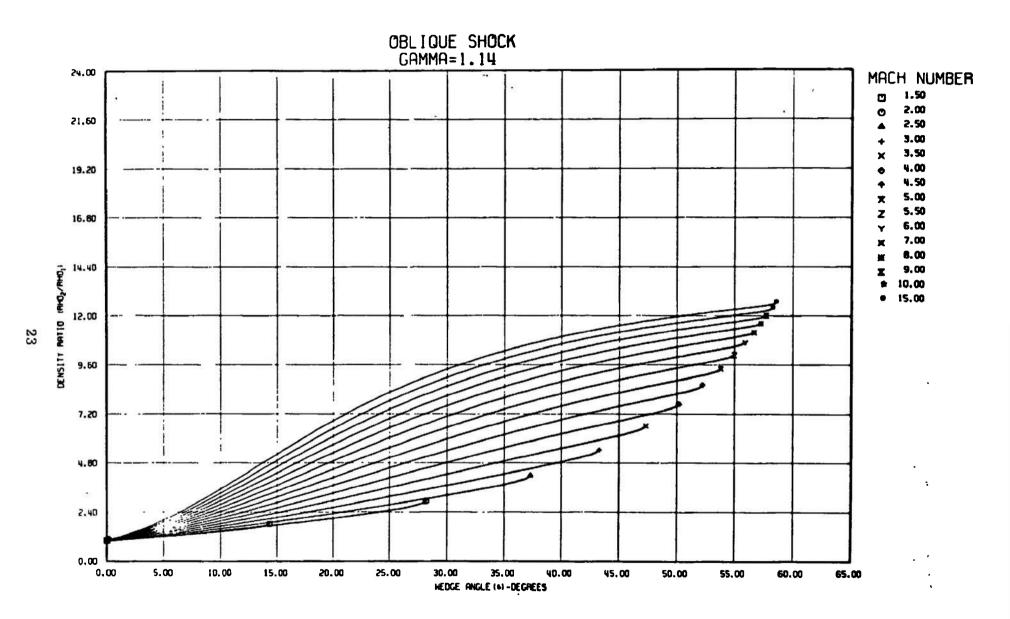


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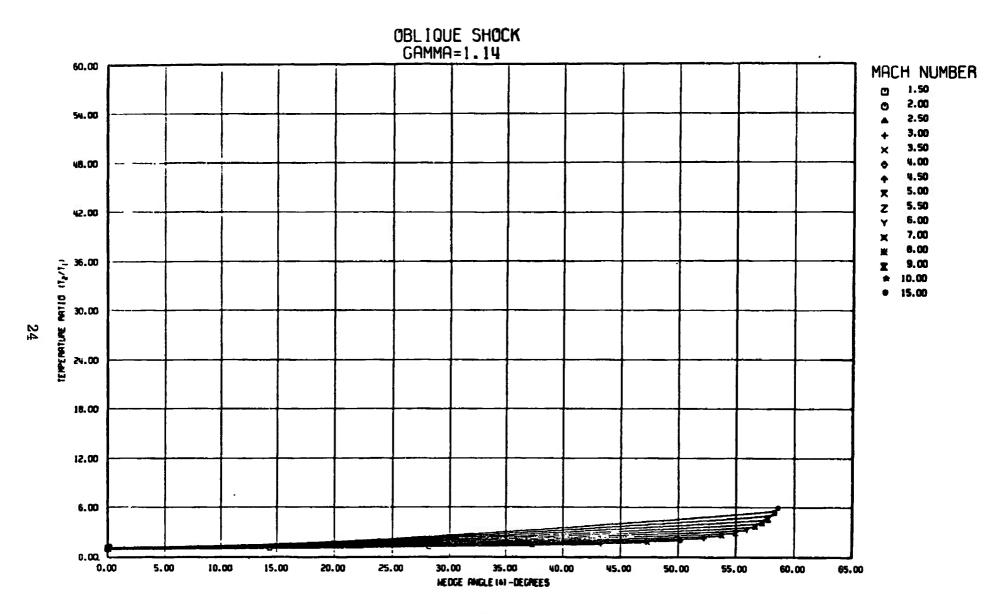


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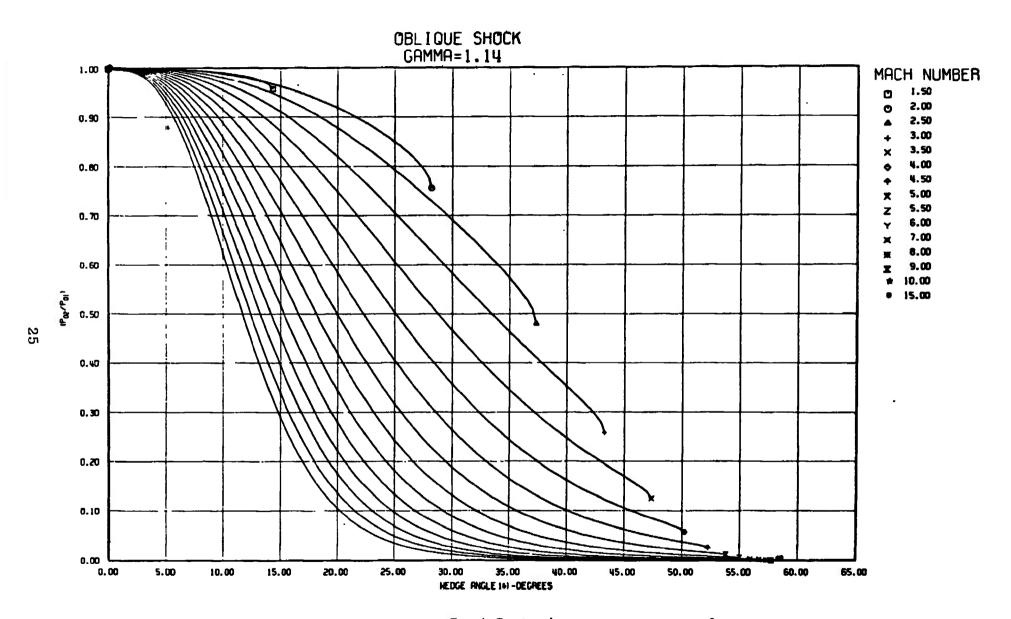


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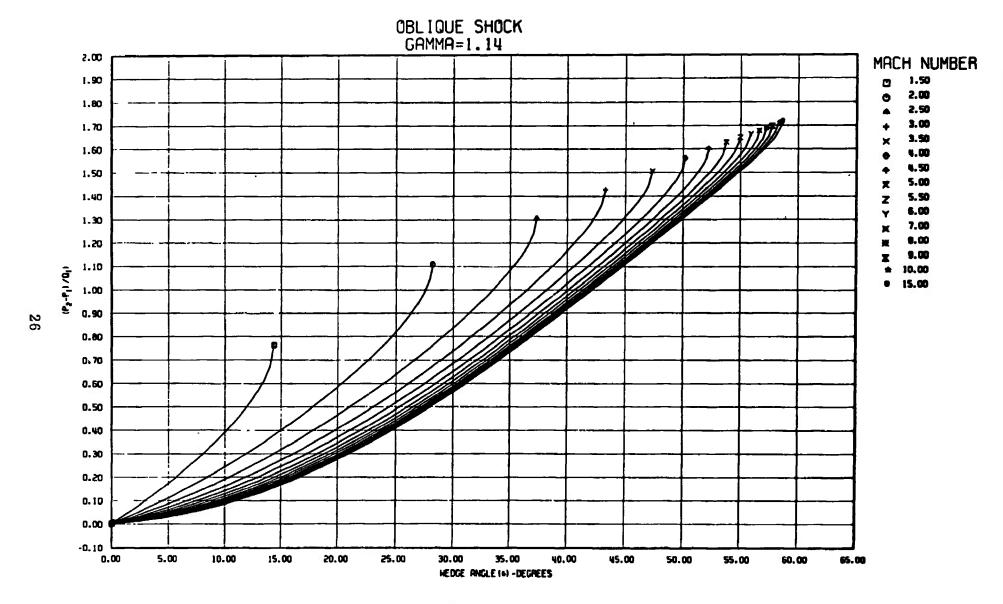


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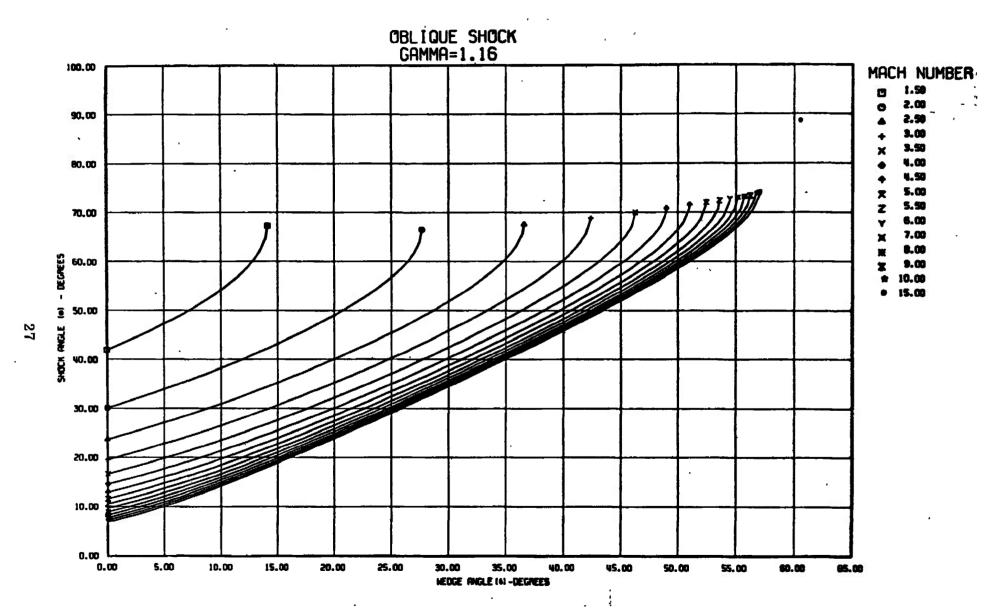


Fig. 5 y = 1.16

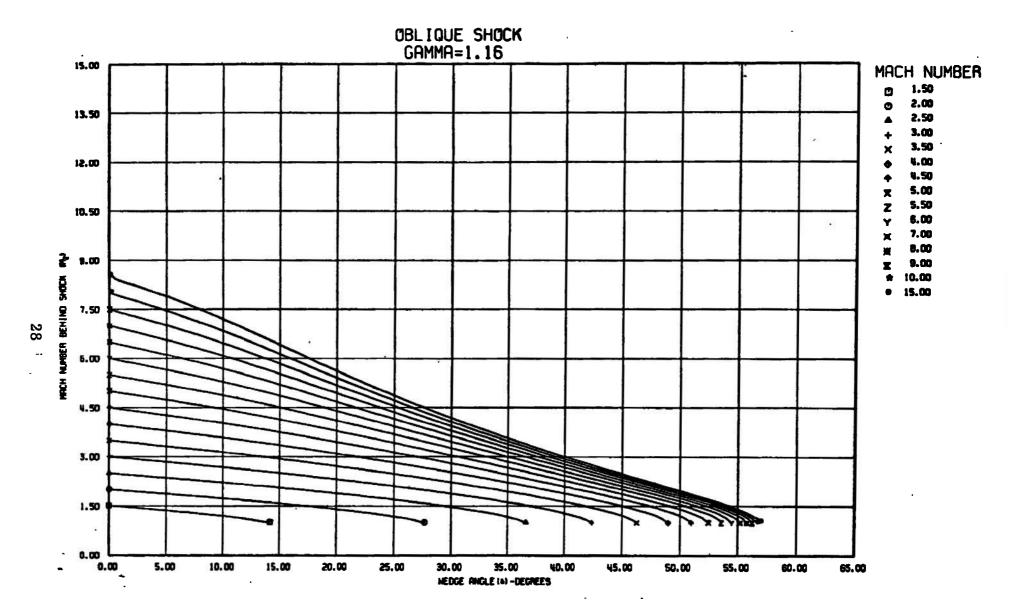


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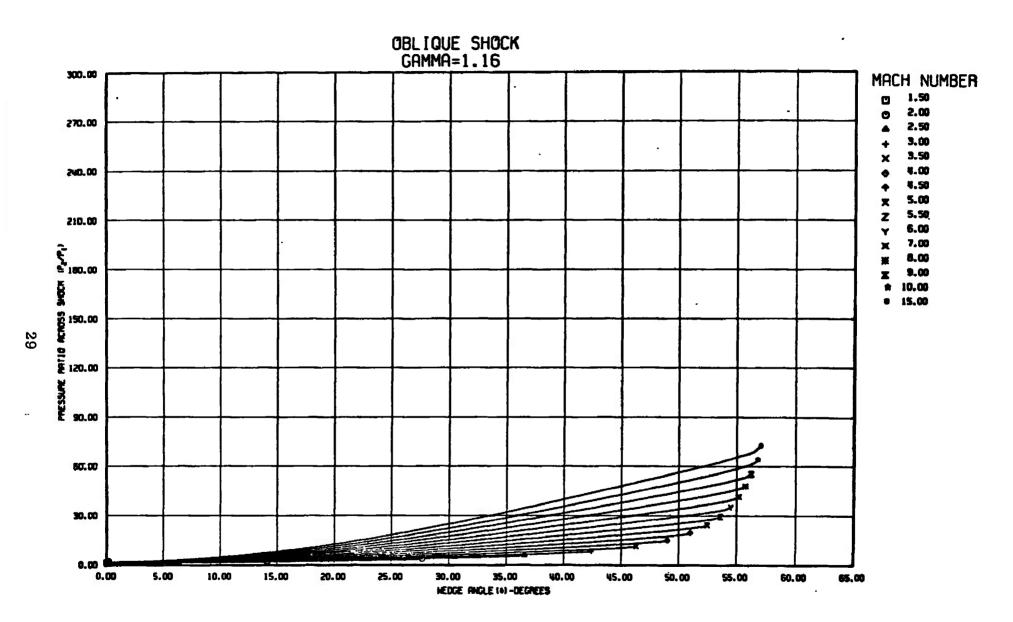


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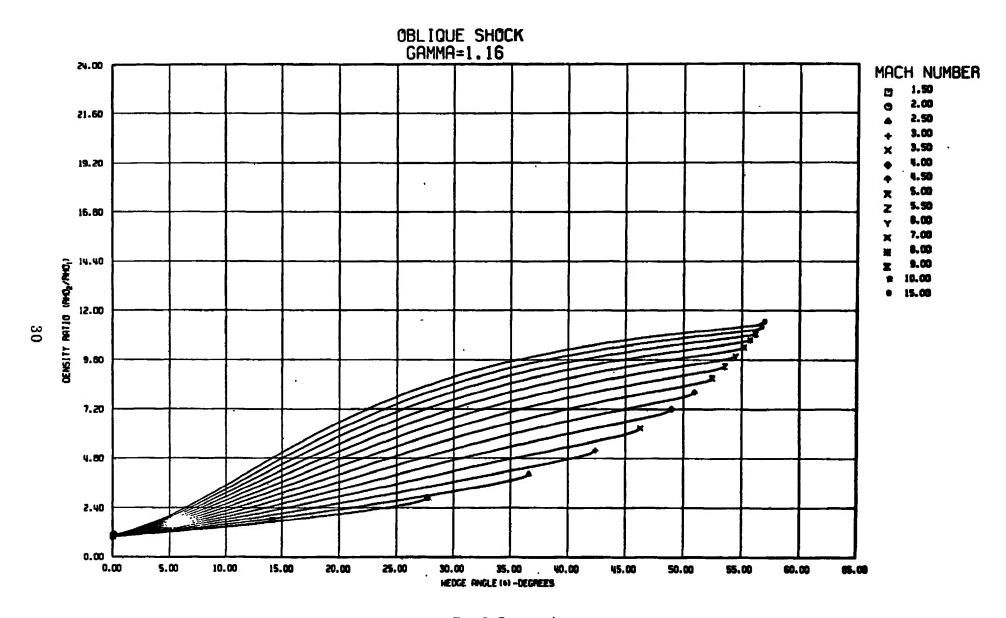


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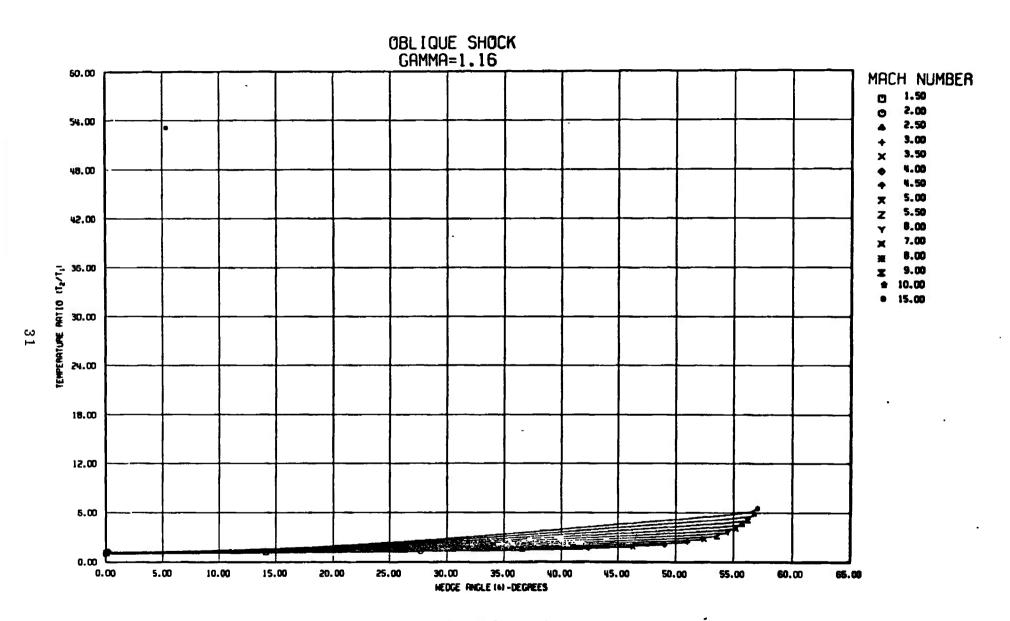


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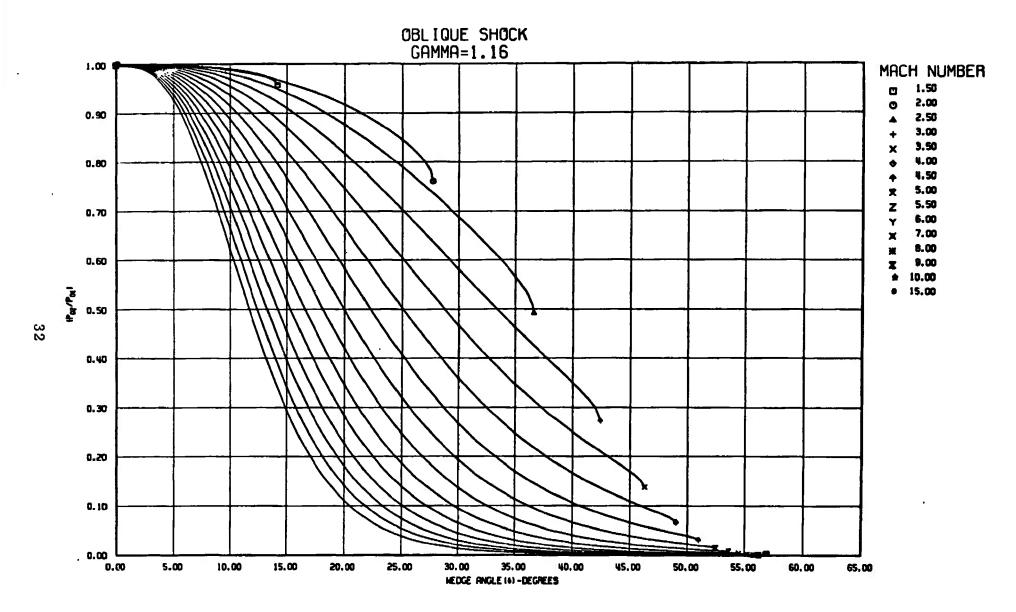


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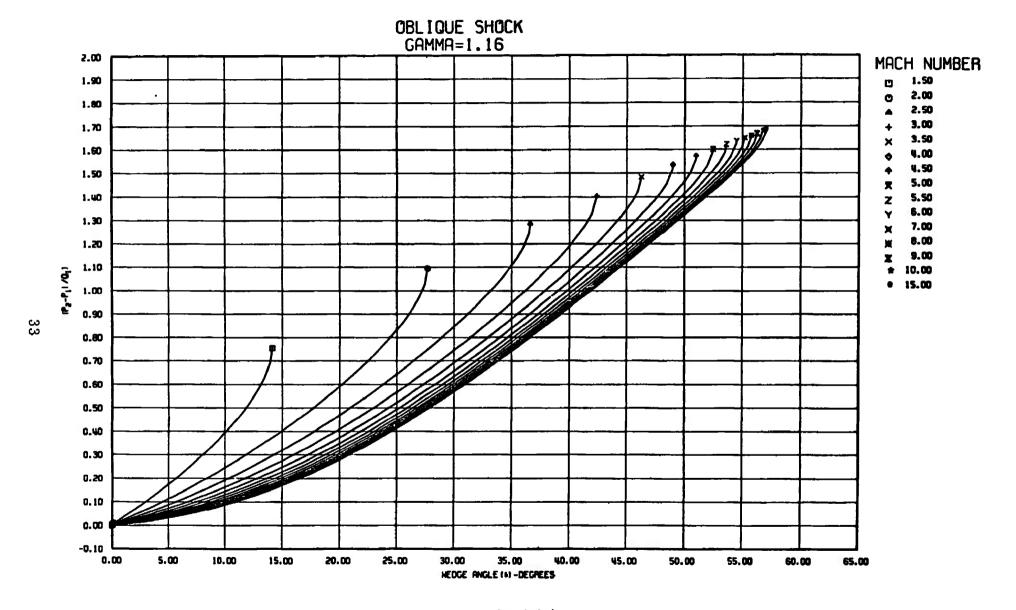


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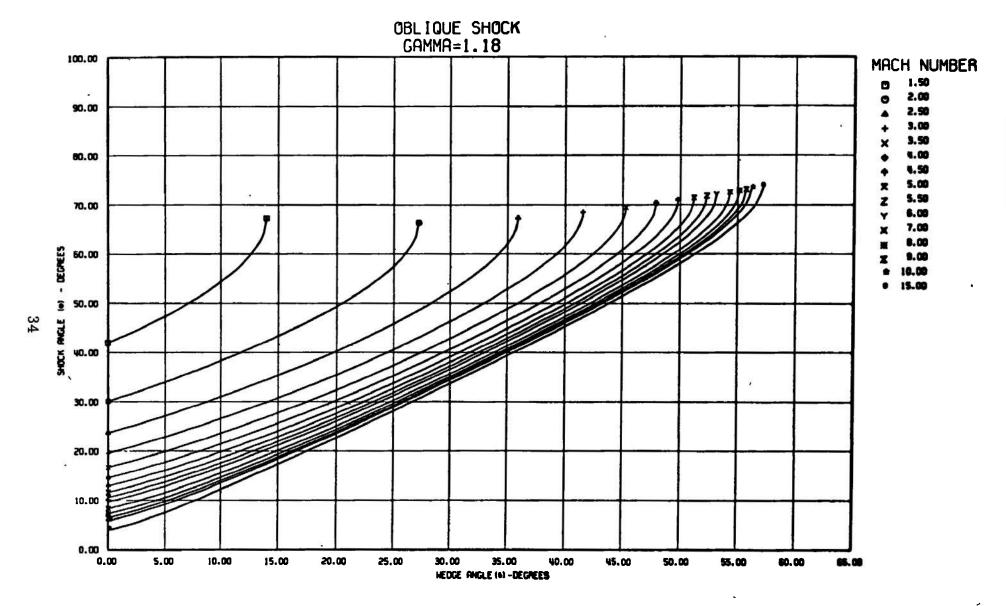


Fig. 6 $\gamma = 1.18$

Fig. 6 Continued

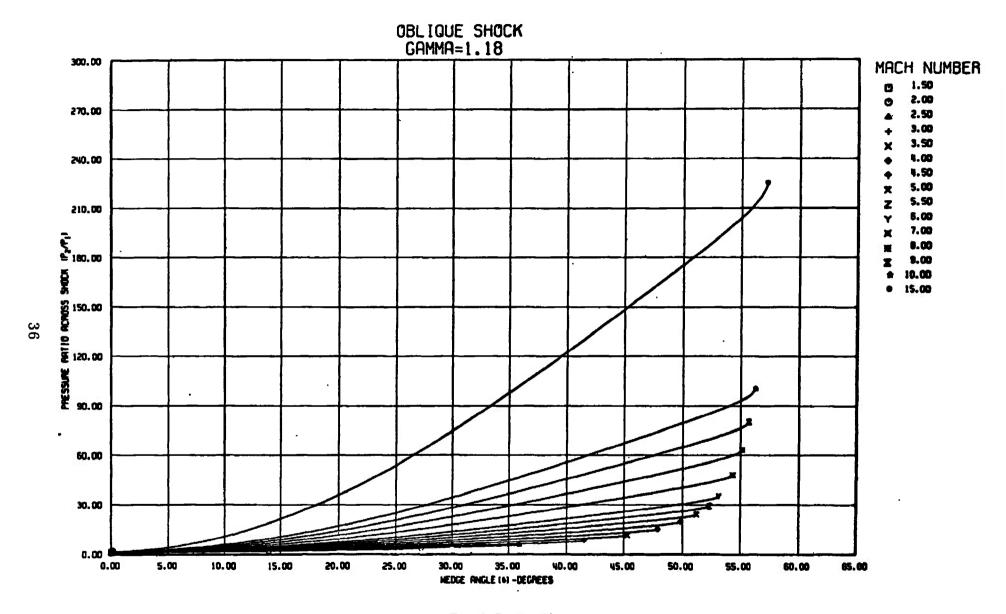


Fig. 6 Continued

Fig. 6 Continued

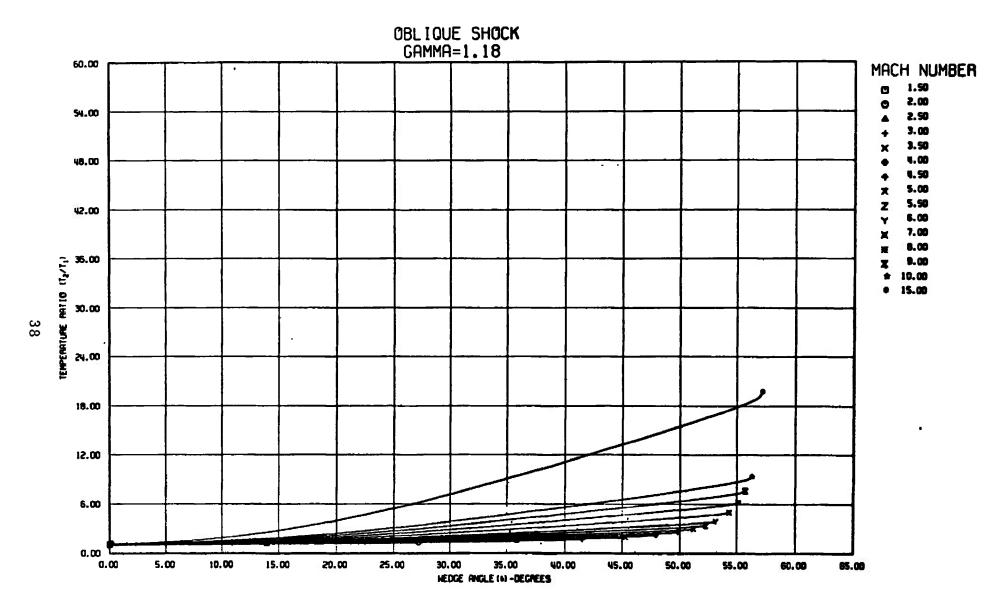


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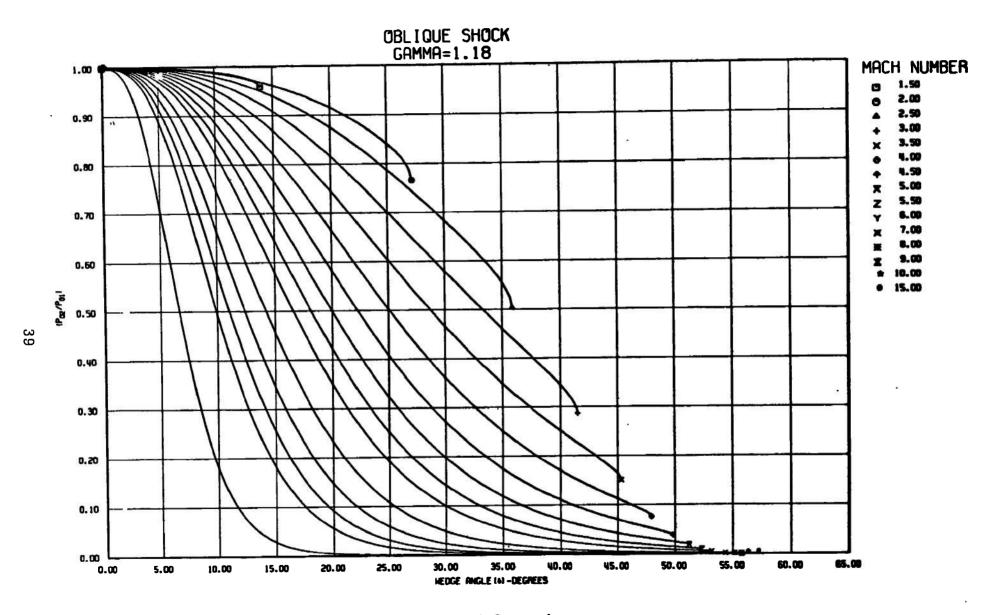


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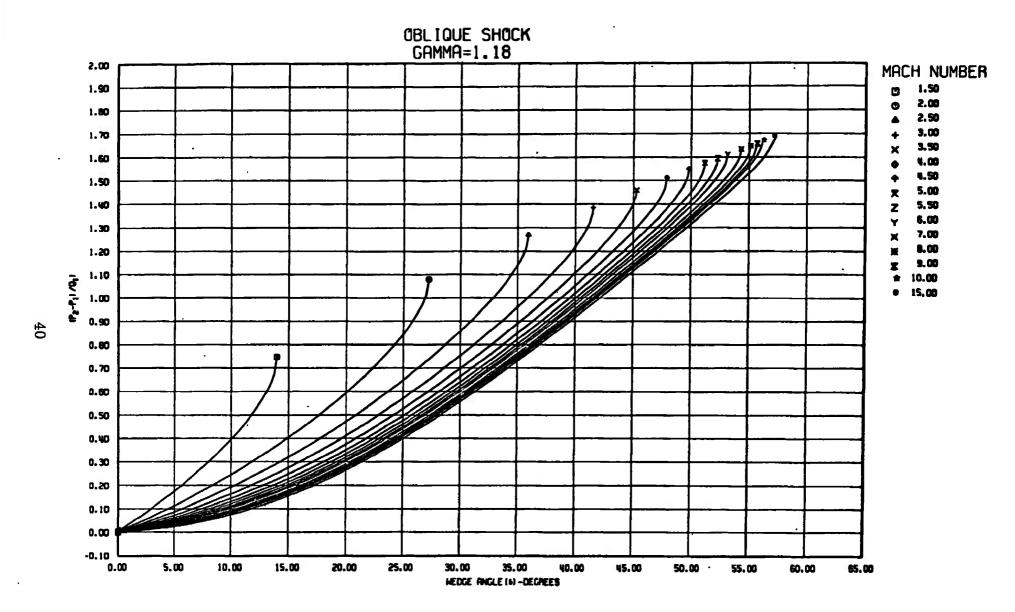


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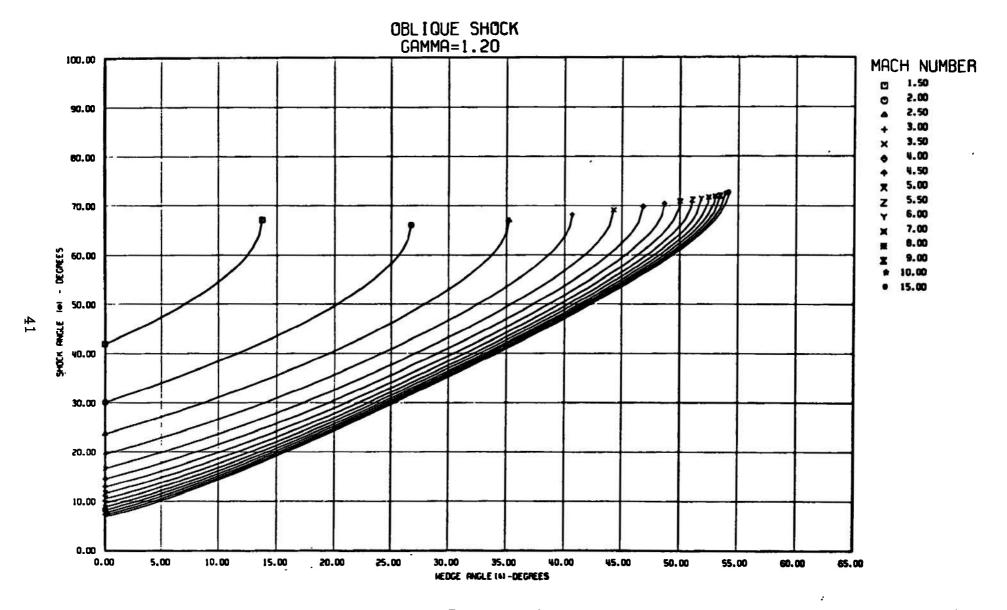


Fig. 7 y = 1.20

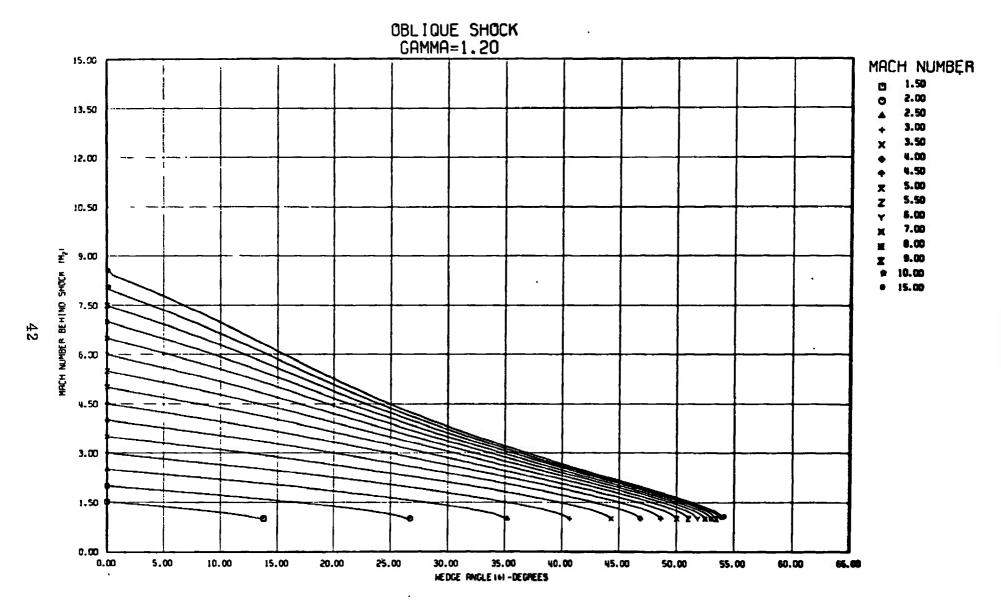


Fig. 7 Continued

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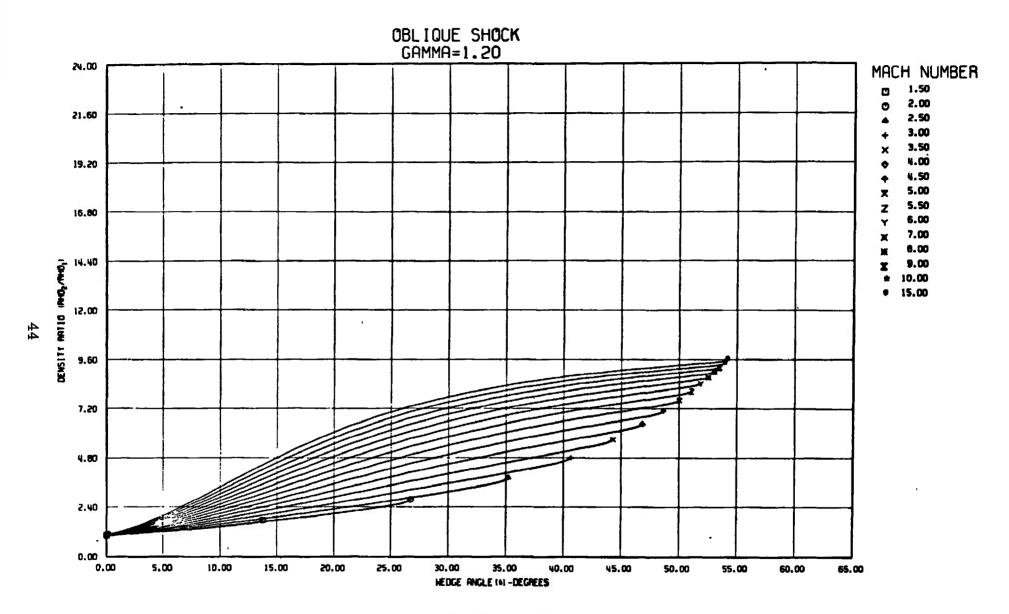


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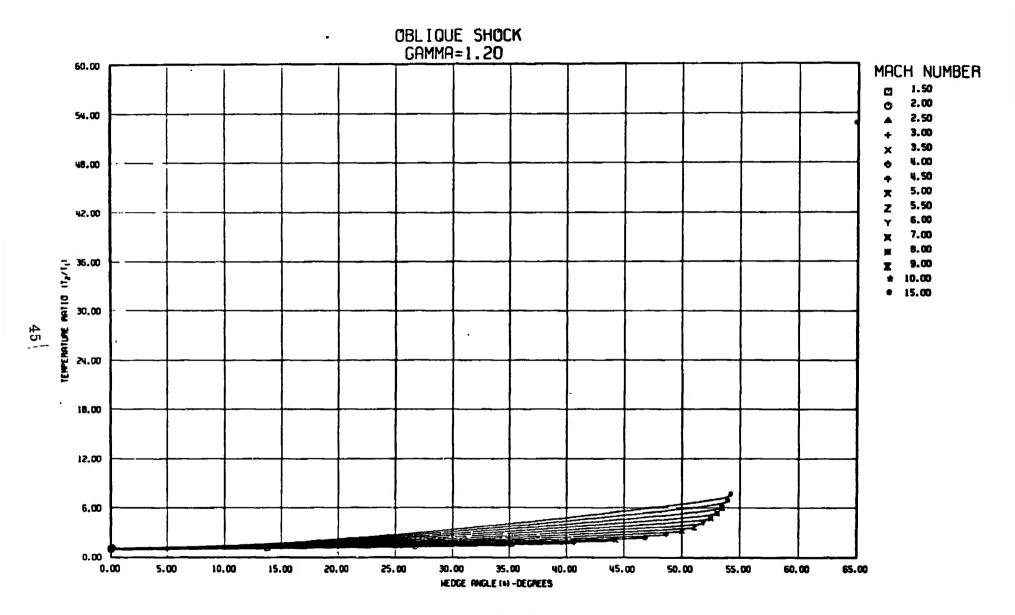


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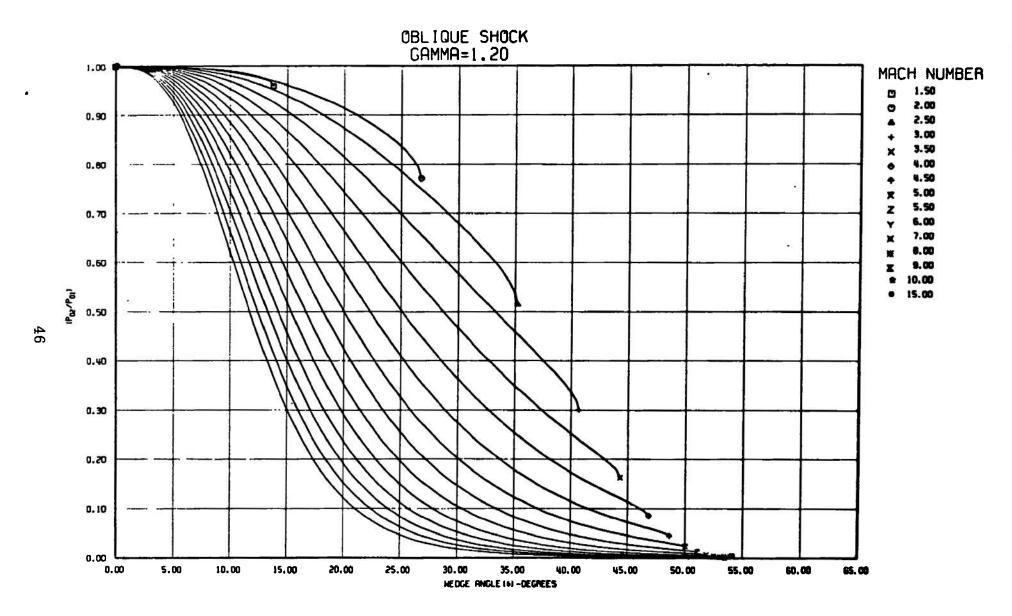


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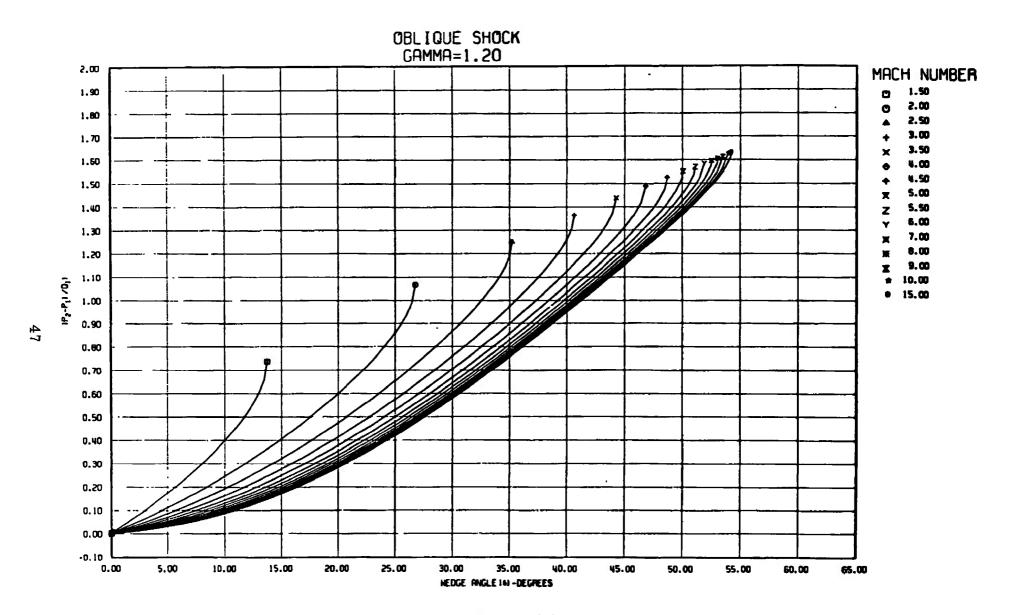


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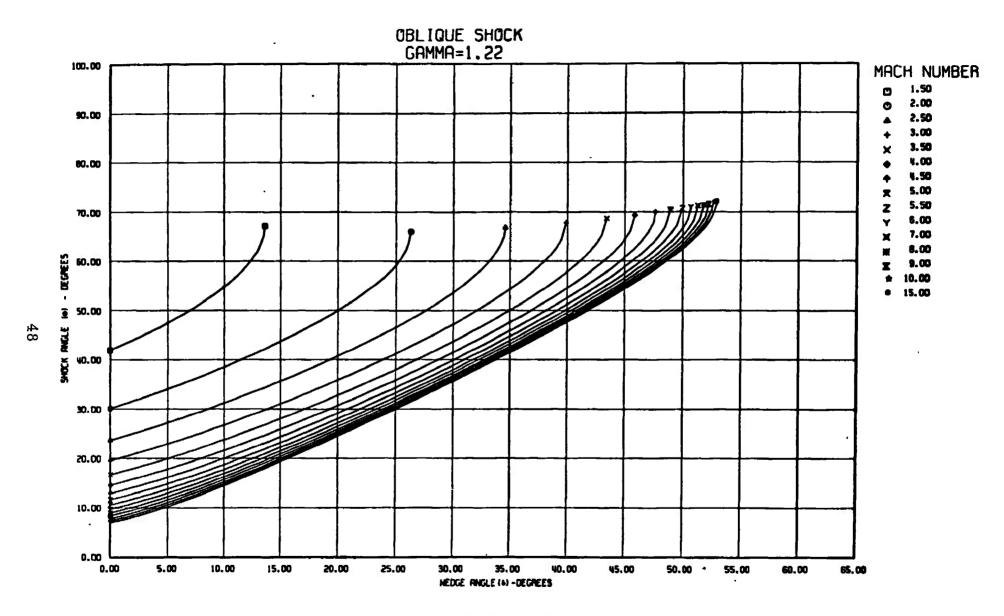


Fig. 8 y = 1.22

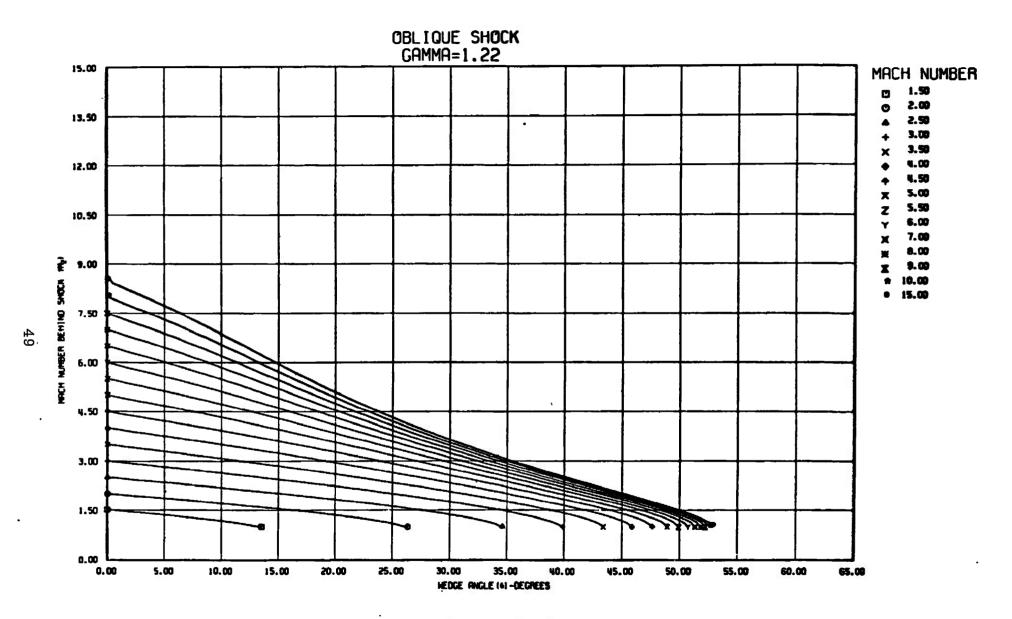


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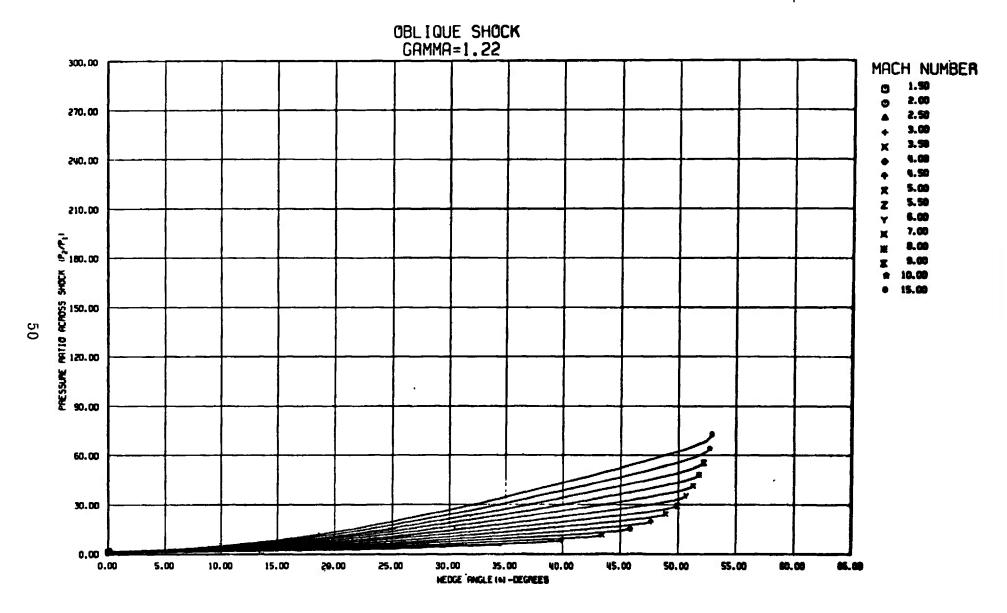


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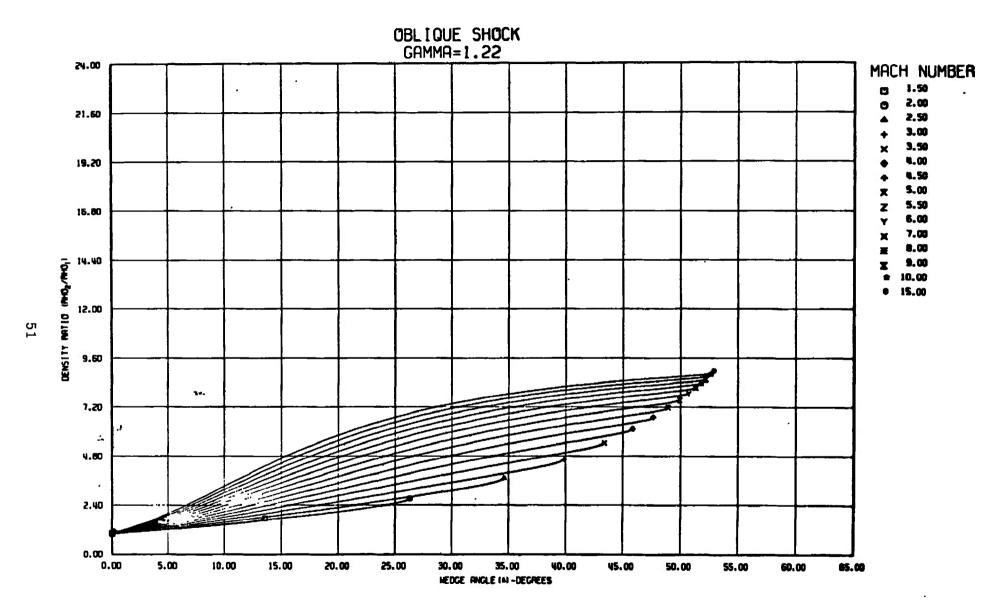


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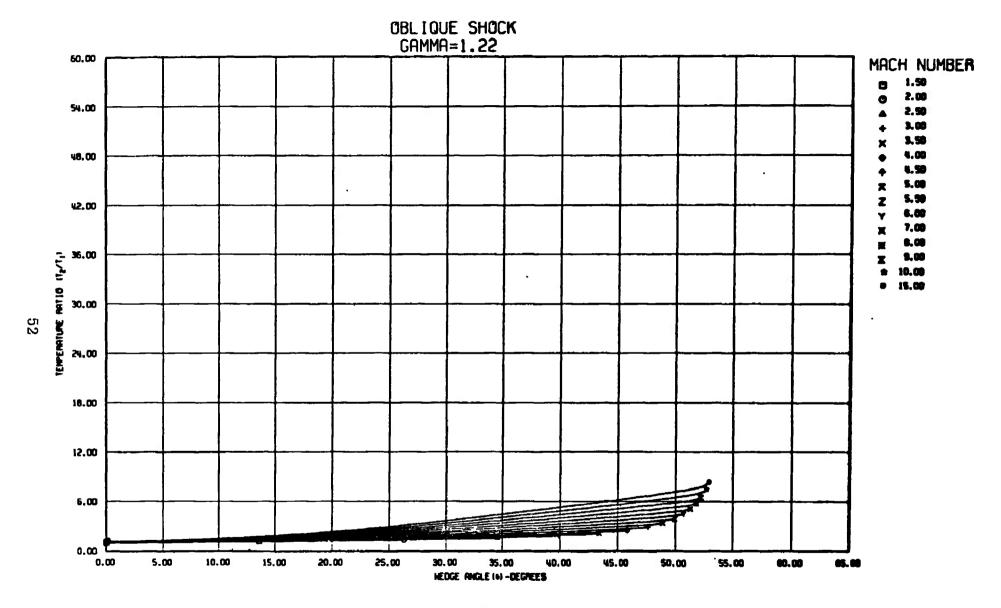


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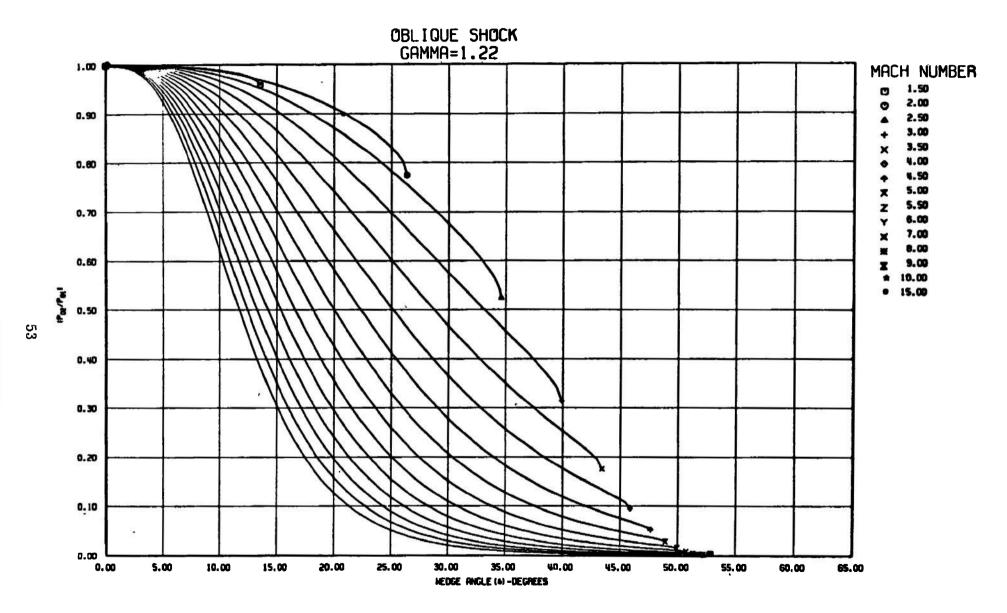


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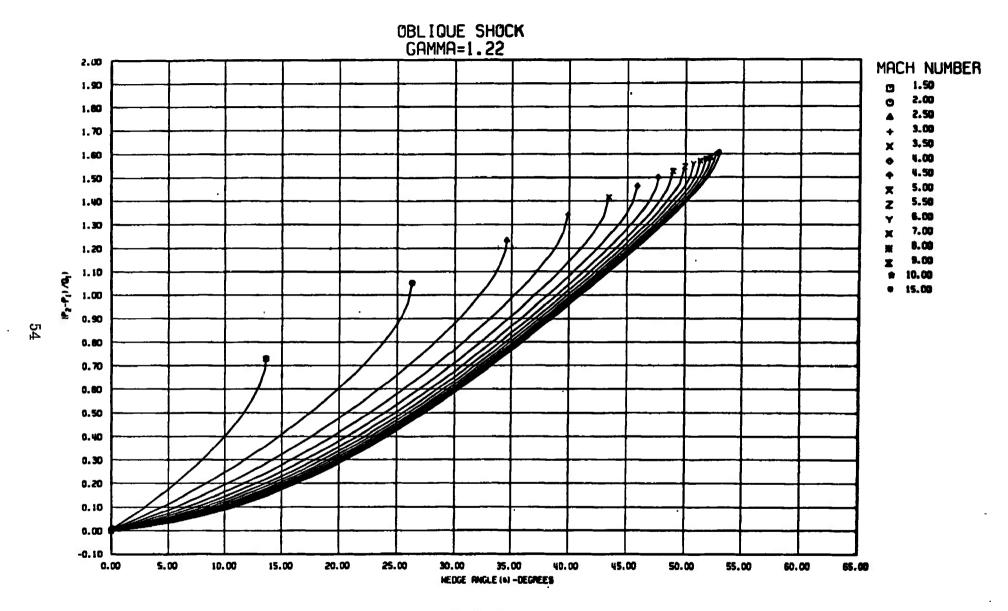


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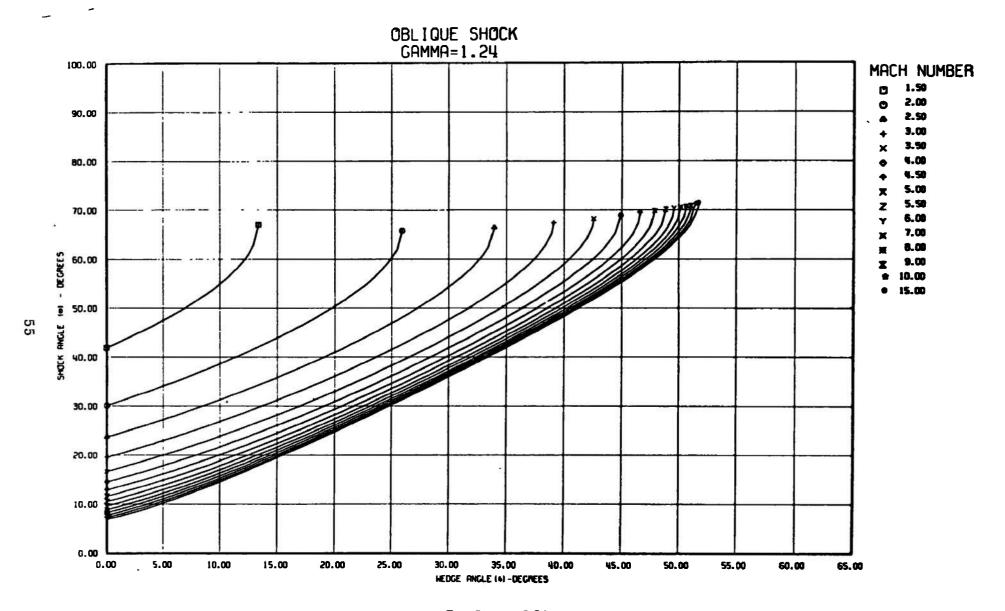


Fig. 9 $\gamma = 1.24$

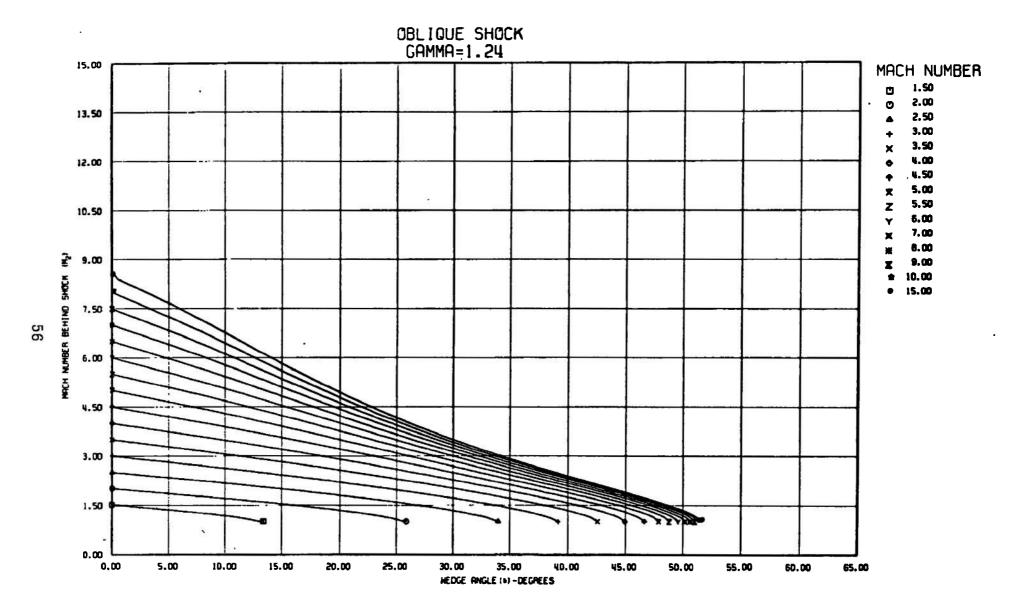


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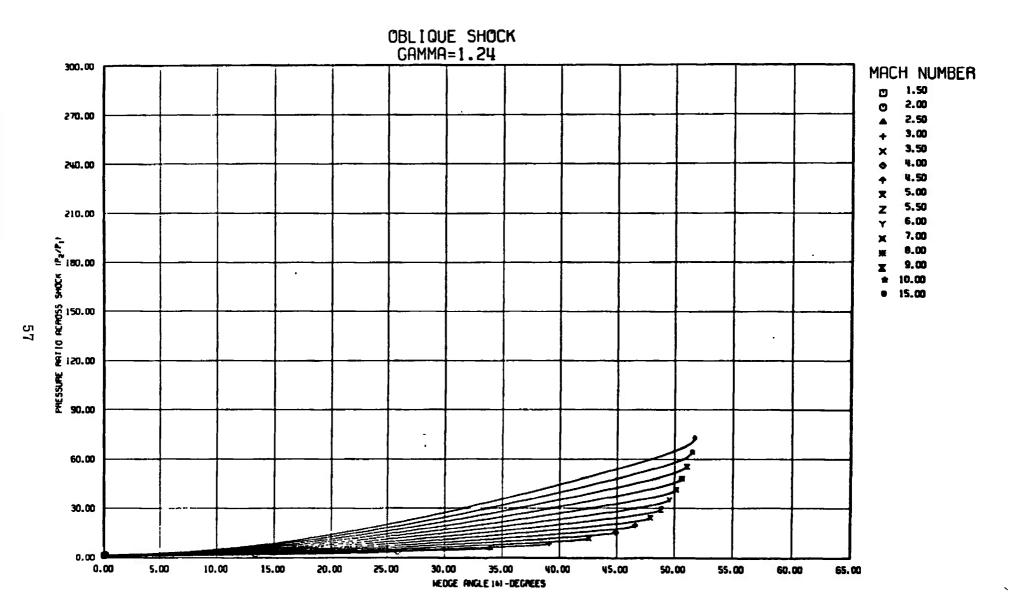


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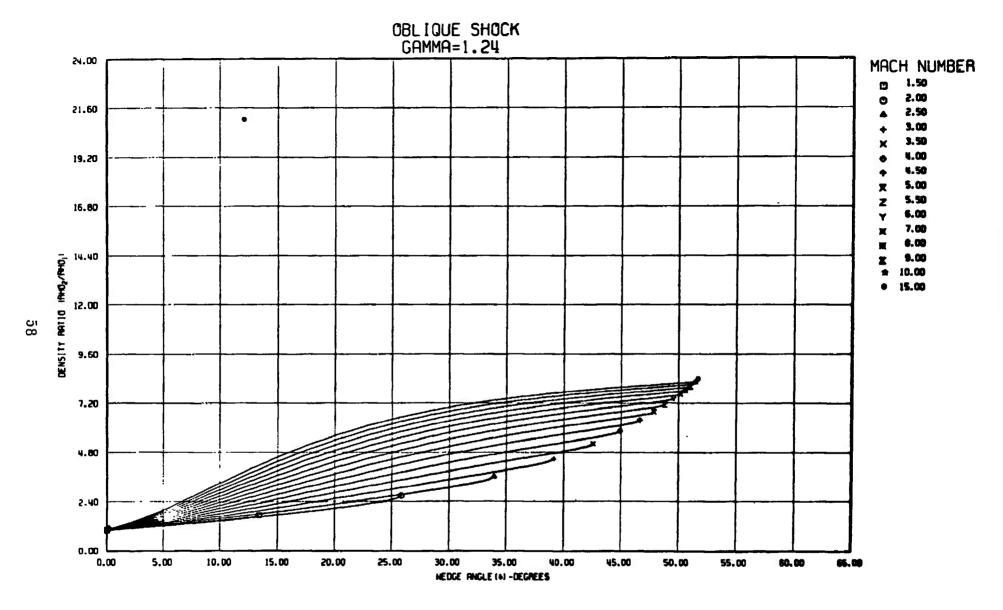


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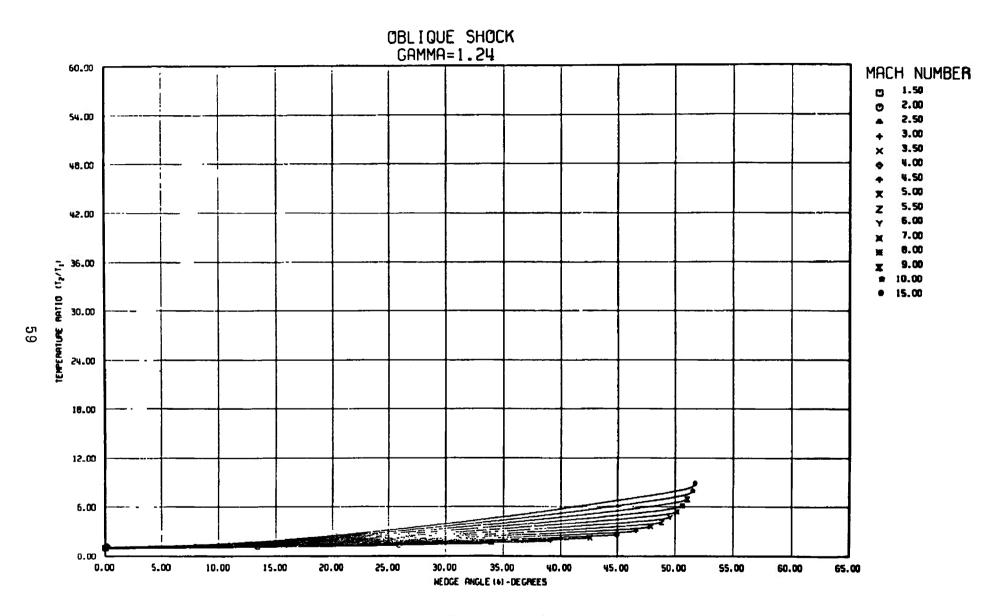


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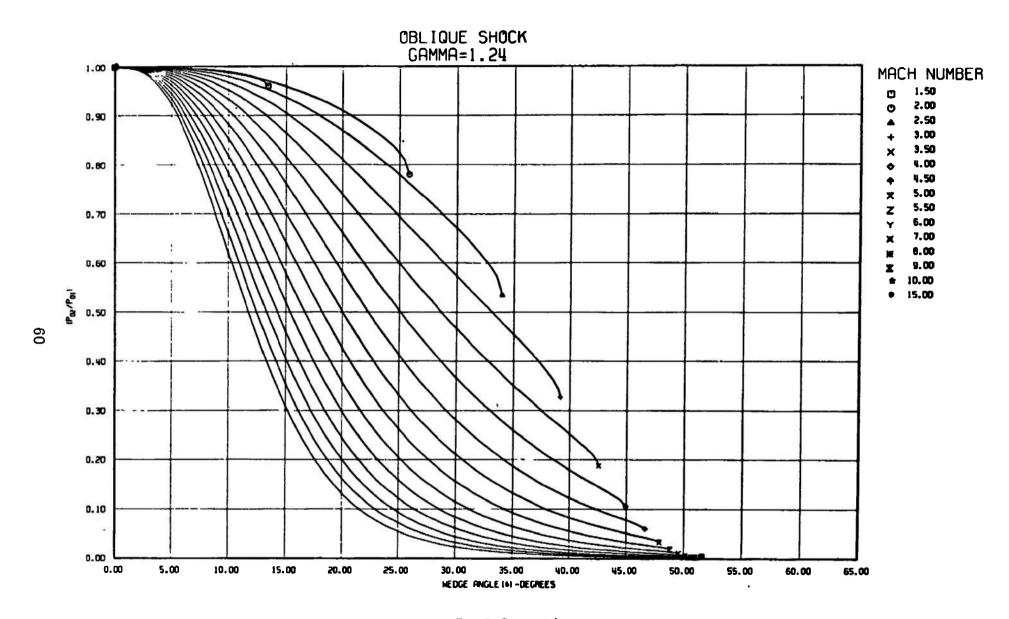


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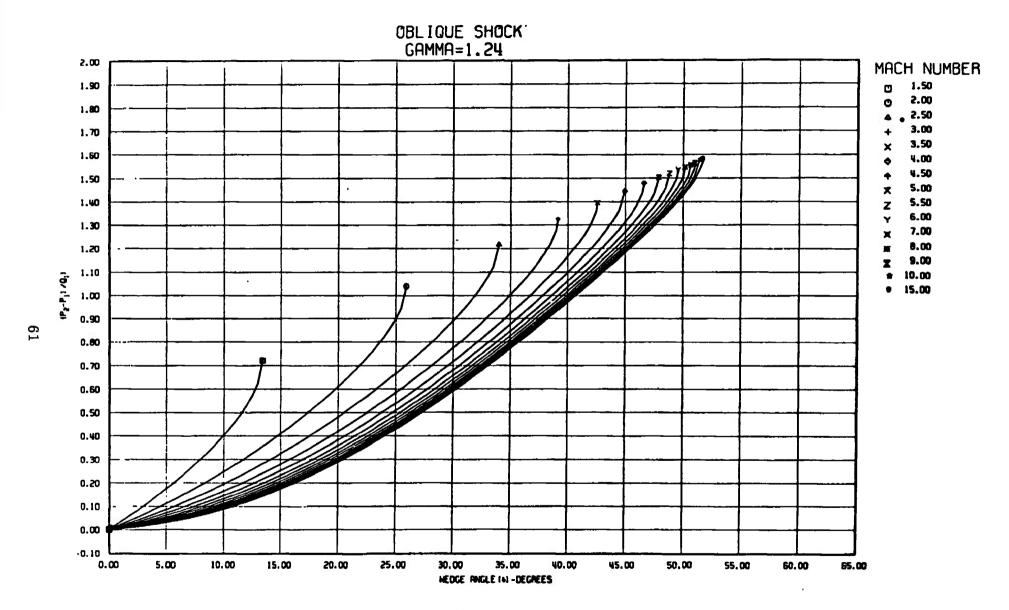


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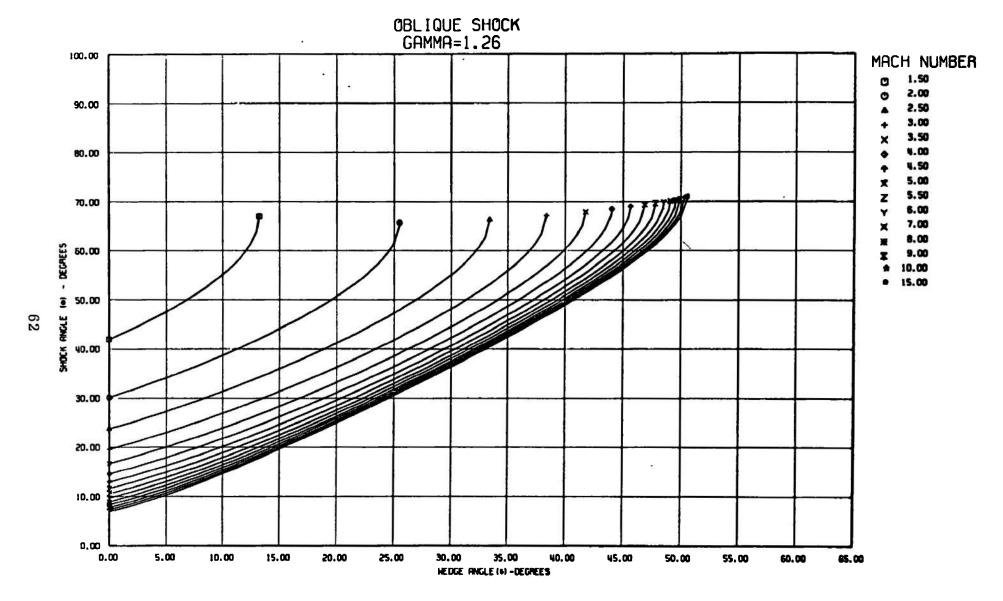


Fig. 10 y - 1.26

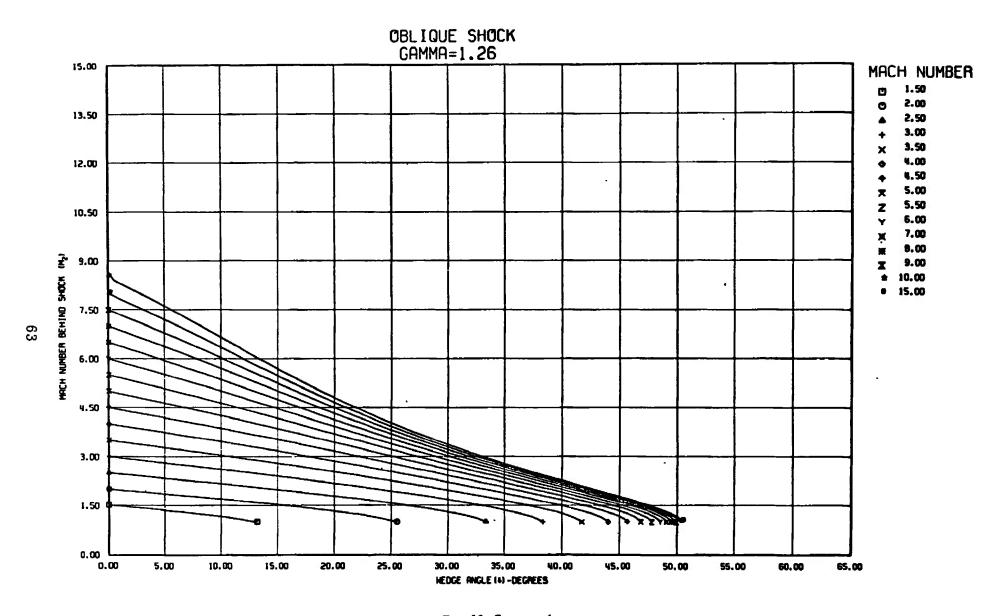


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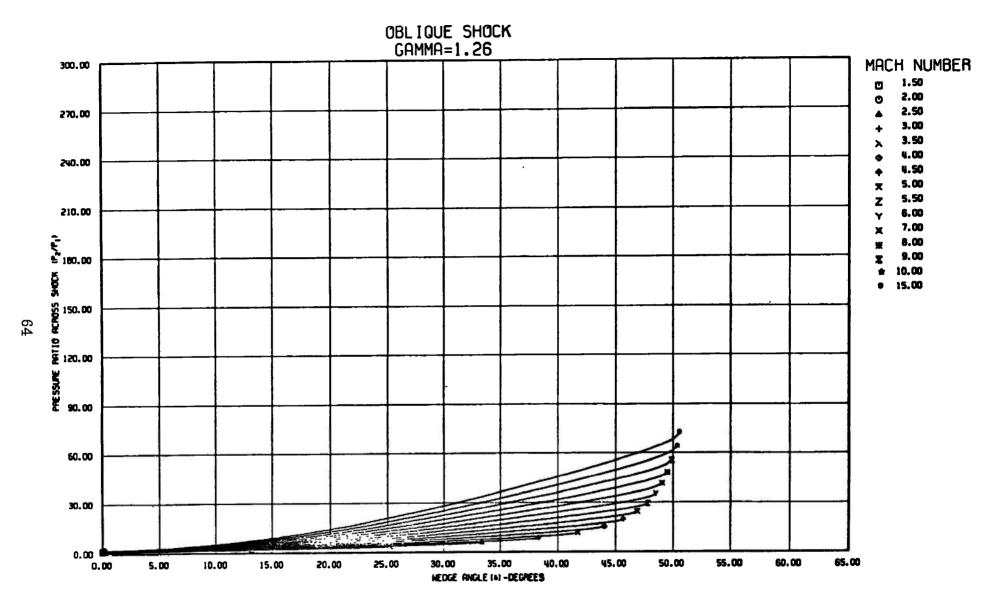


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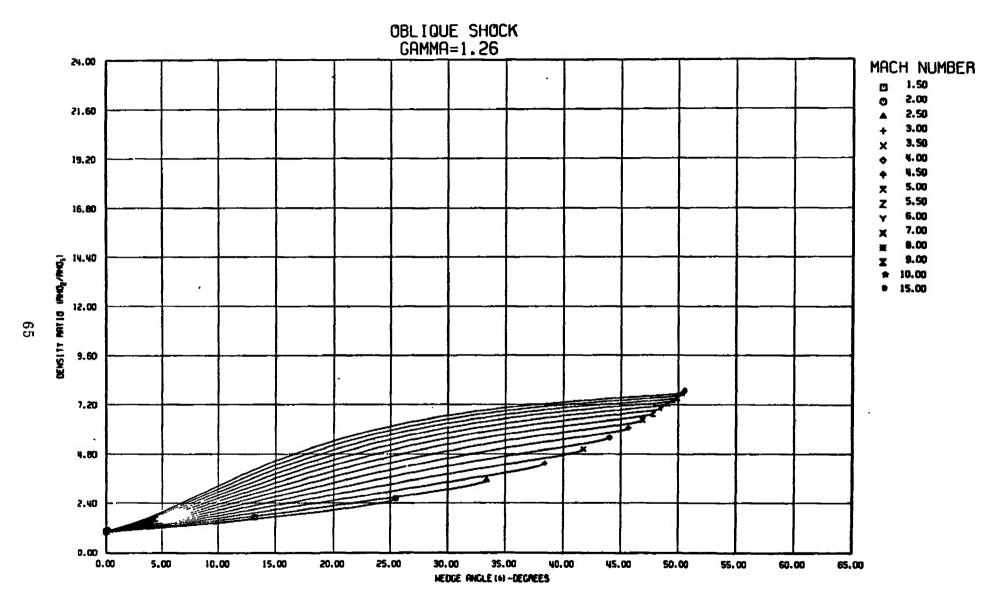


Fig. 10 Continued

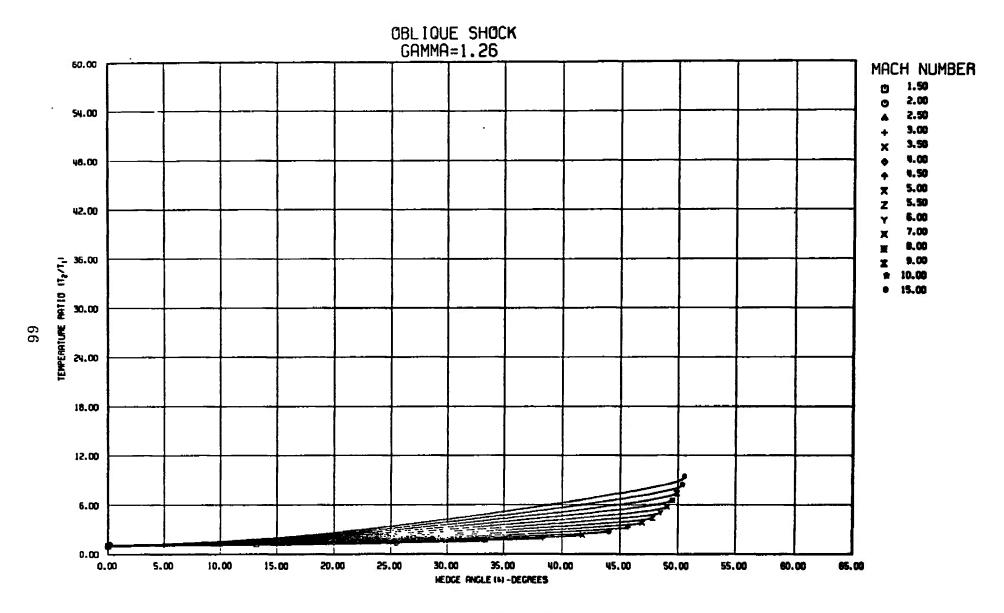


Fig. 10 Continued

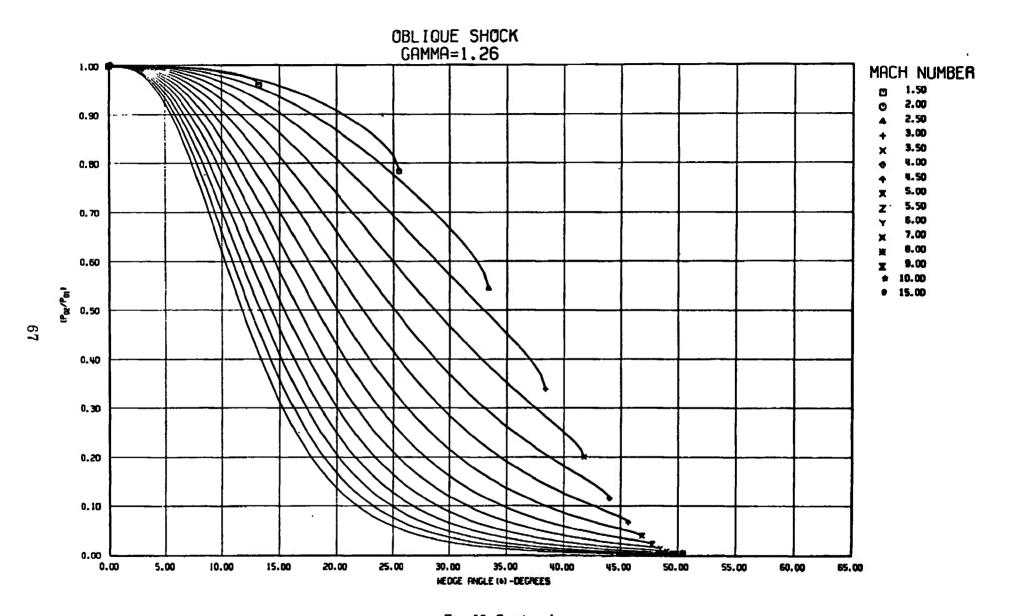


Fig. 10 Continued

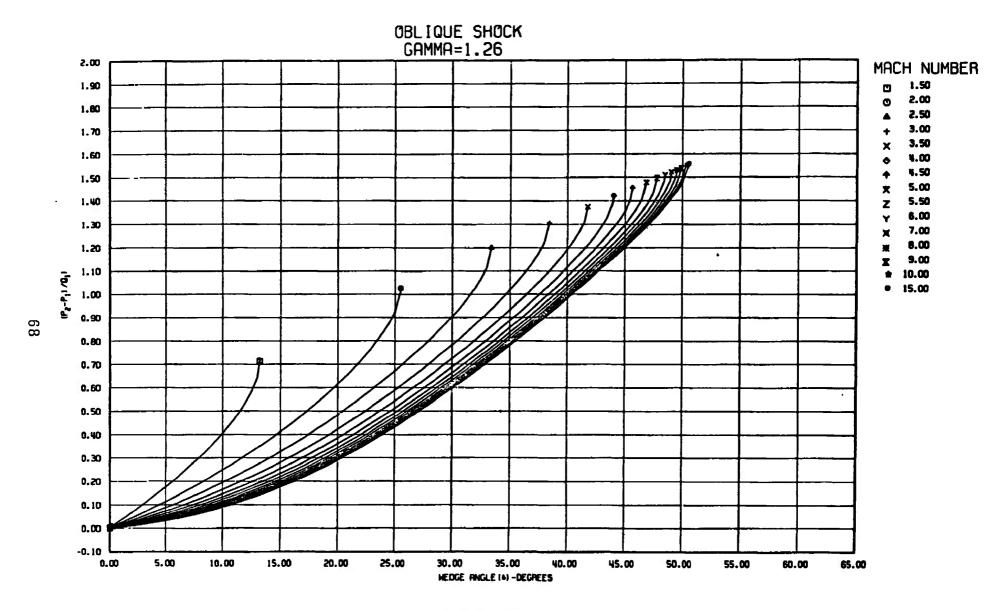


Fig. 10 Concluded

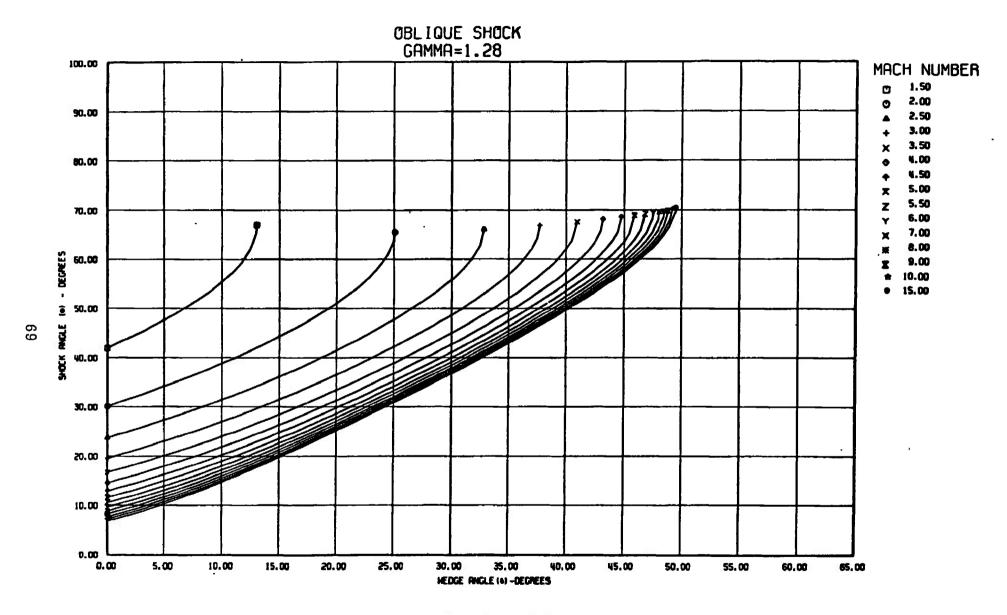


Fig. 11 $\gamma = 1.28$

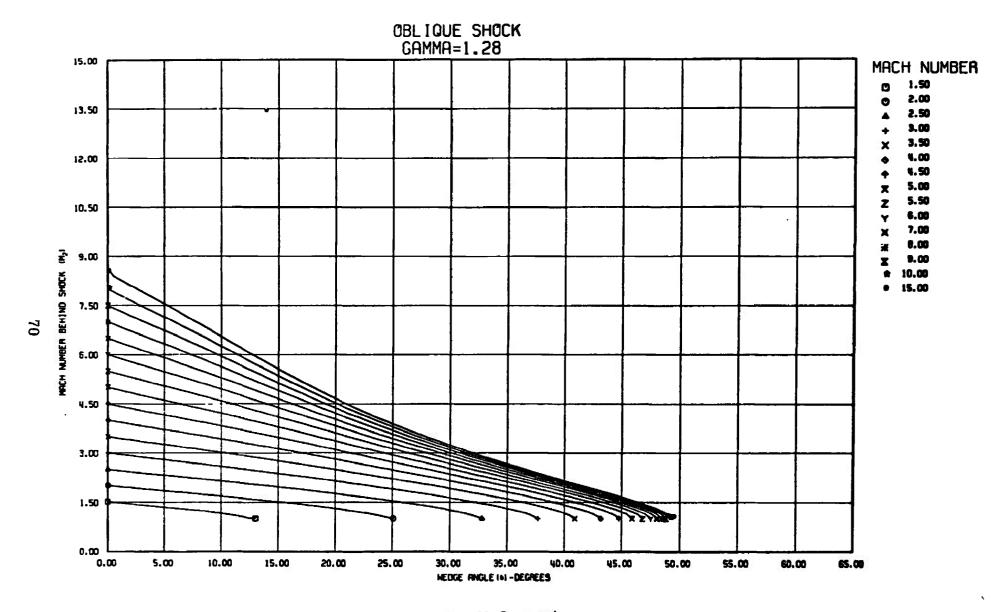


Fig. 11 Continued

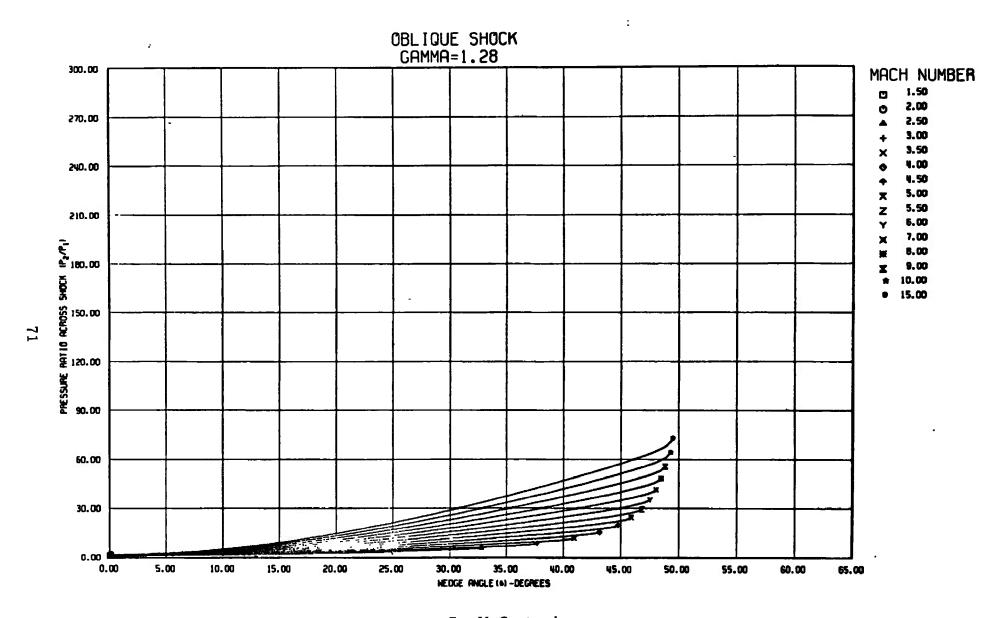


Fig. 11 Continued

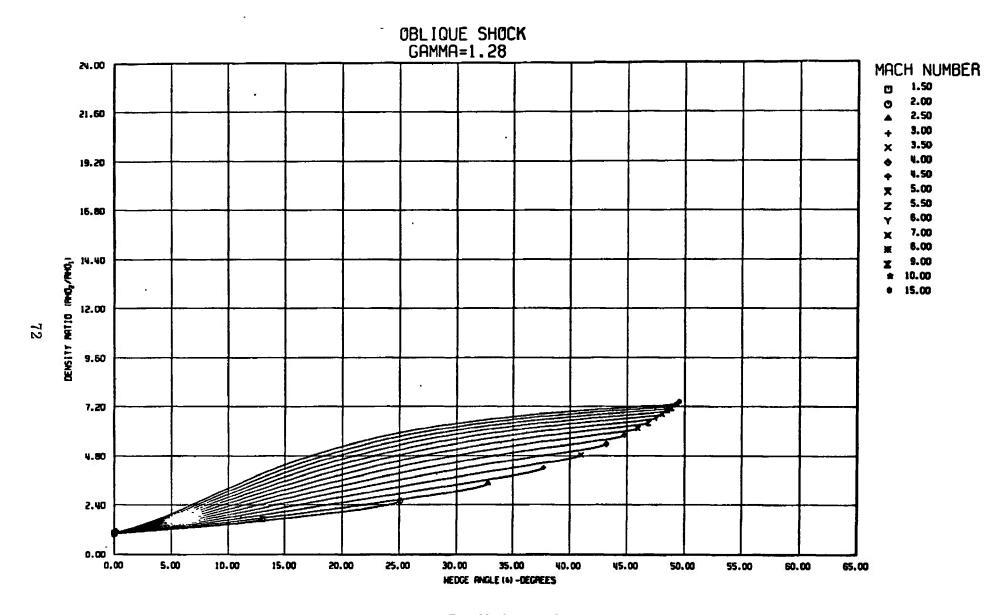


Fig. 11 Continued

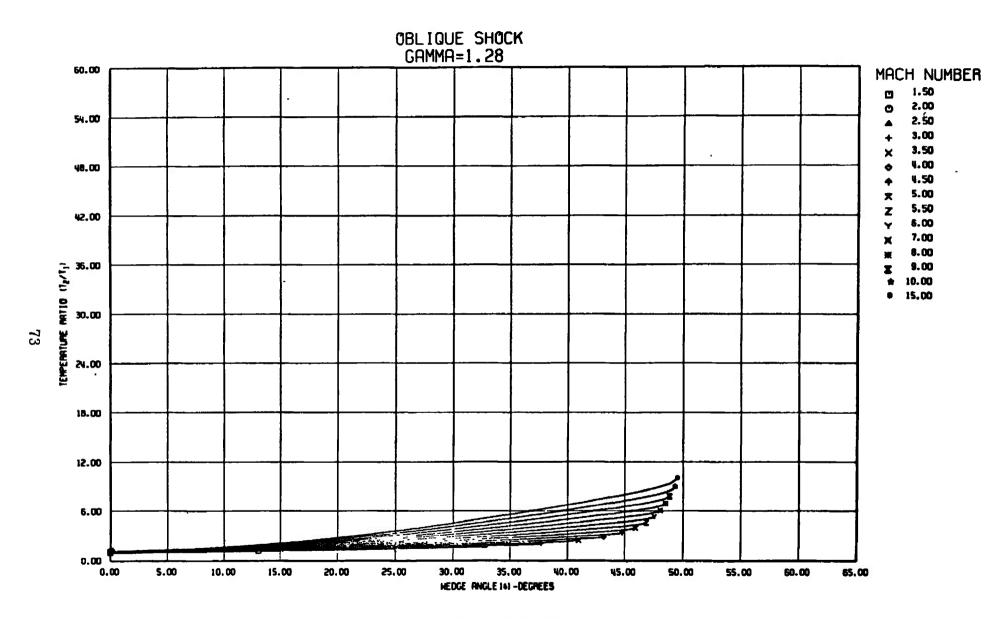


Fig. 11 Continued

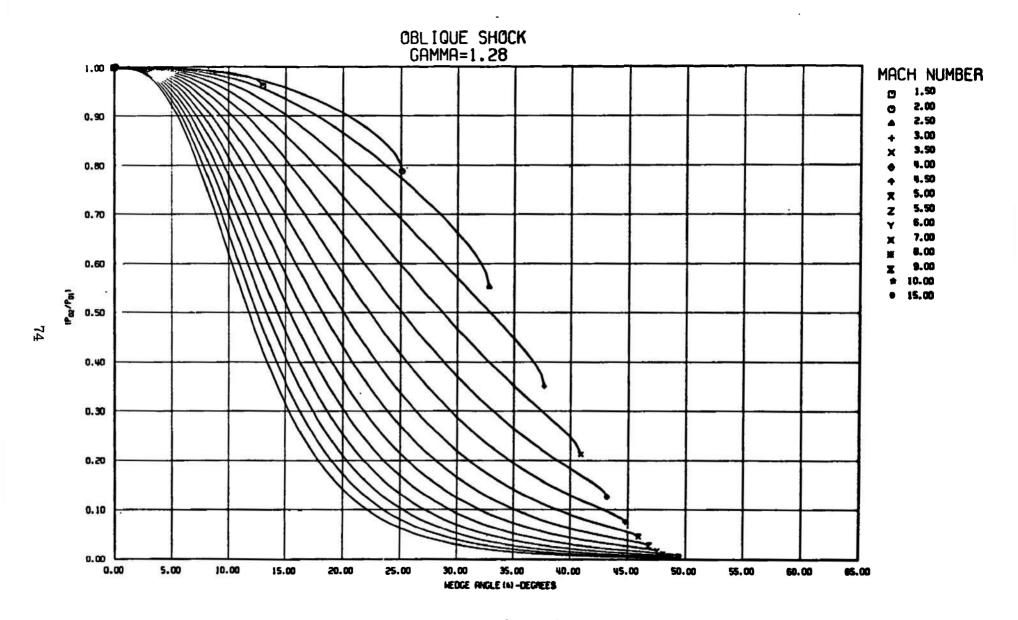


Fig. 11 Continued

Fig. 11 Concluded

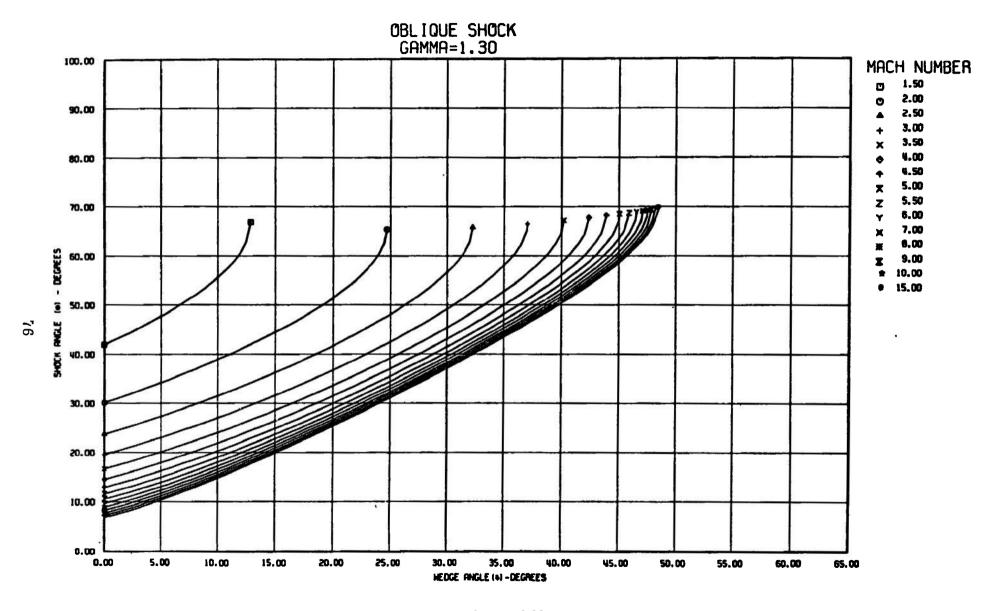


Fig. 12 y = 1.30

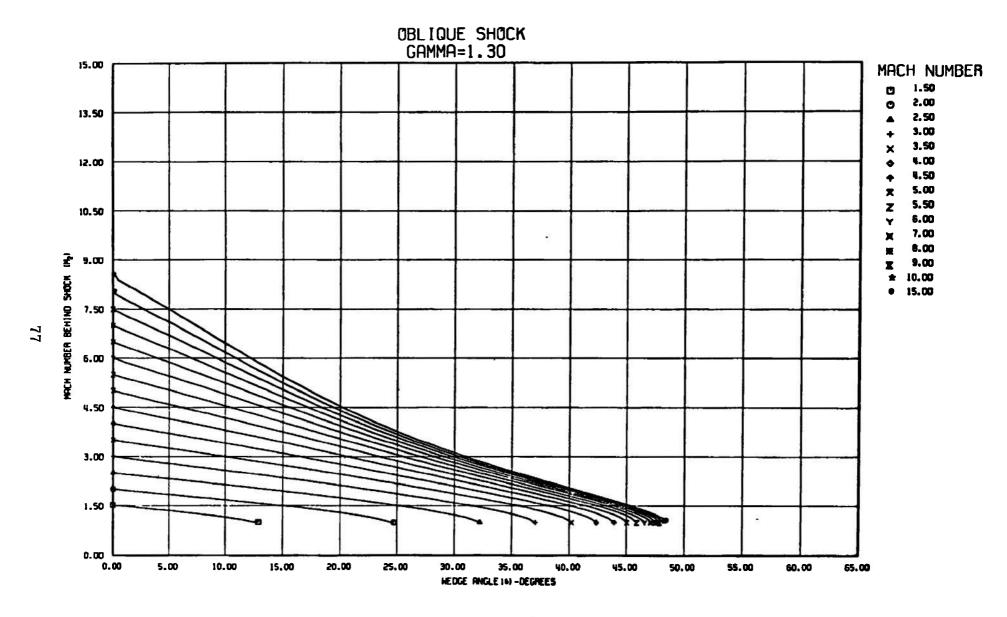


Fig. 12 Continued

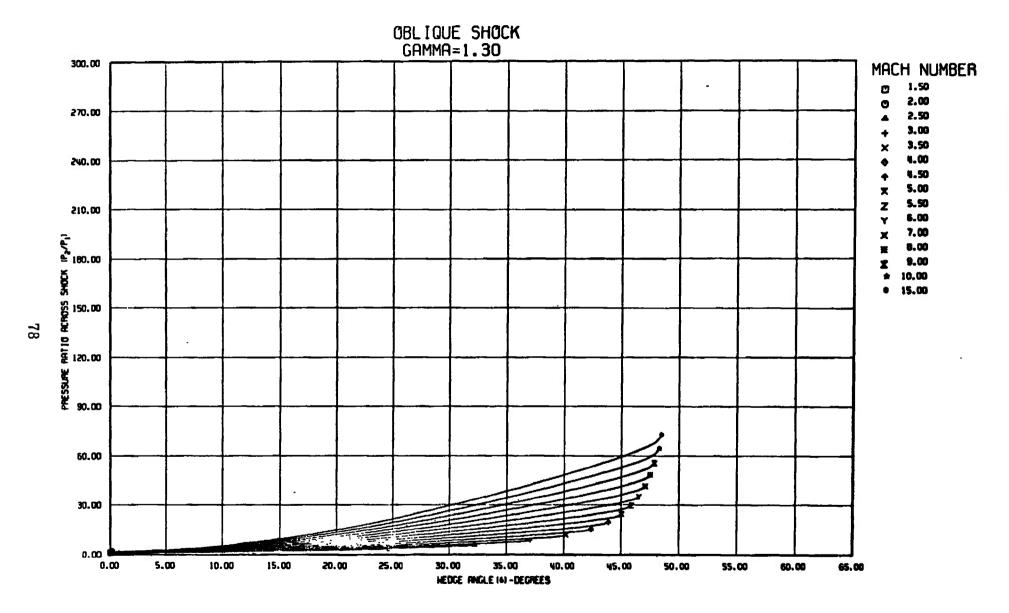


Fig. 12 Continued

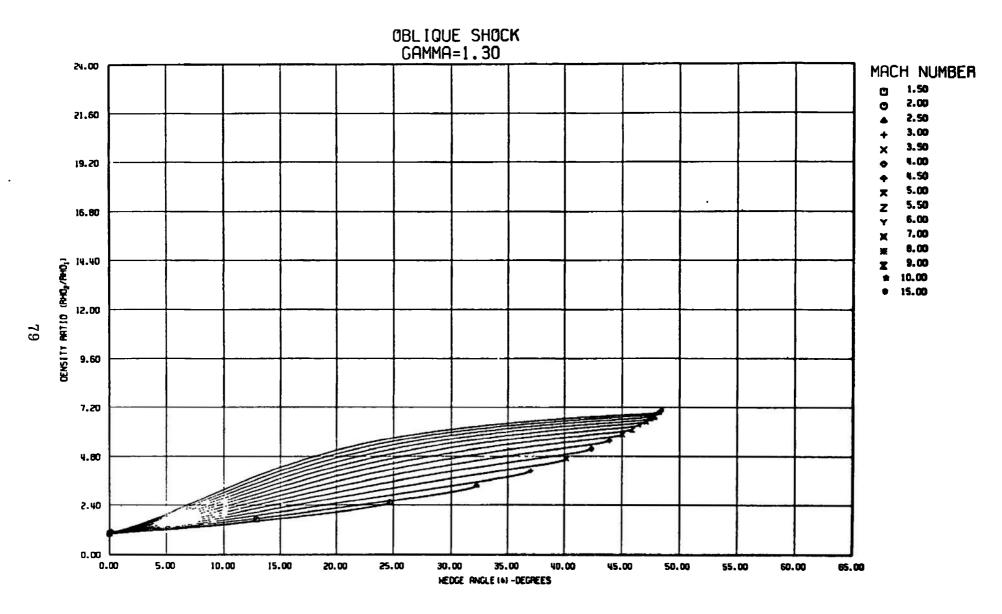


Fig. 12 Continued

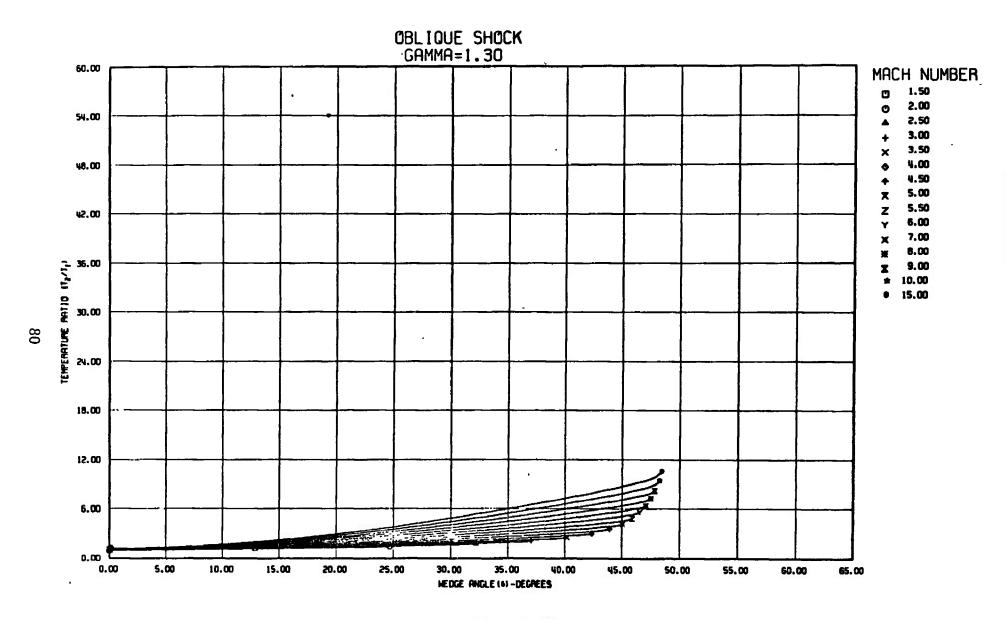


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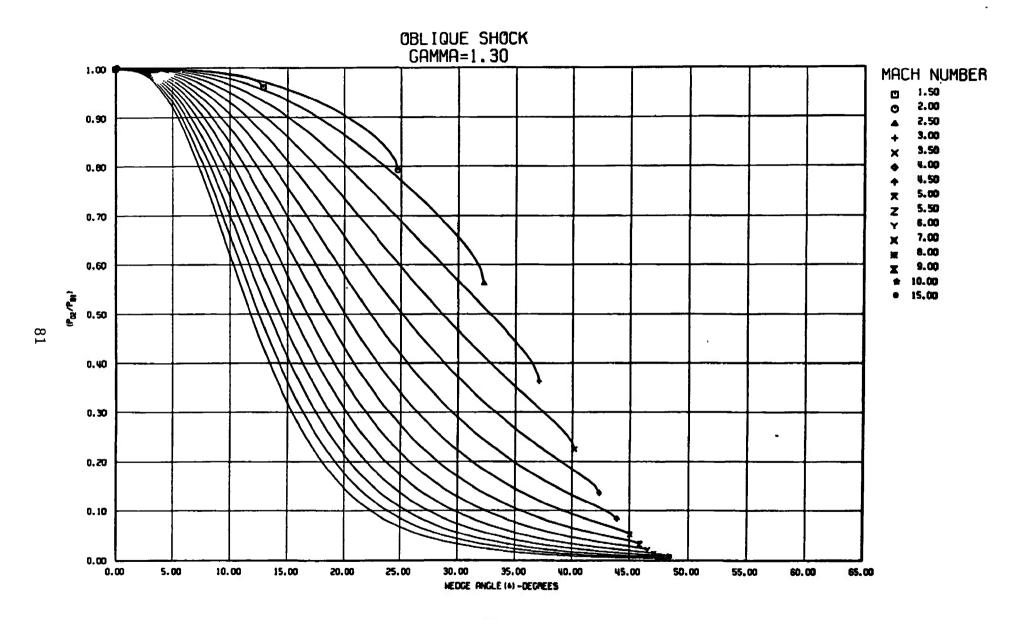


Fig. 12 Continued

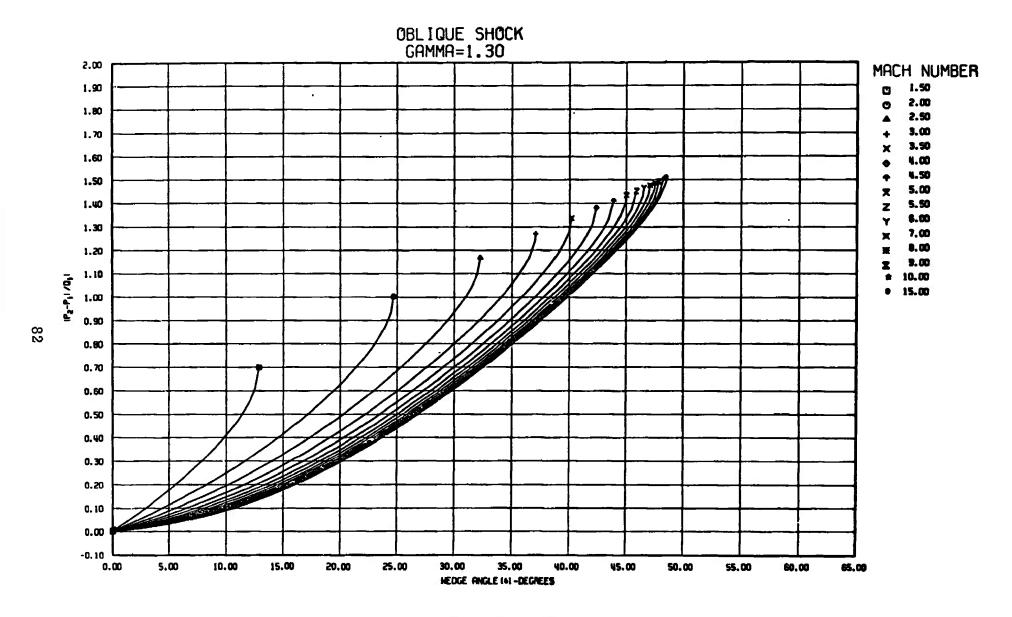


Fig. 12 Concluded

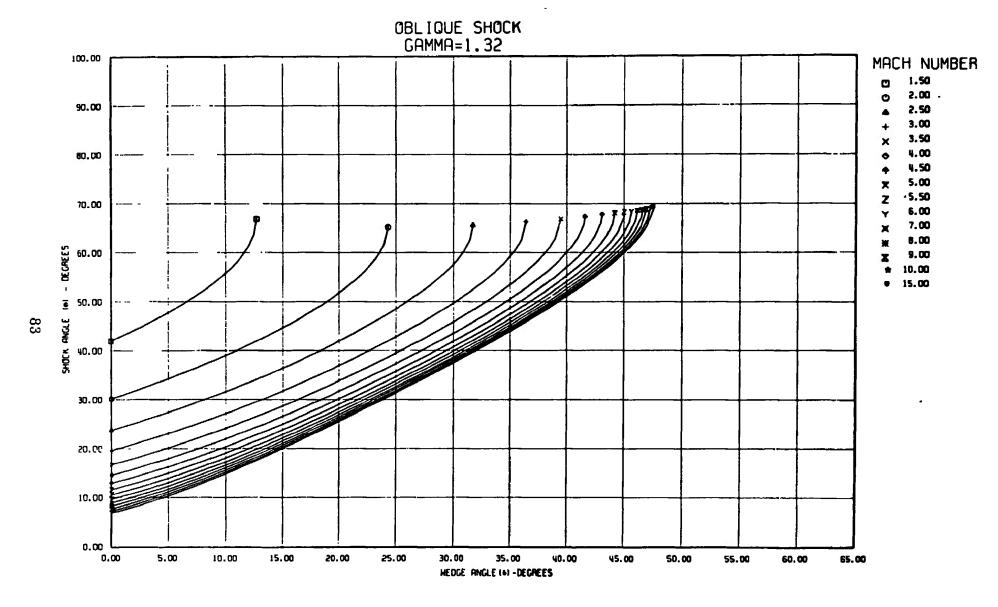


Fig. 13 $\gamma = 1.32$

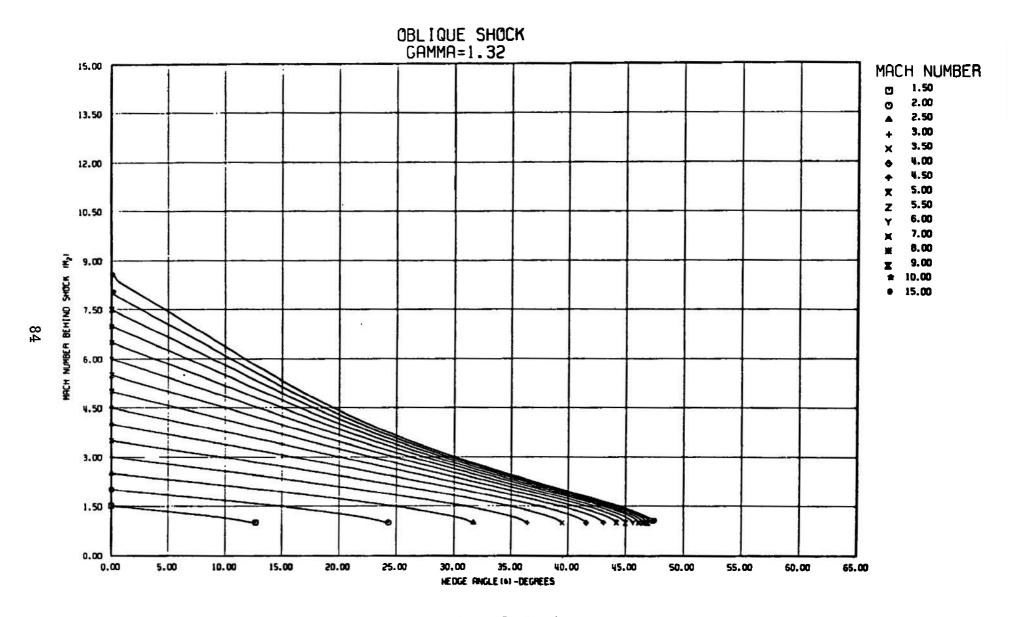


Fig. 13 Continued

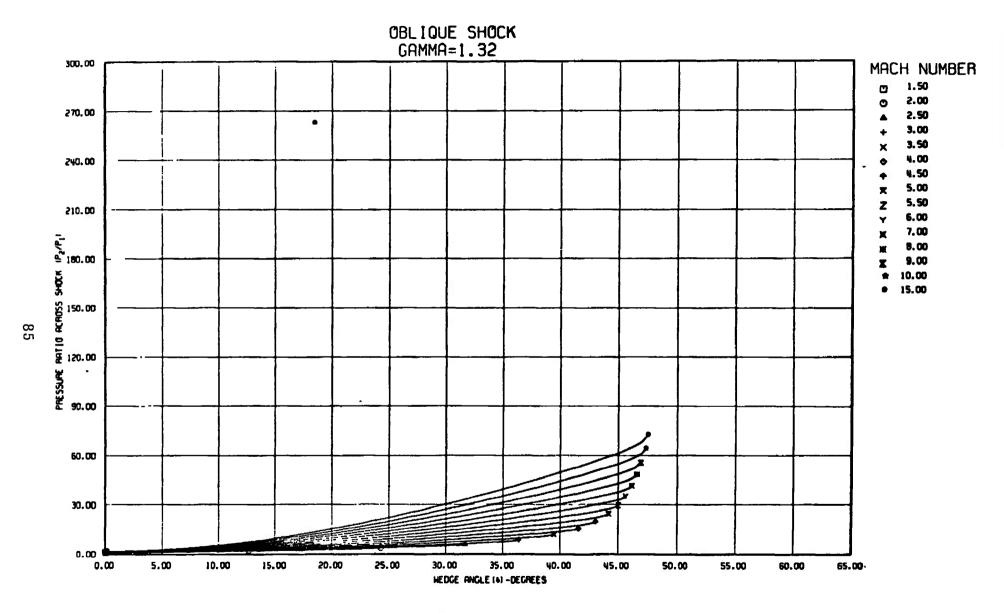


Fig. 13 Continued

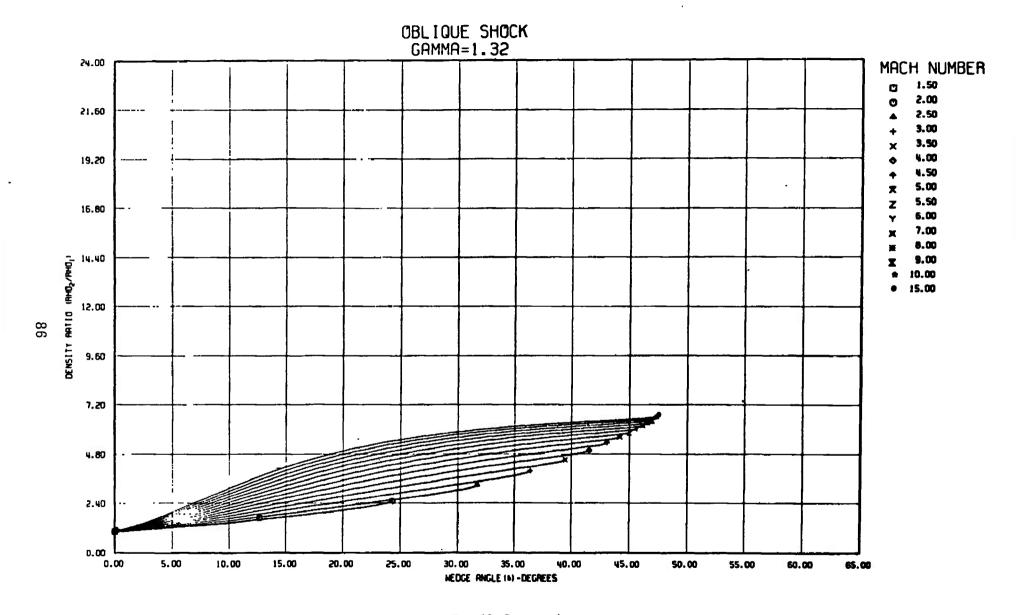


Fig. 13 Continued

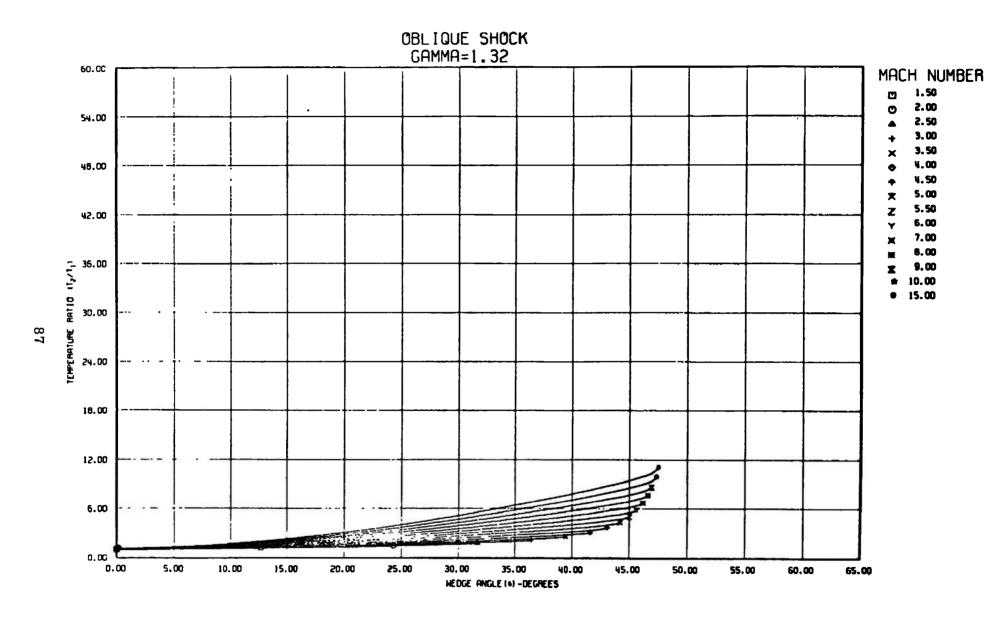


Fig. 13 Continued

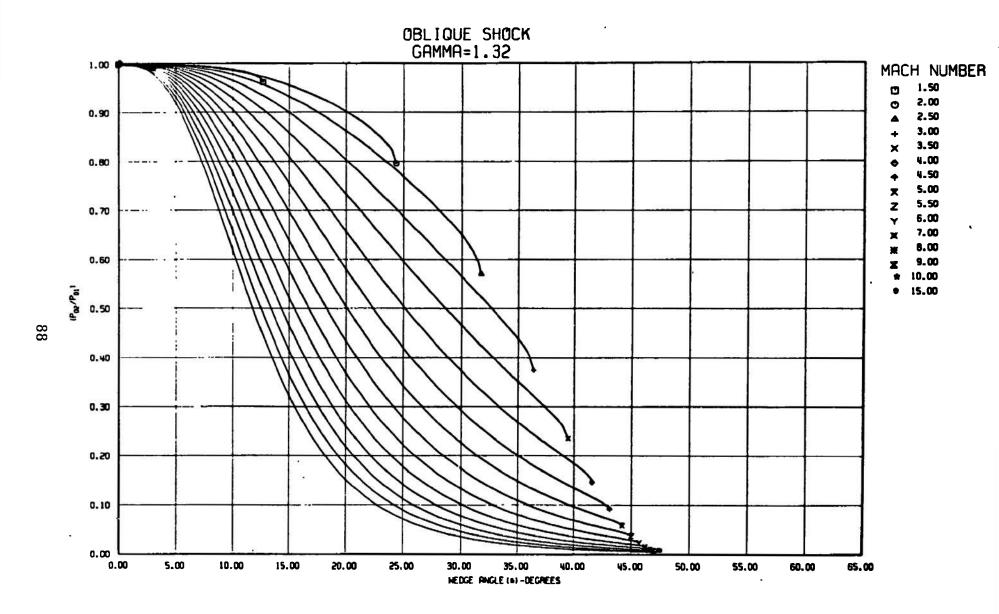


Fig. 13 Continued

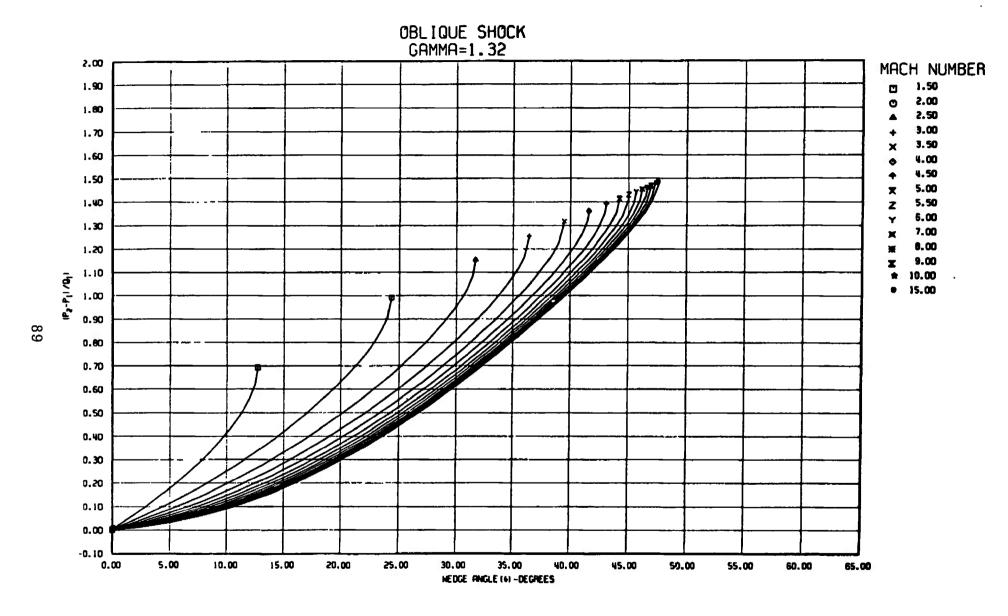


Fig. 13 Concluded

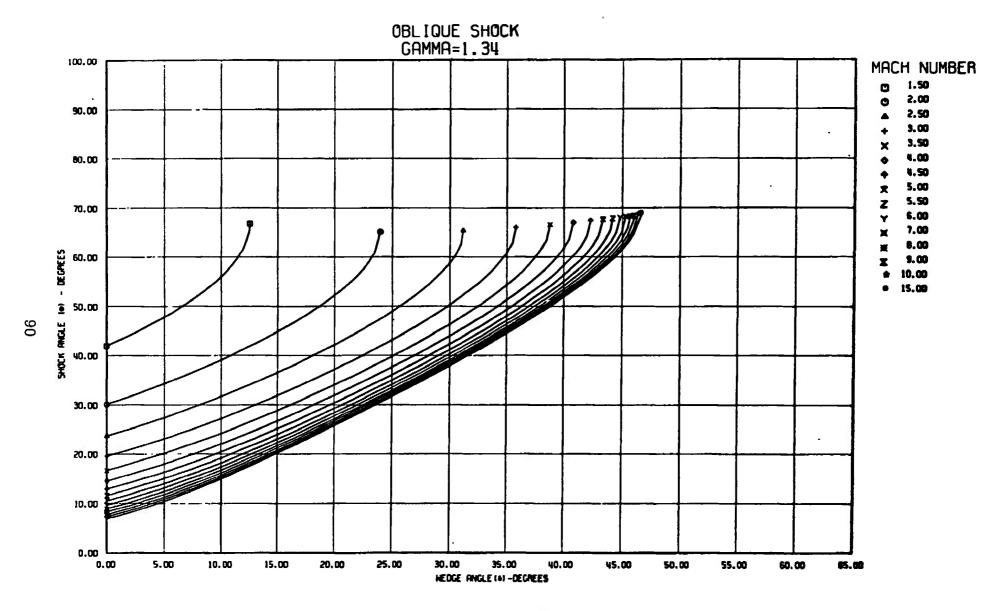


Fig. 14 y - 1.34

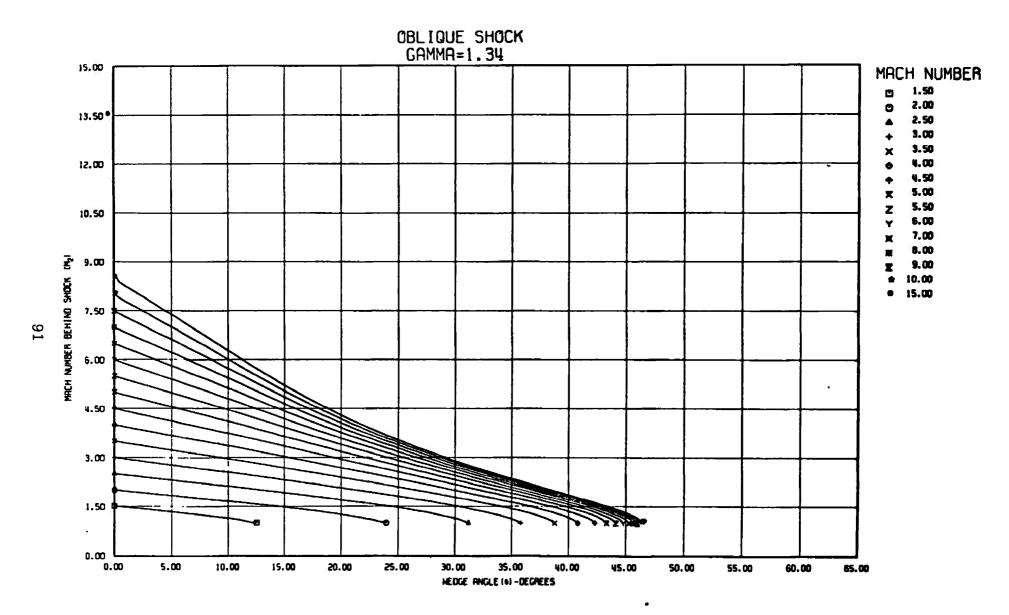


Fig. 14 Continued

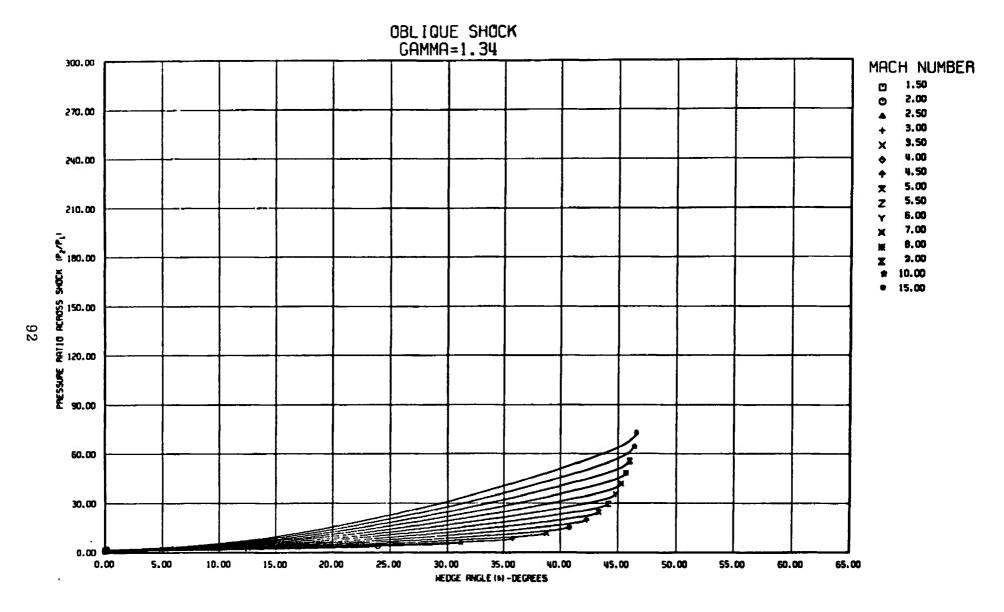


Fig. 14 Continued

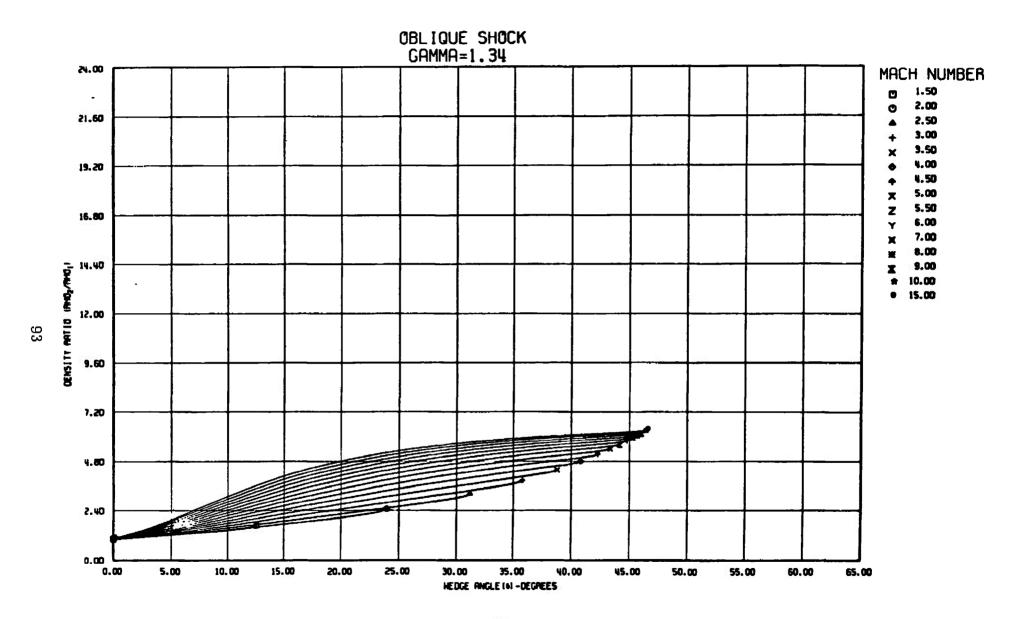


Fig. 14 Continued

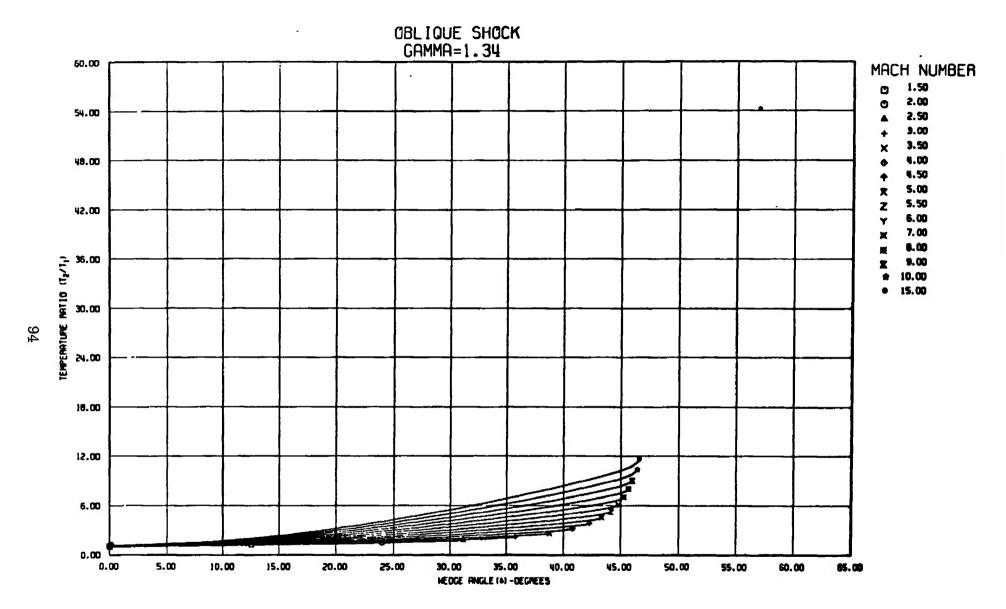


Fig. 14 Continued

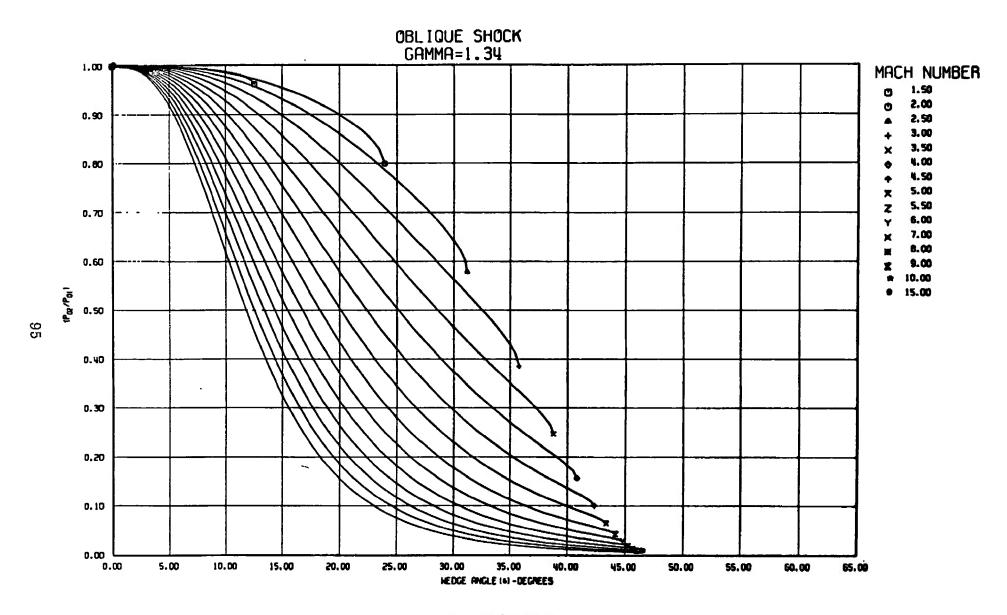


Fig. 14 Continued

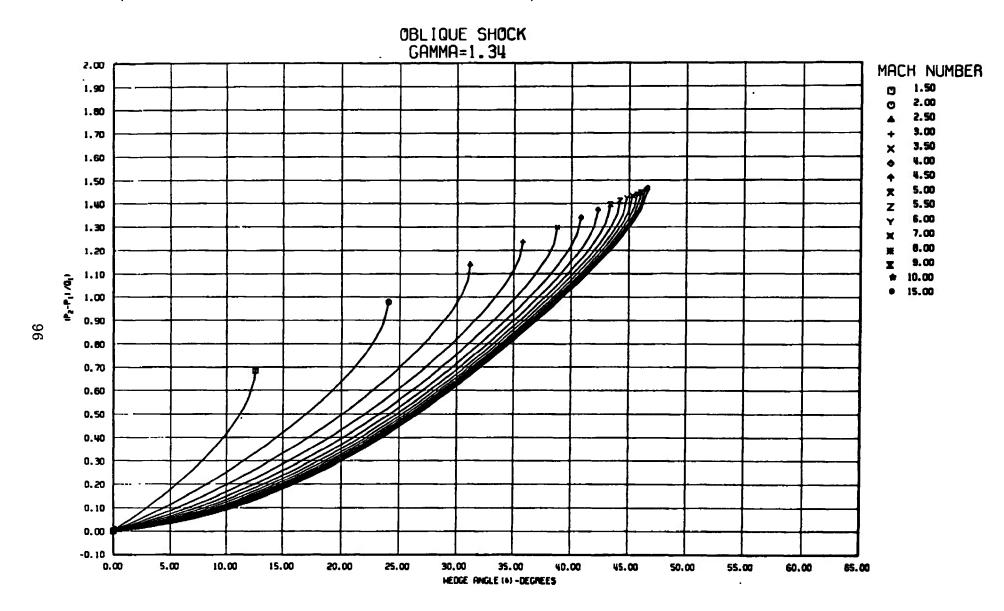


Fig. 14 Concluded

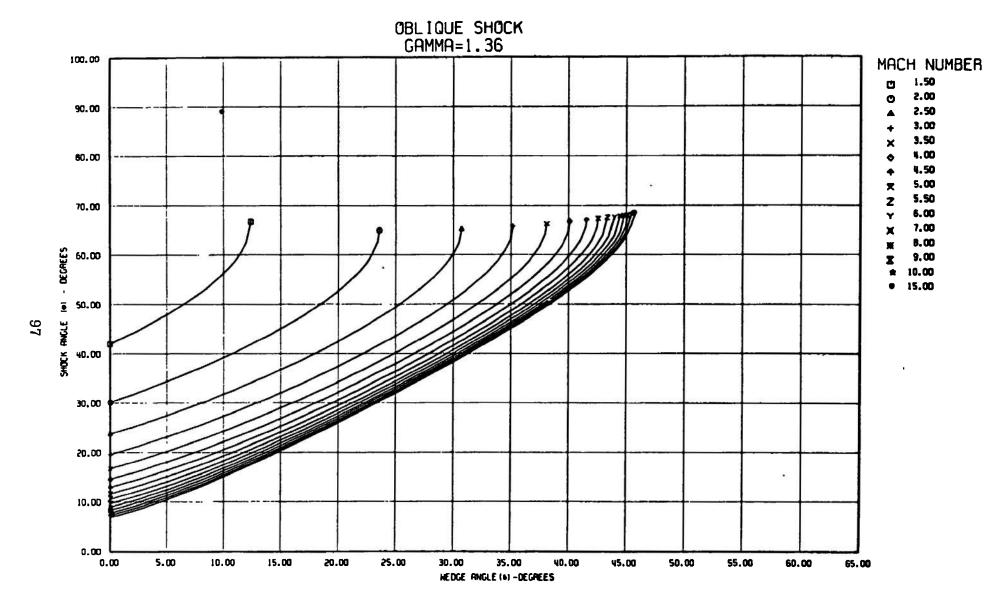


Fig. 15 y - 1.36

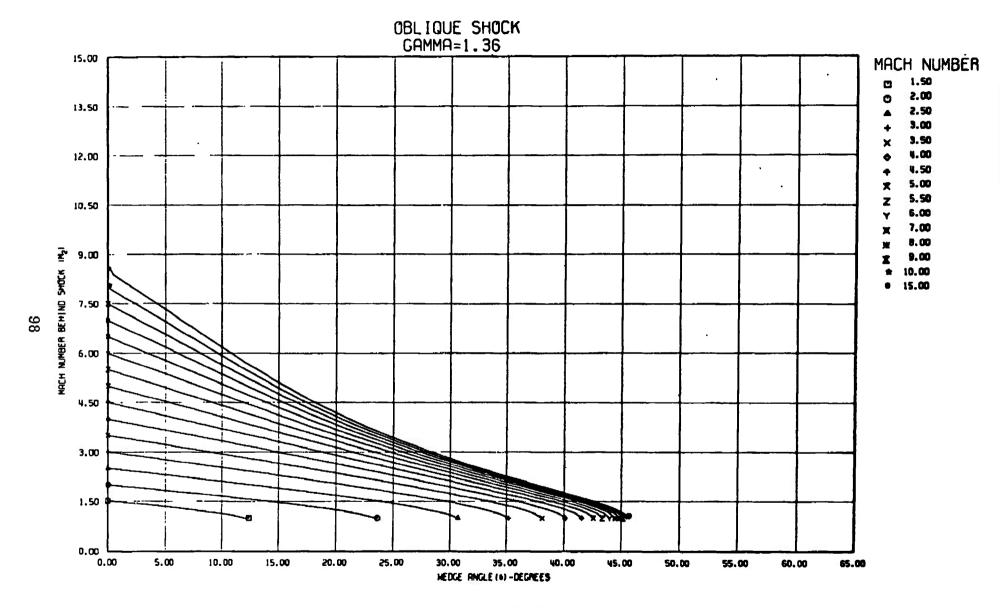


Fig. 15 Continued

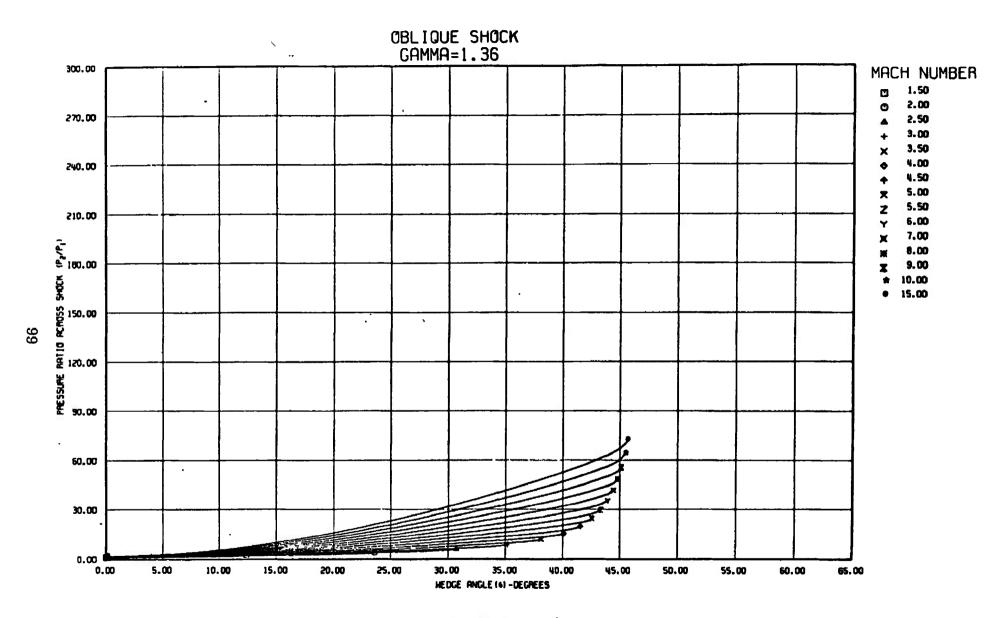


Fig. 15 Continued

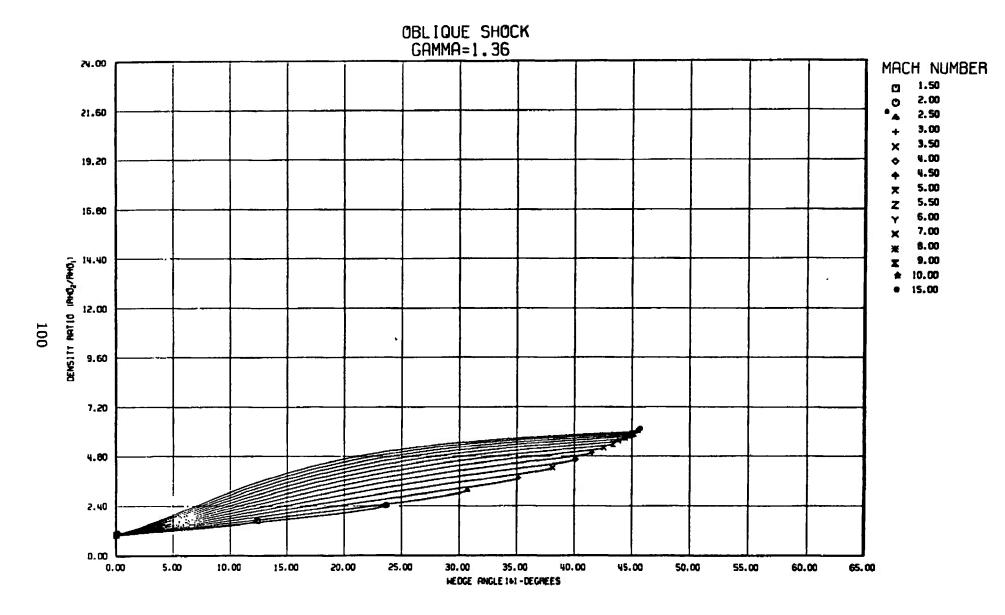


Fig. 15 Continued

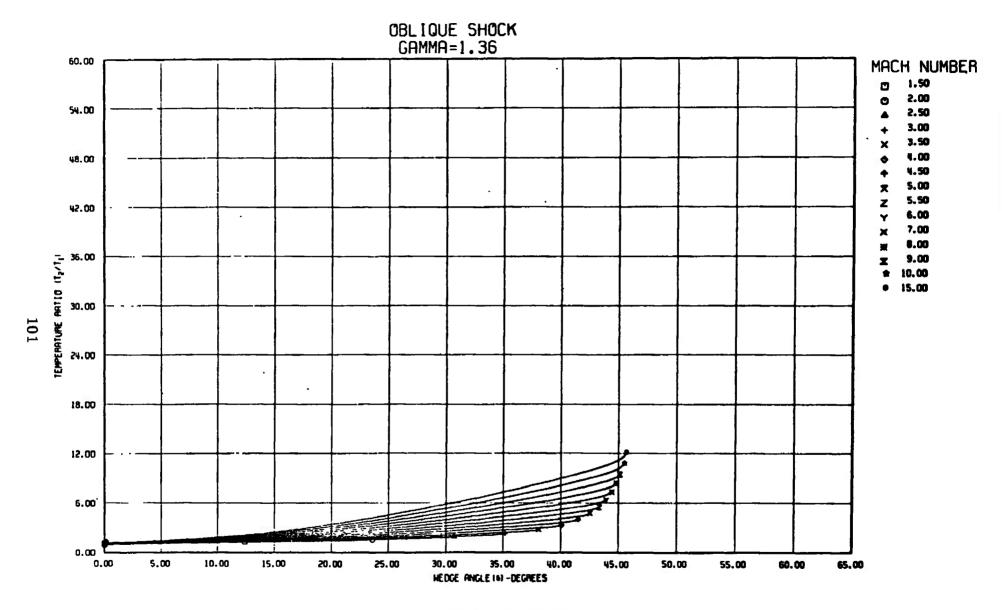


Fig. 15 Continued

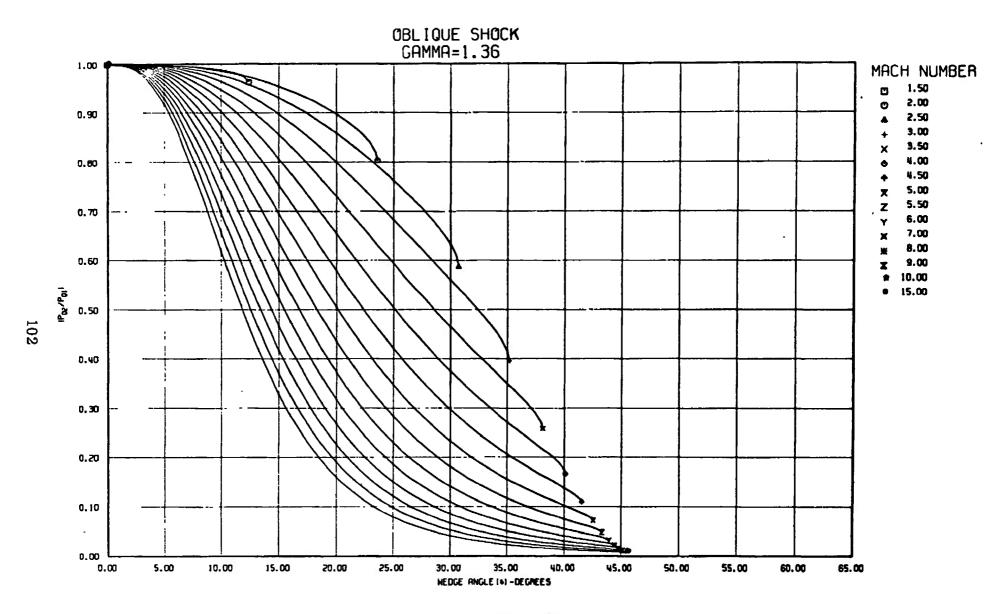


Fig. 15 Continued

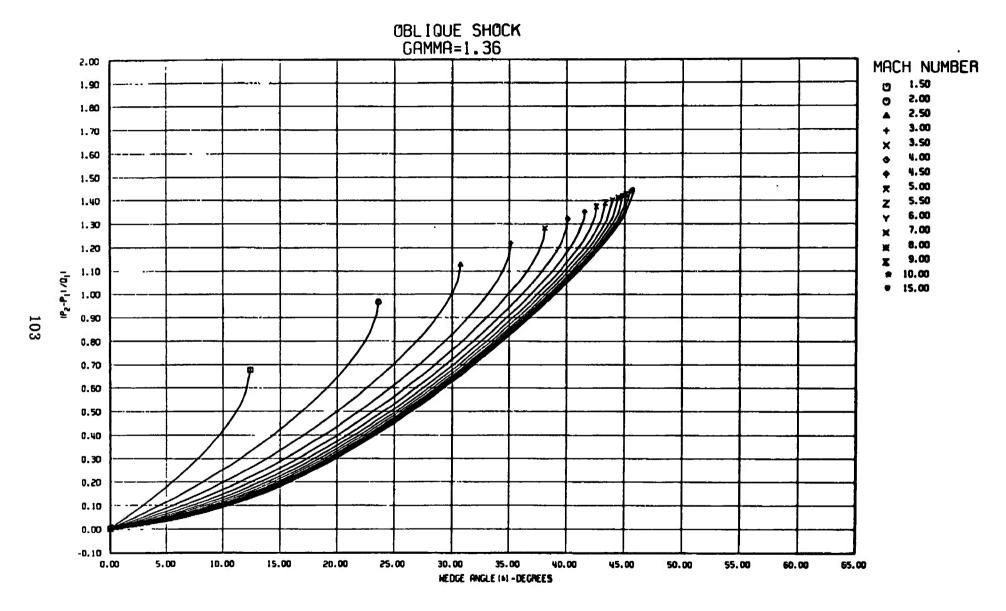


Fig. 15 Concluded

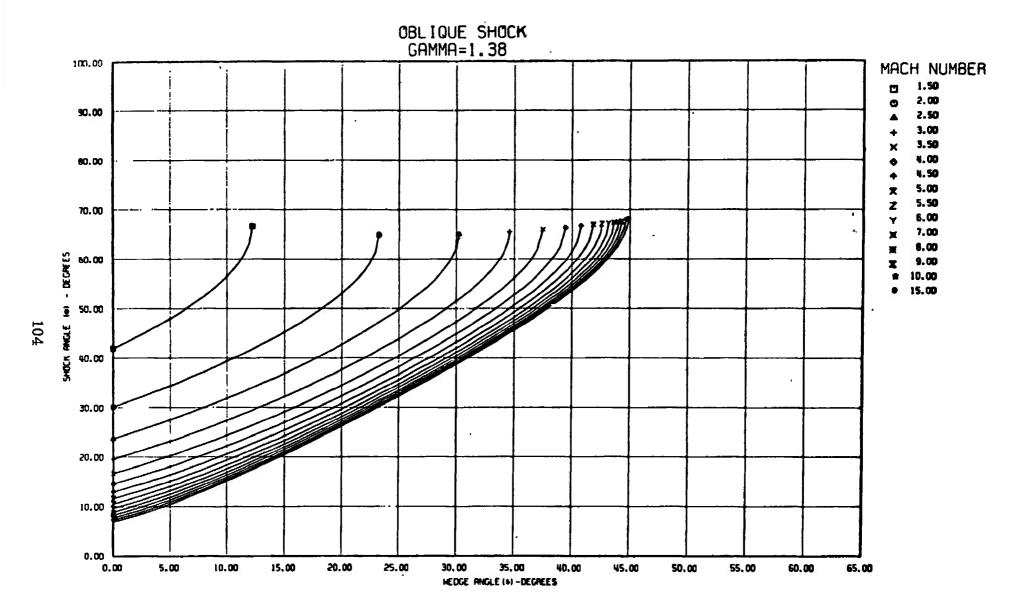


Fig. 16 $\gamma = 1.38$

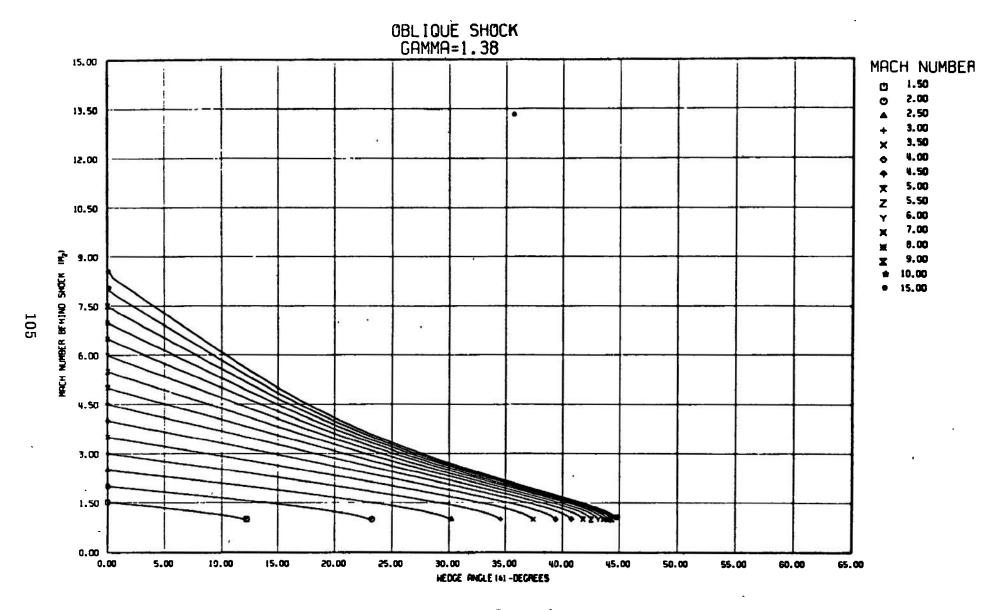


Fig. 16 Continued

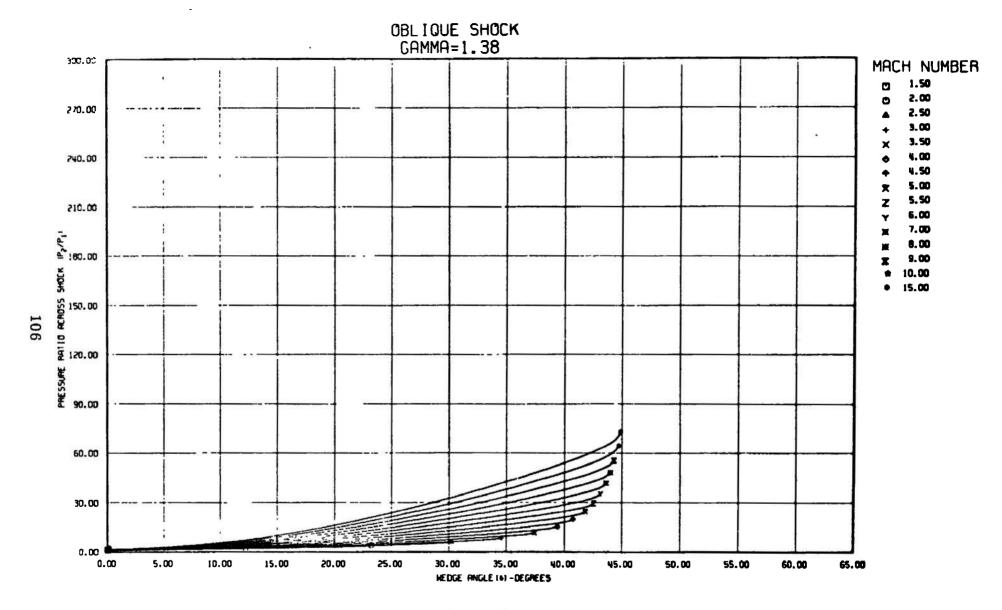


Fig. 16 Continued

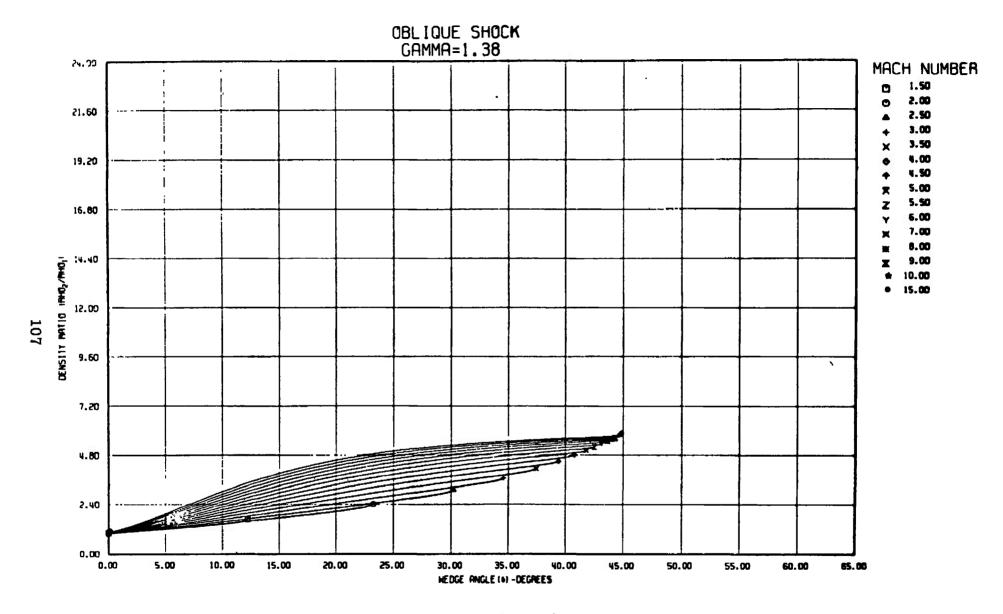


Fig. 16 Continued

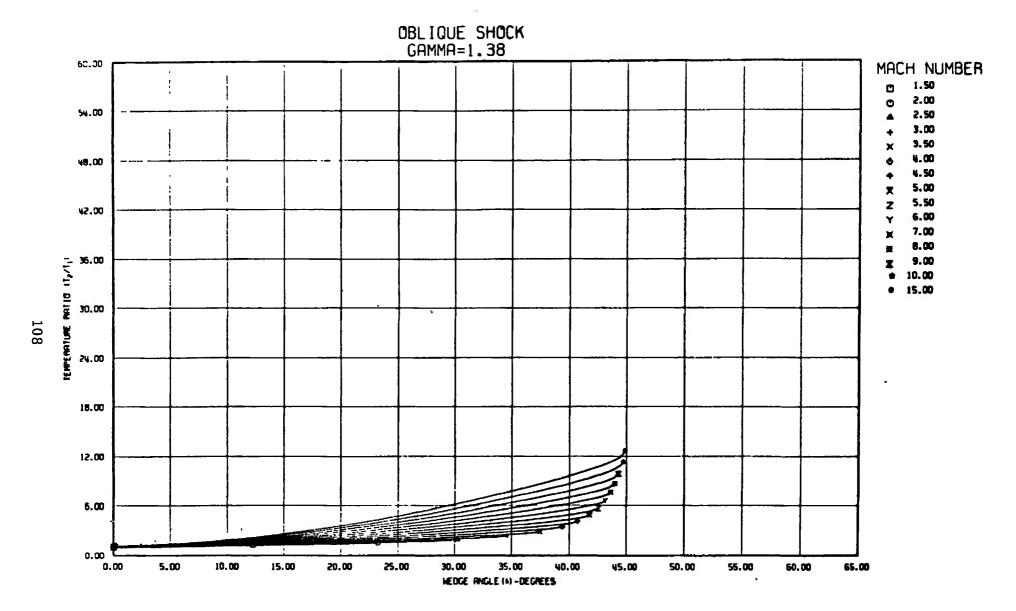


Fig. 16 Continued

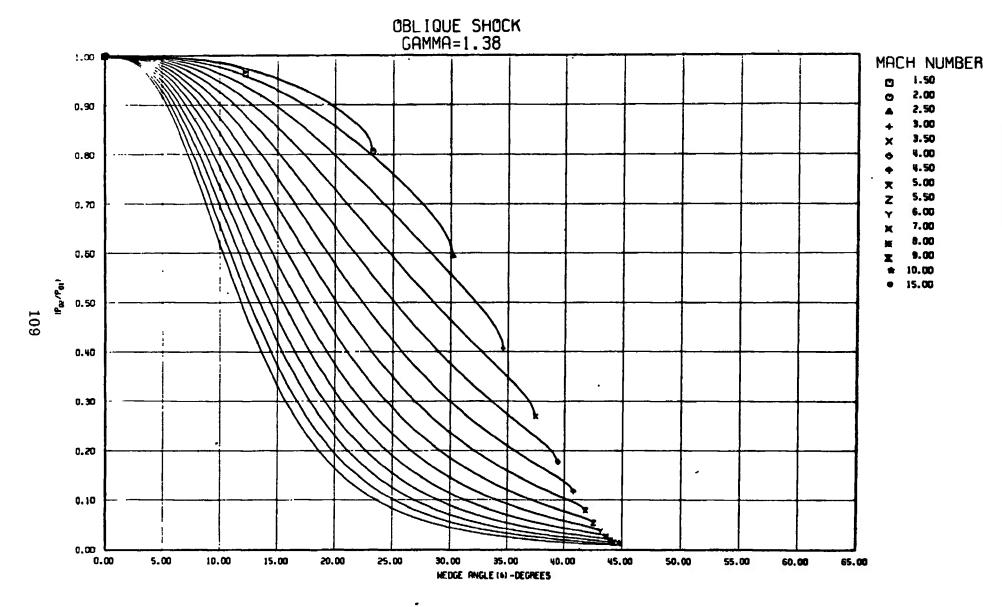


Fig. 16 Continued

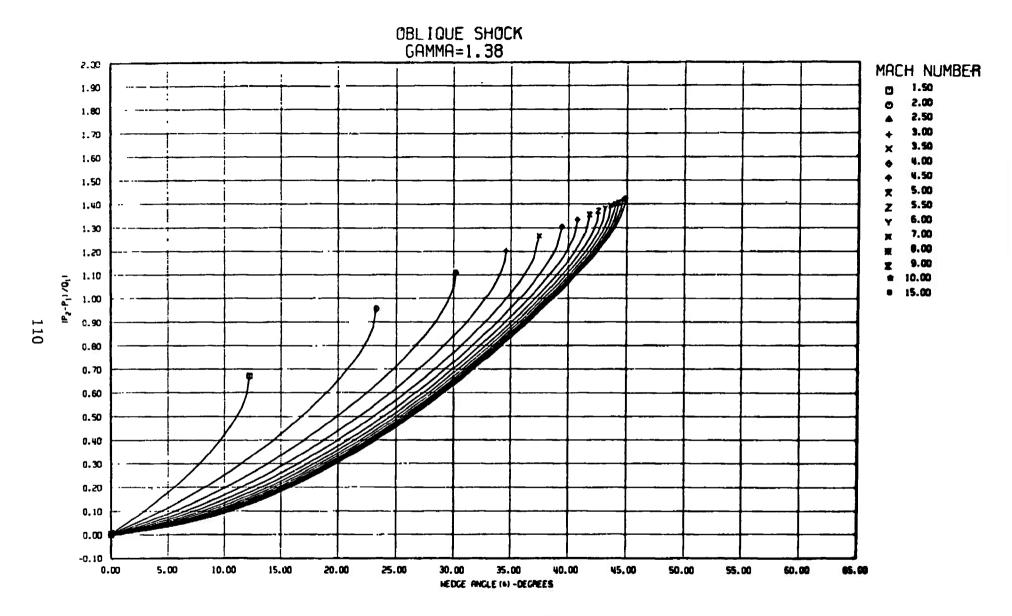


Fig. 16 Concluded

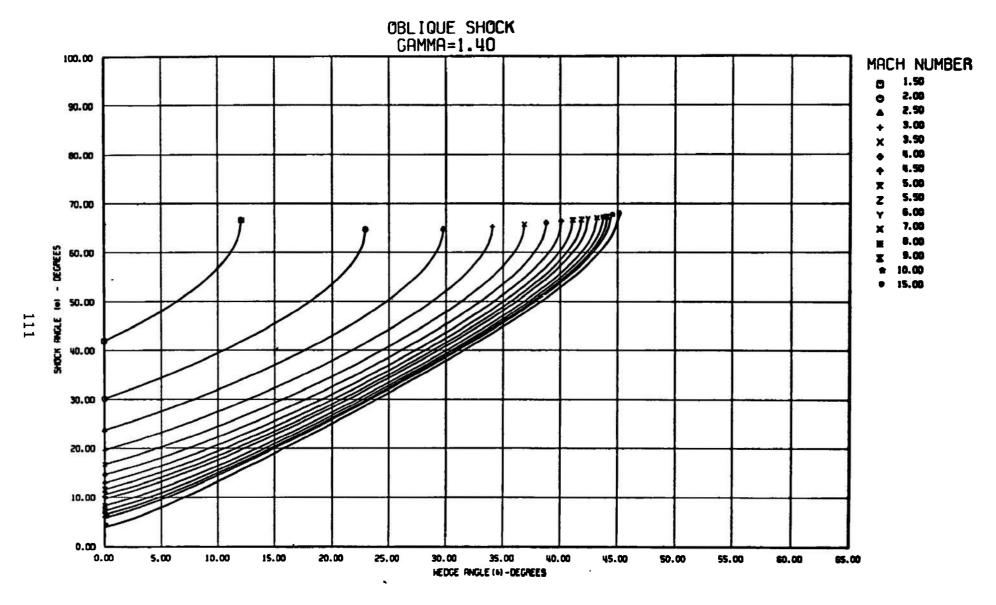


Fig. 17 y = 1.40

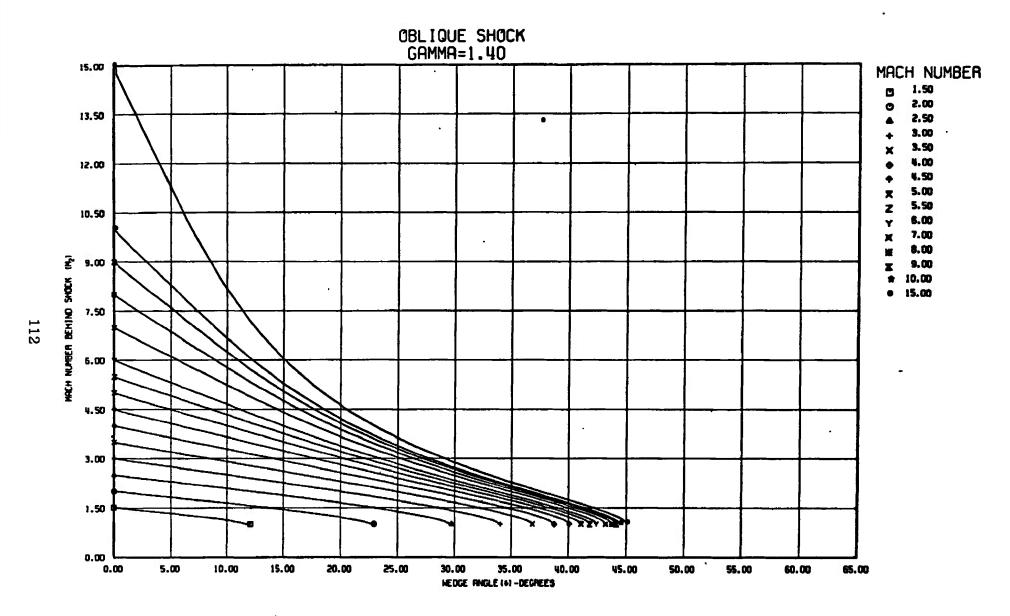


Fig. 17 Continued

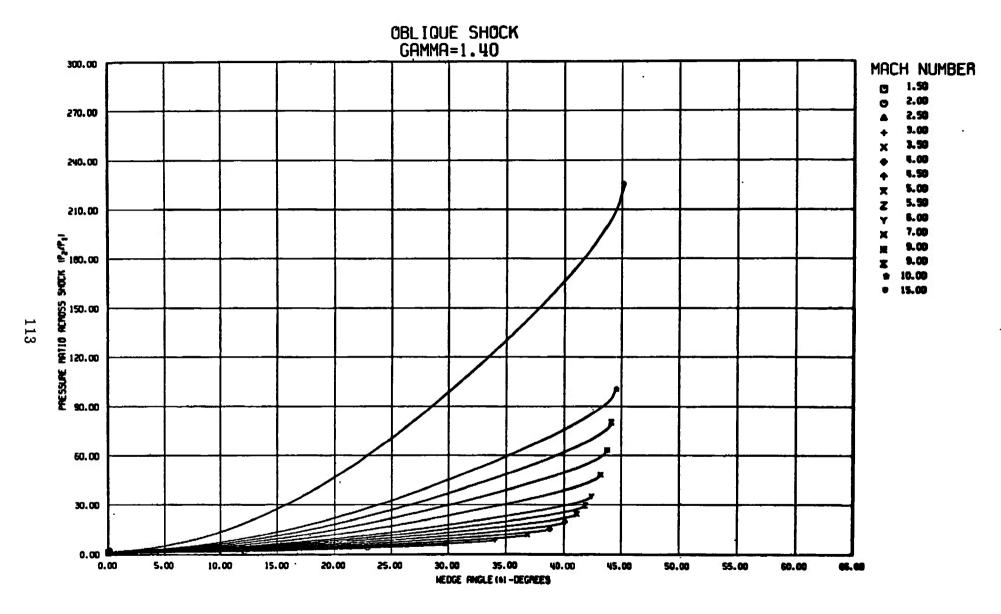


Fig. 17 Continued

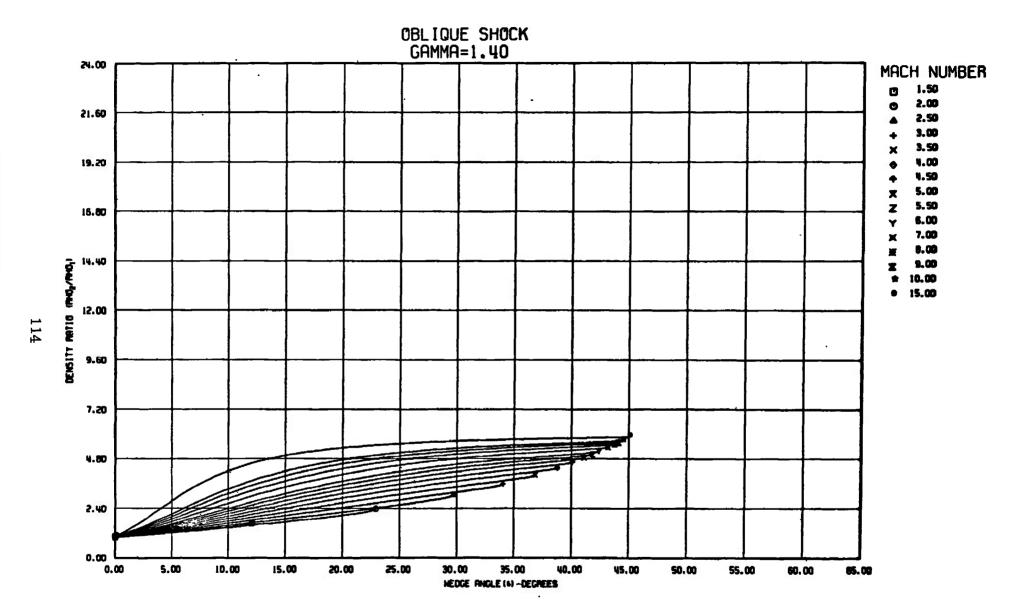


Fig. 17 Continued

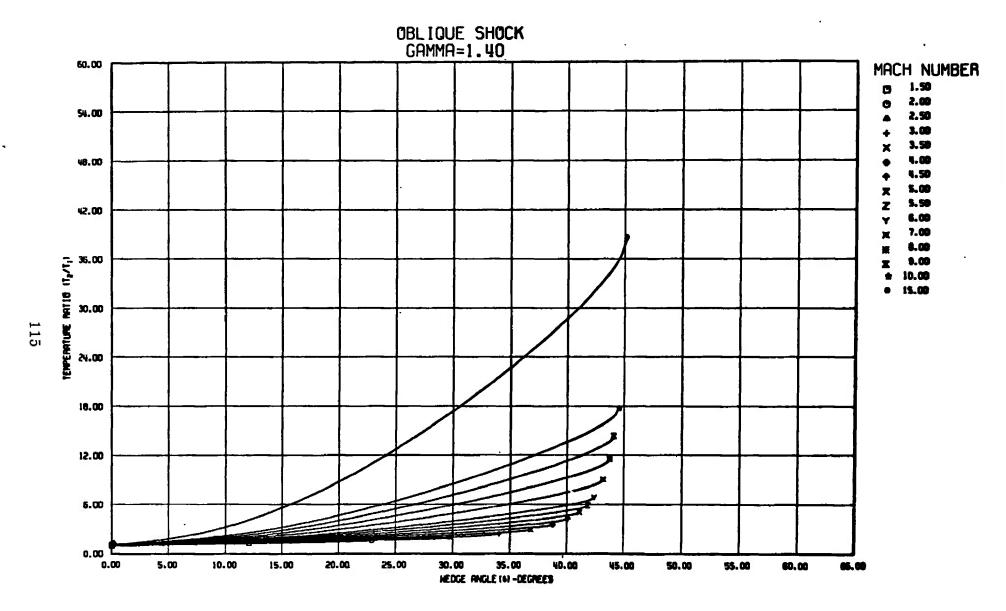


Fig. 17 Continued

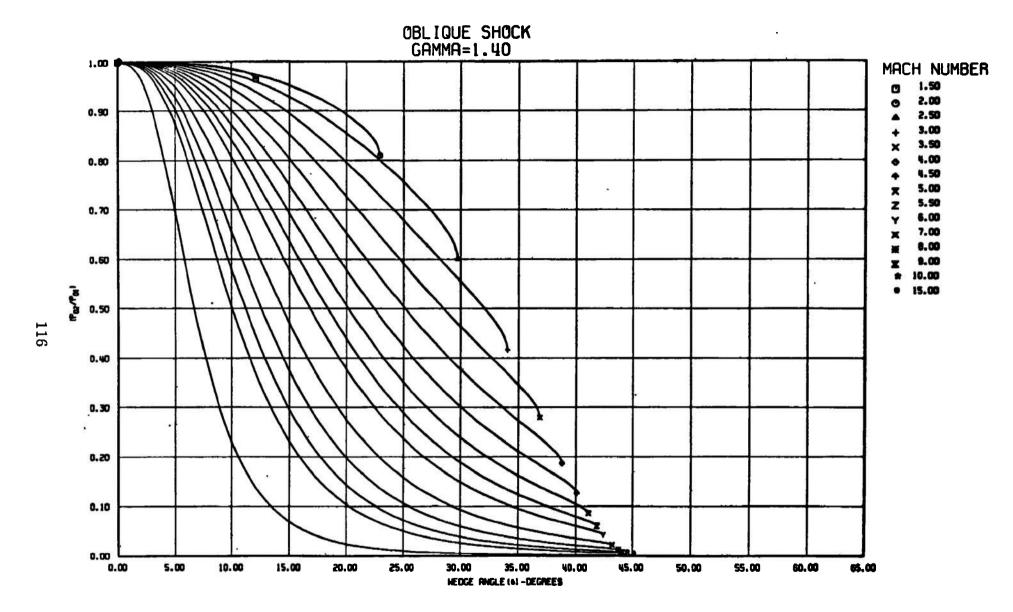


Fig. 17 Continued

Fig. 17 Concluded

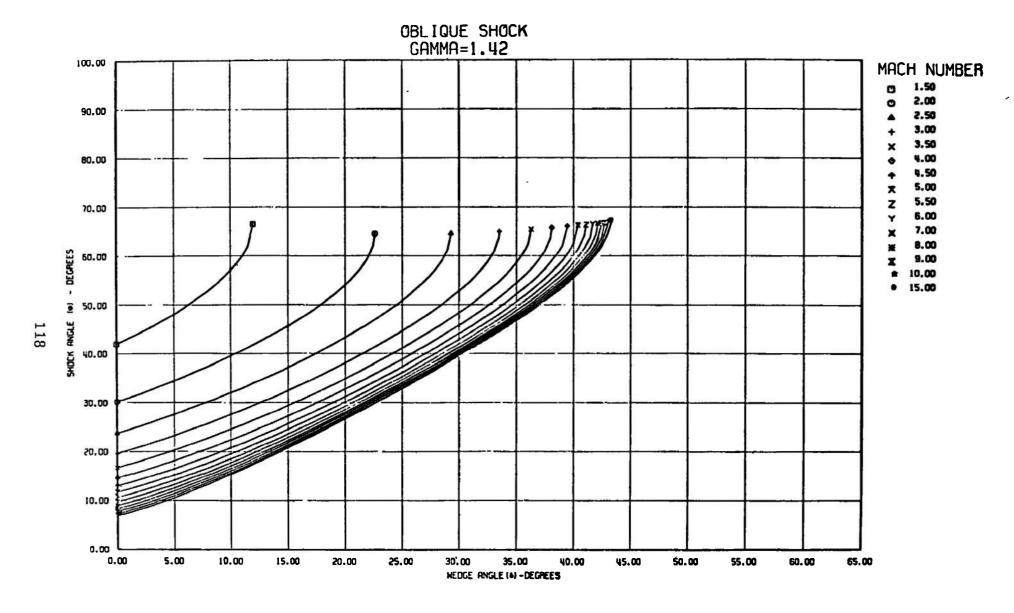


Fig. 18 y = 1.42

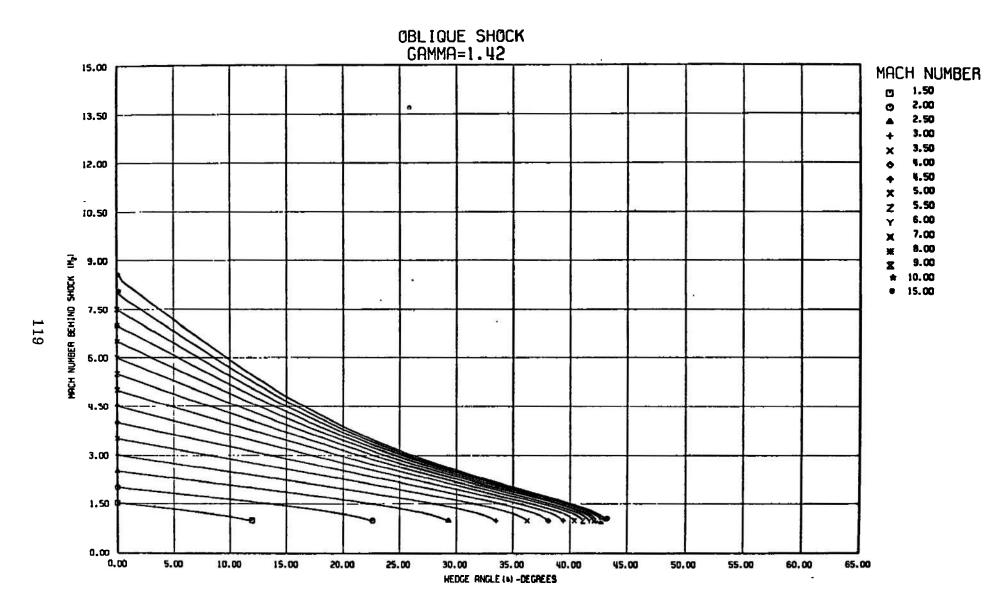


Fig. 18 Continued

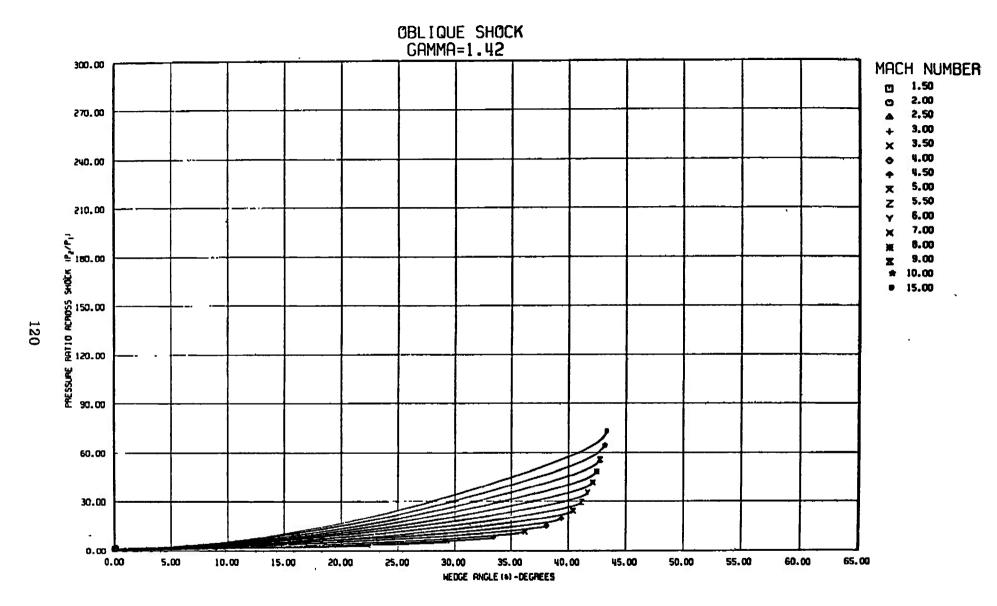


Fig. 18 Continued

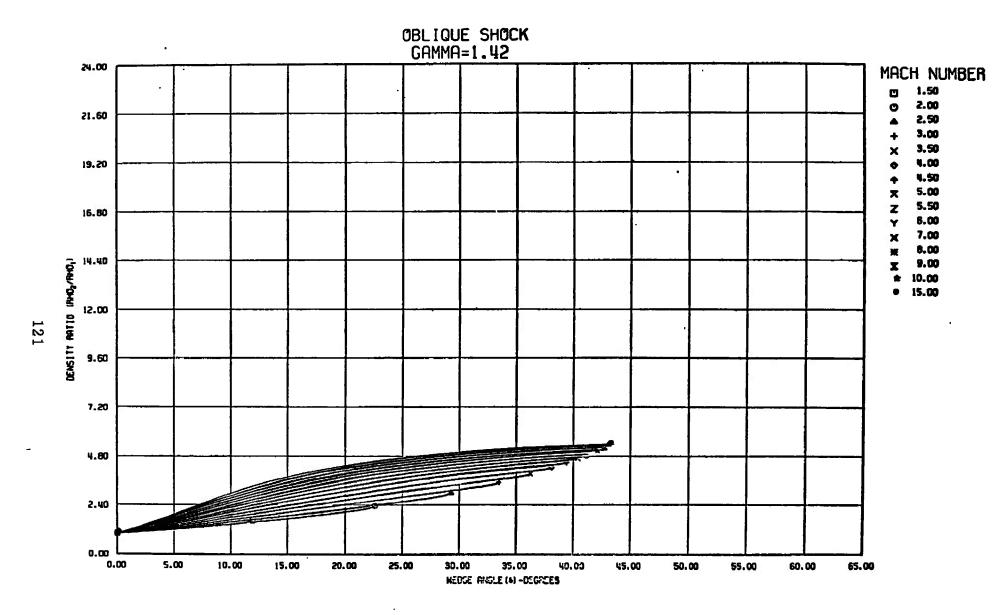


Fig. 18 Continued

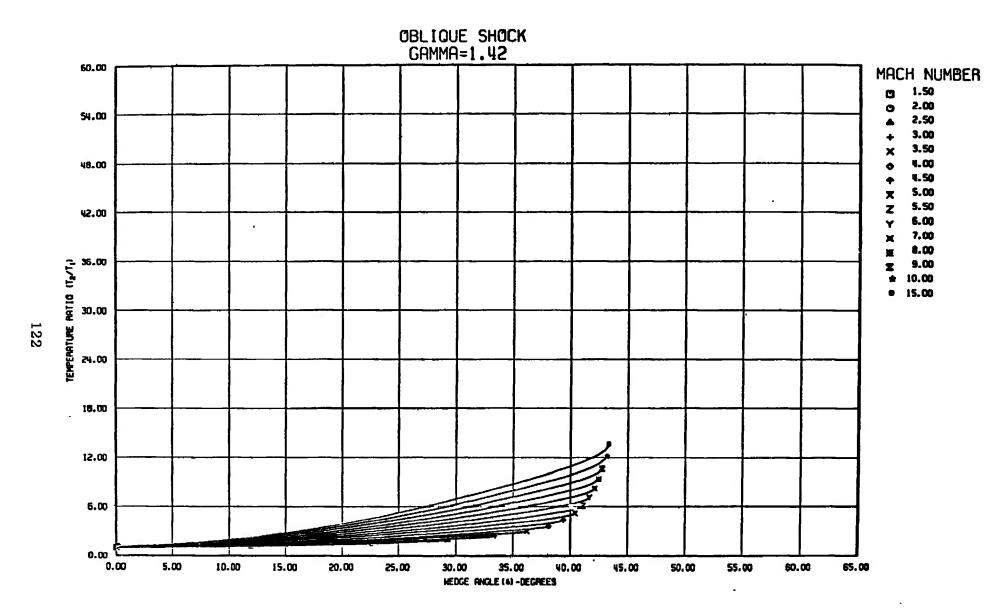


Fig. 18 Continued

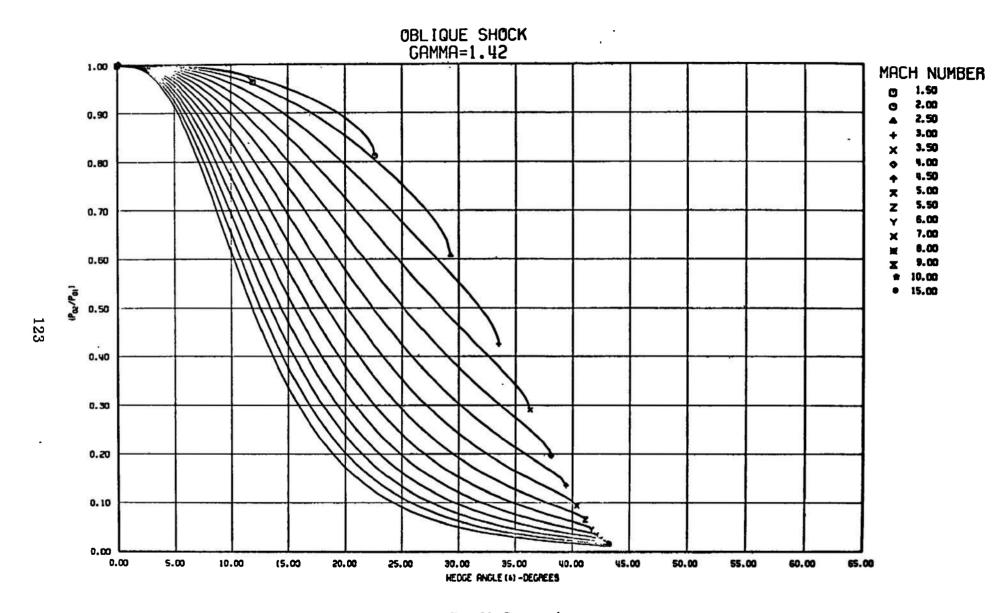


Fig. 18 Continued

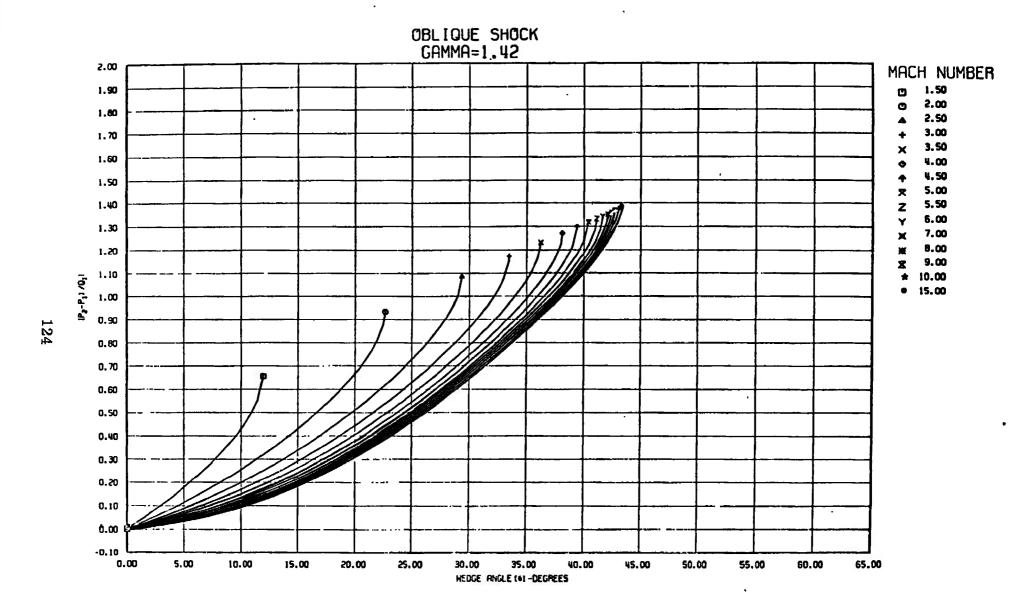


Fig. 18 Concluded

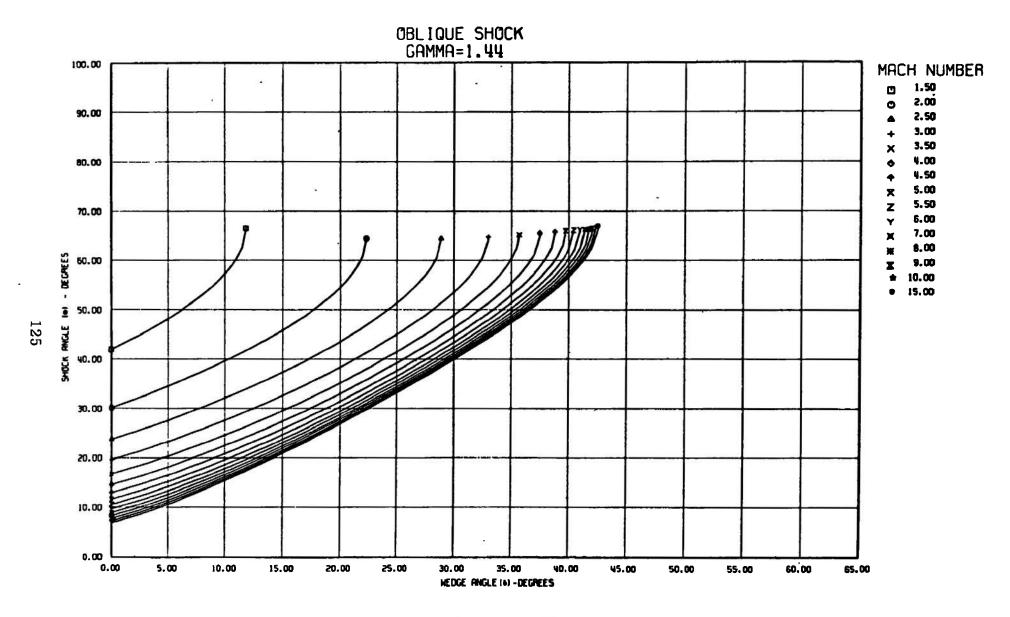


Fig. 19 y = 1.44

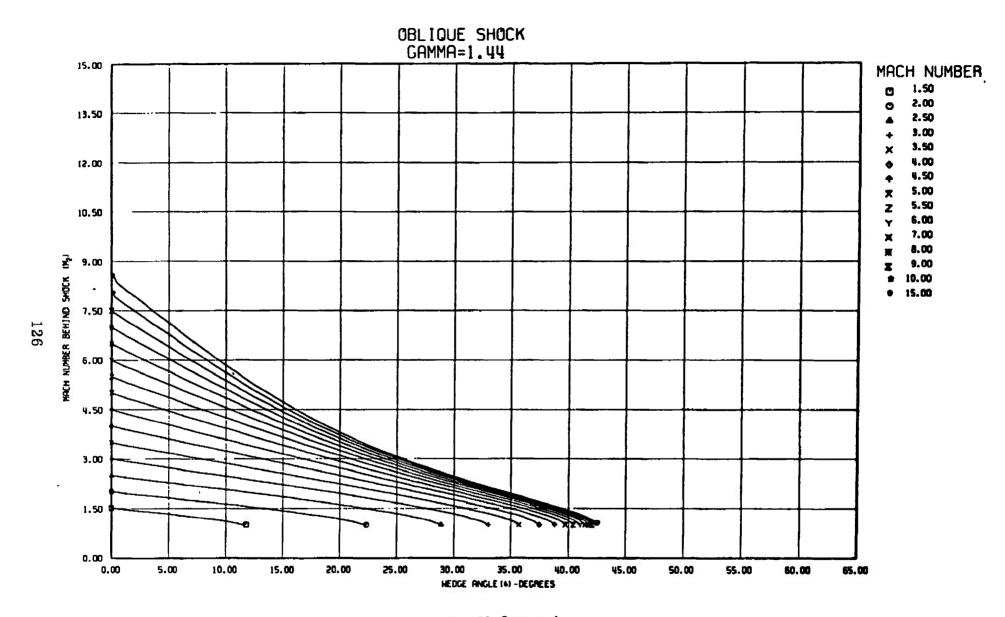


Fig. 19 Continued

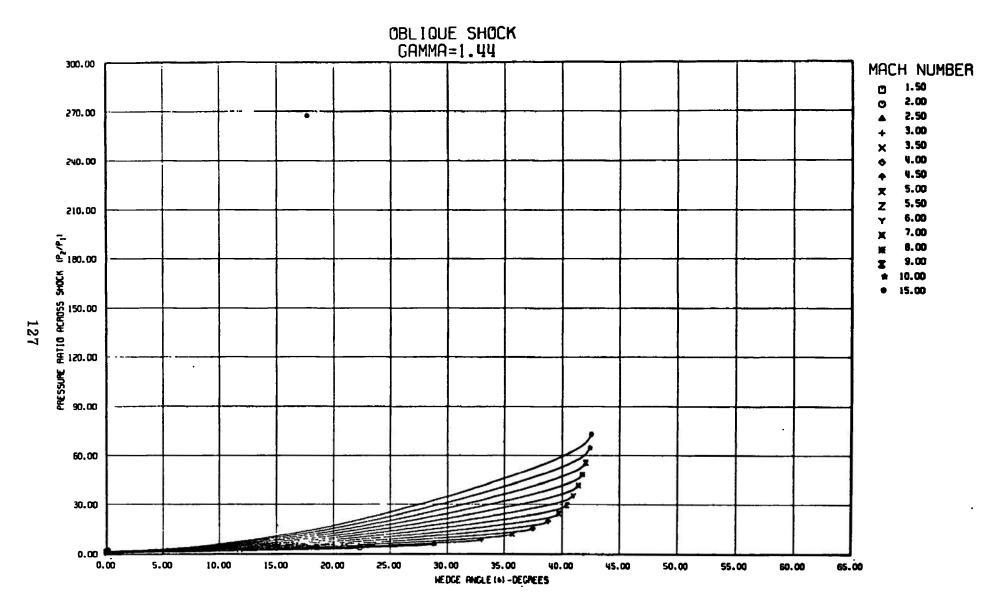


Fig. 19 Continued

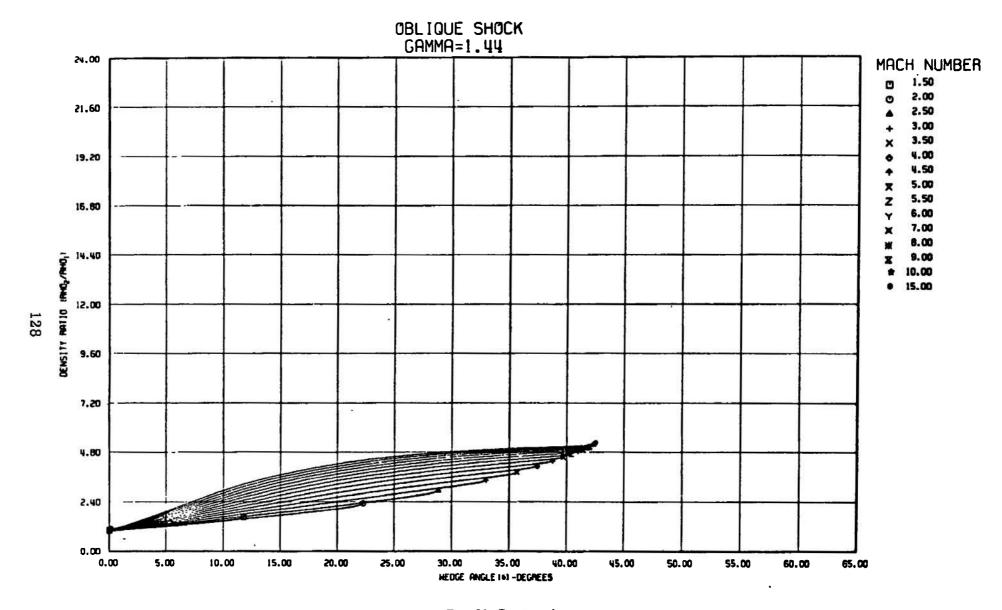


Fig. 19 Continued

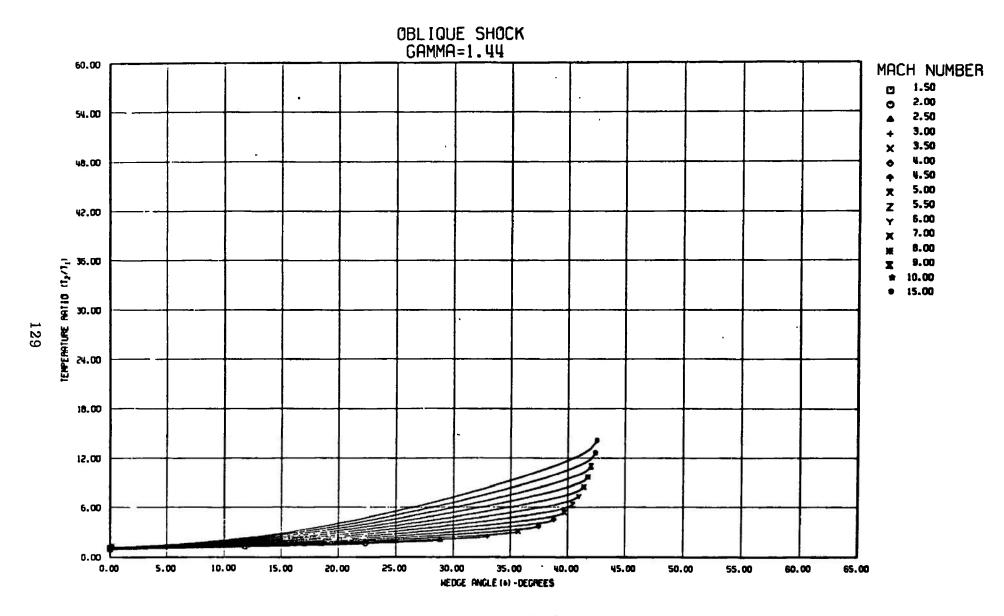


Fig. 19 Continued

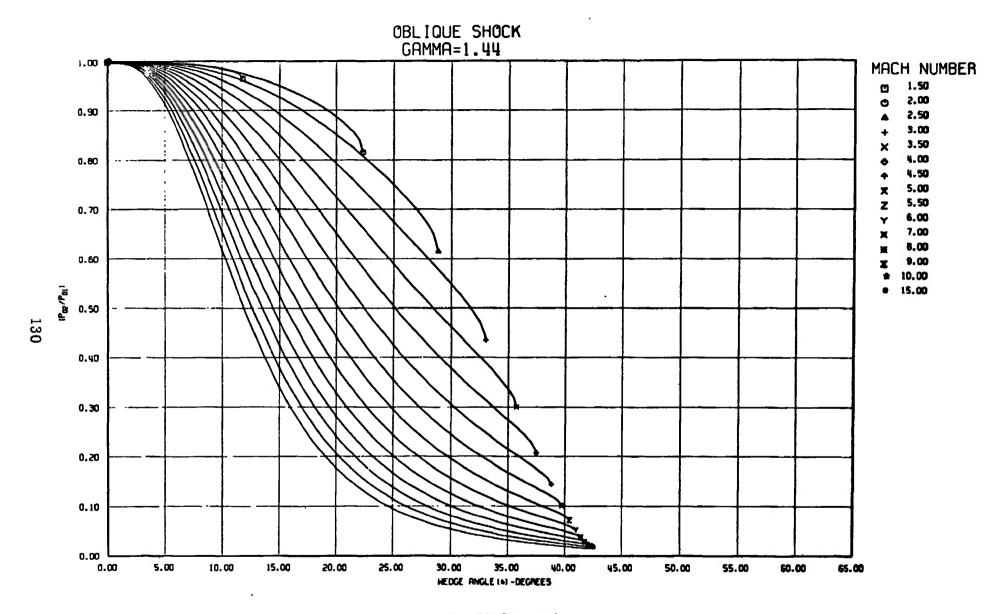


Fig. 19 Continued

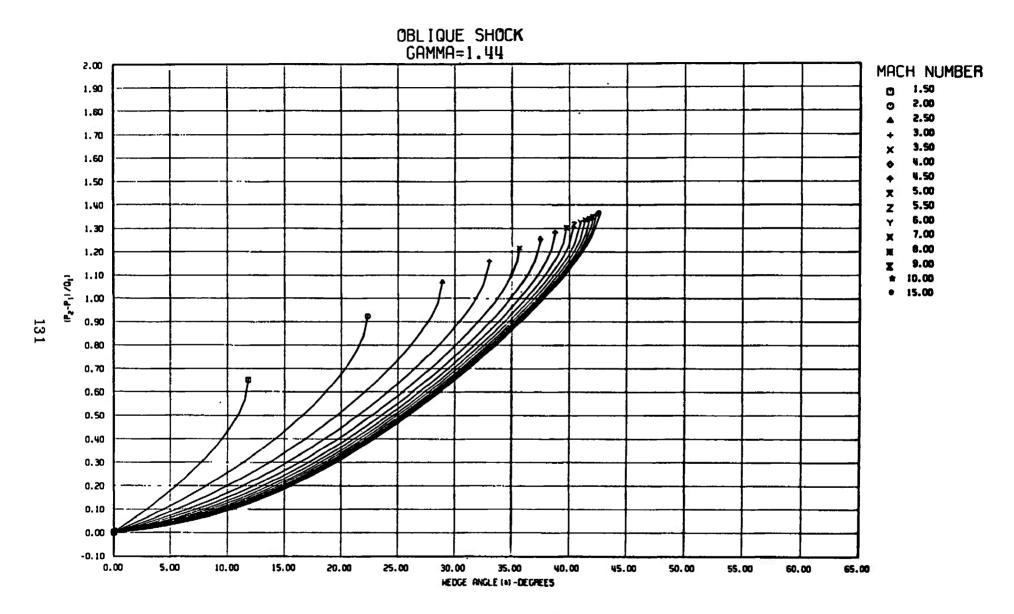


Fig. 19 Concluded

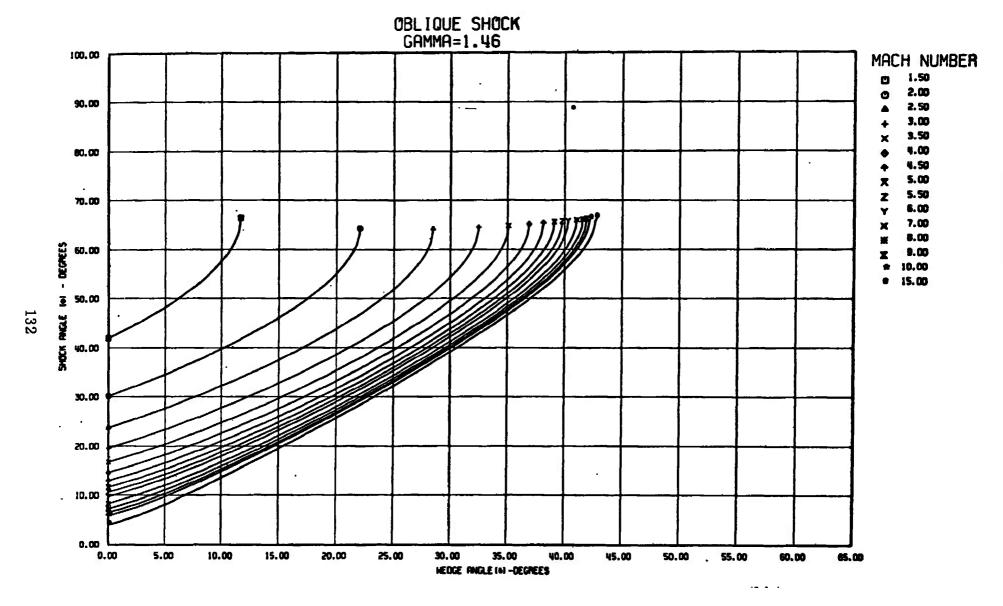


Fig. 20 $\gamma = 1.46$

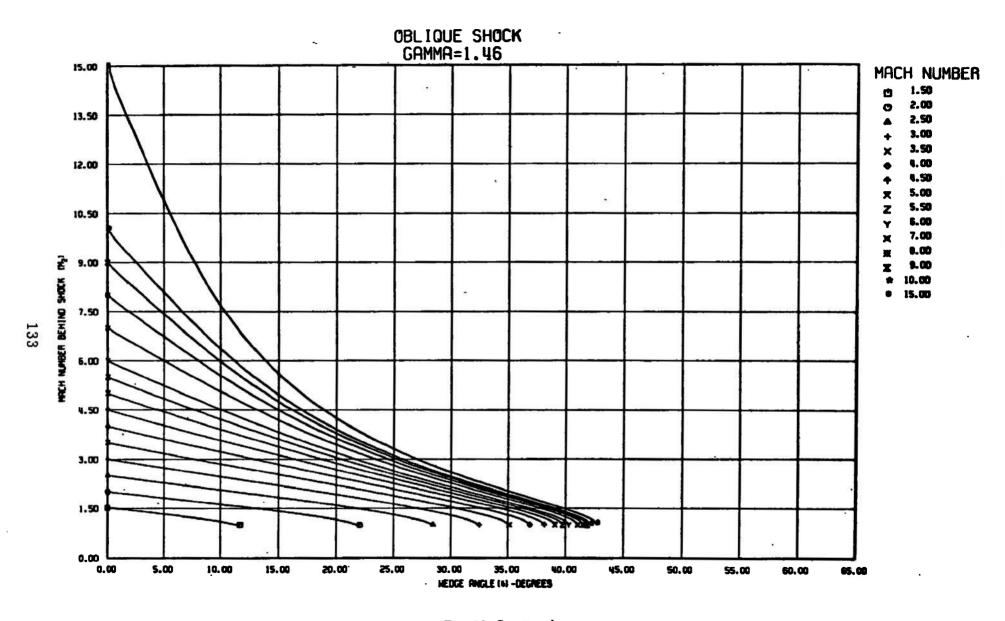


Fig. 20 Continued

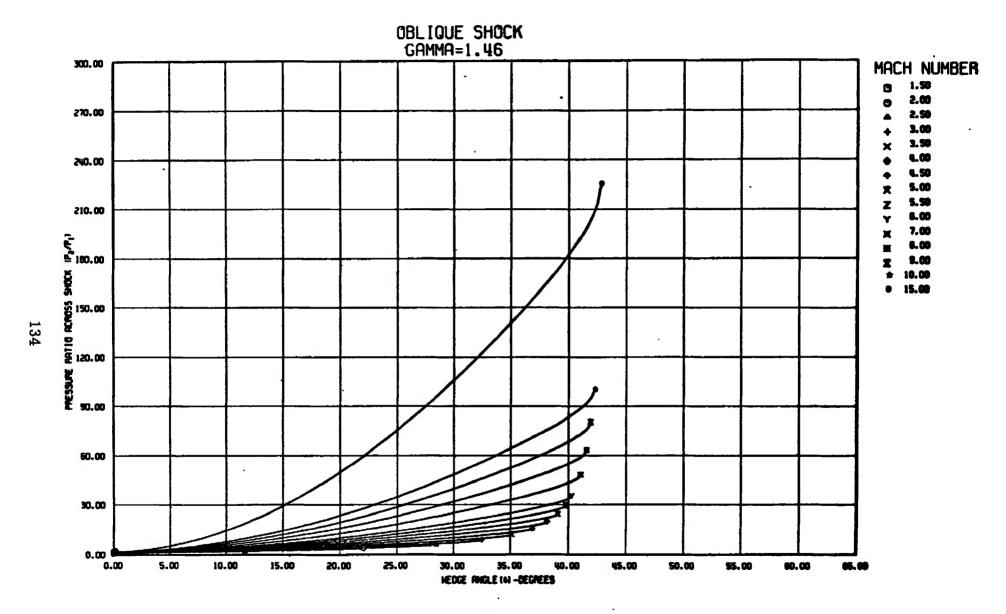


Fig. 20 Continued

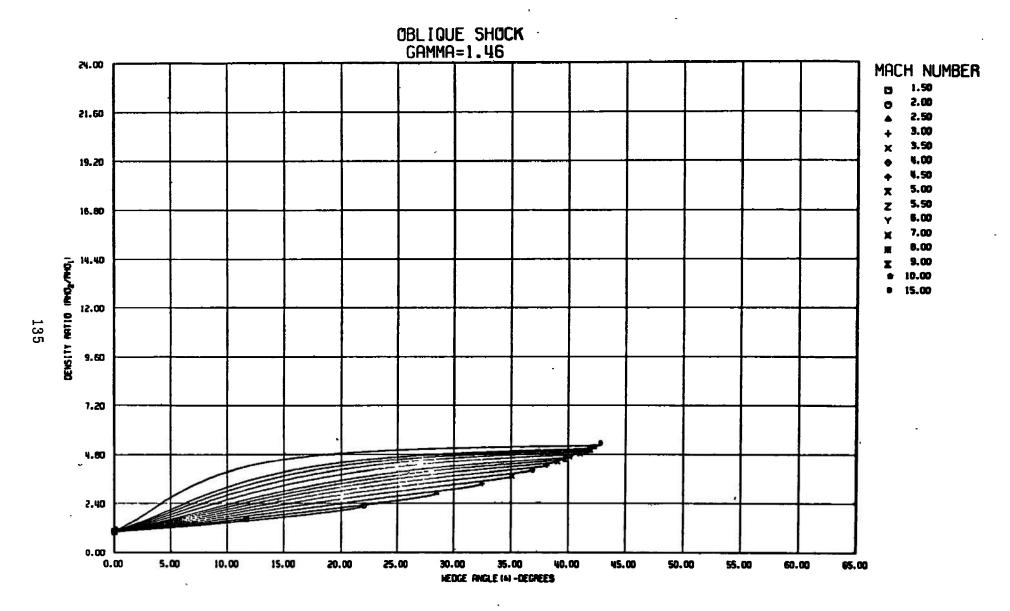


Fig. 20 Continued

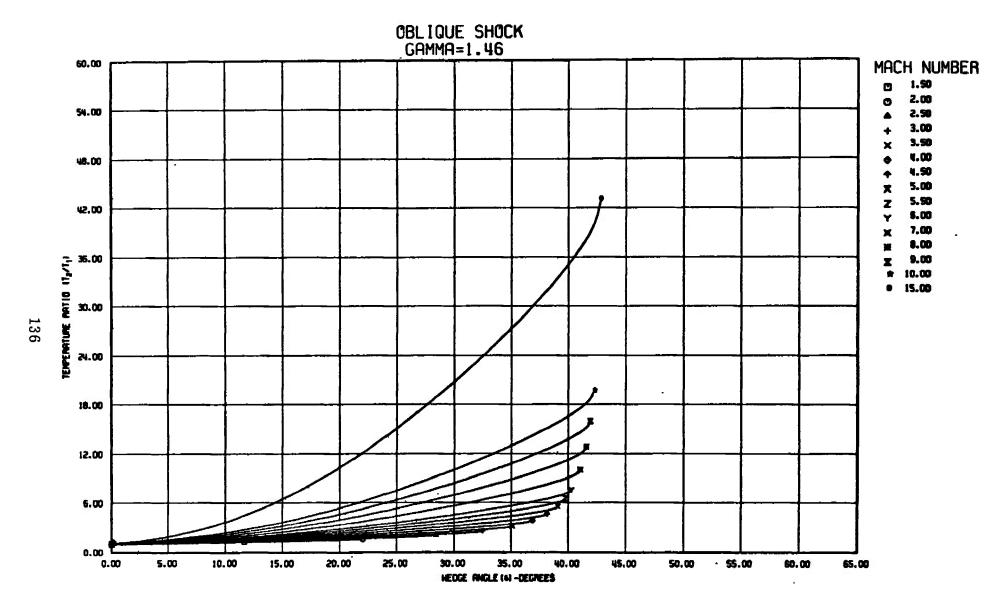


Fig. 20 Continued

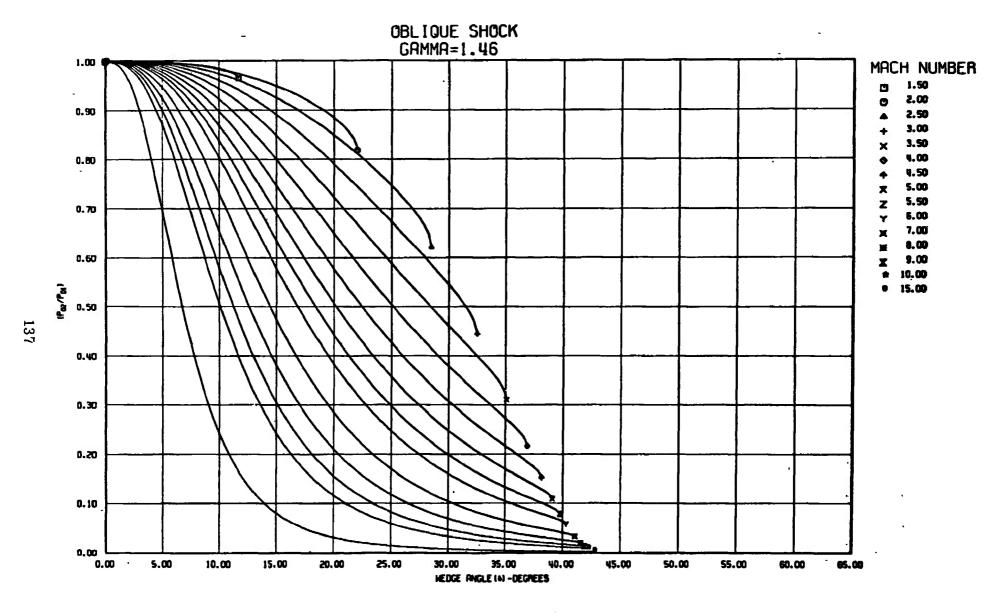


Fig. 20 Continued

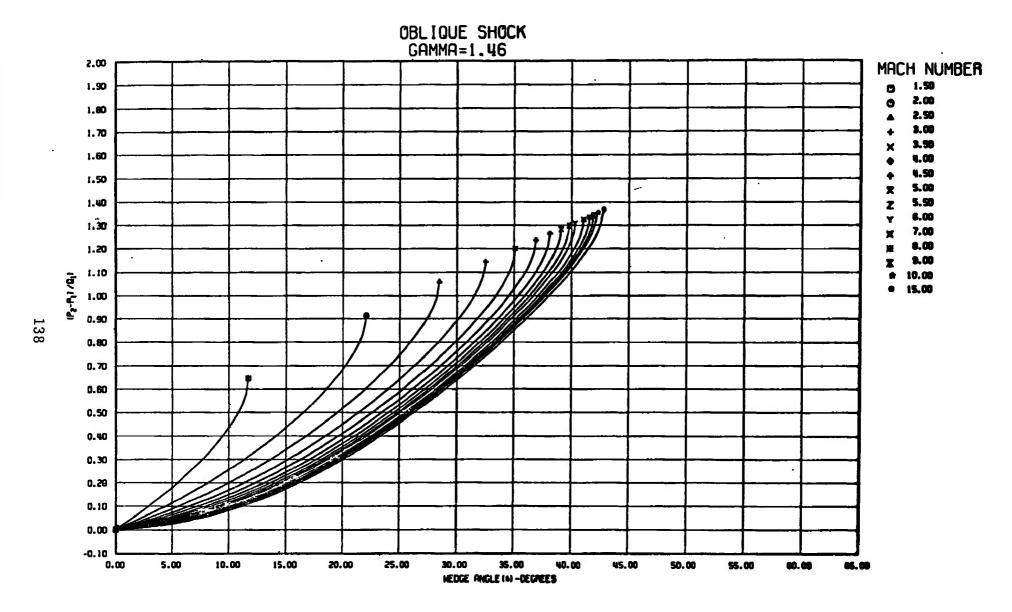


Fig. 20 Concluded

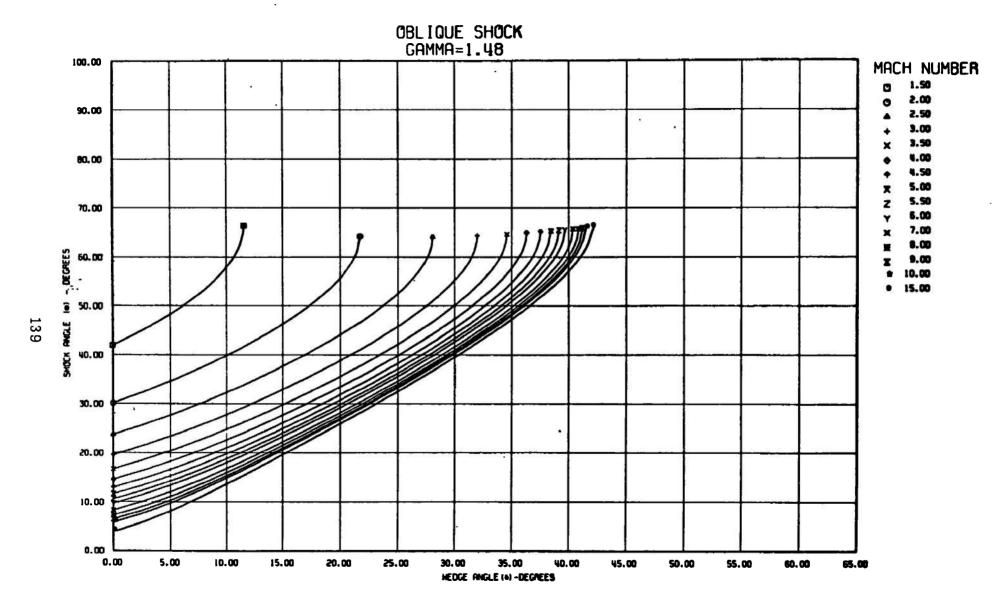


Fig. 21 y - 1.48

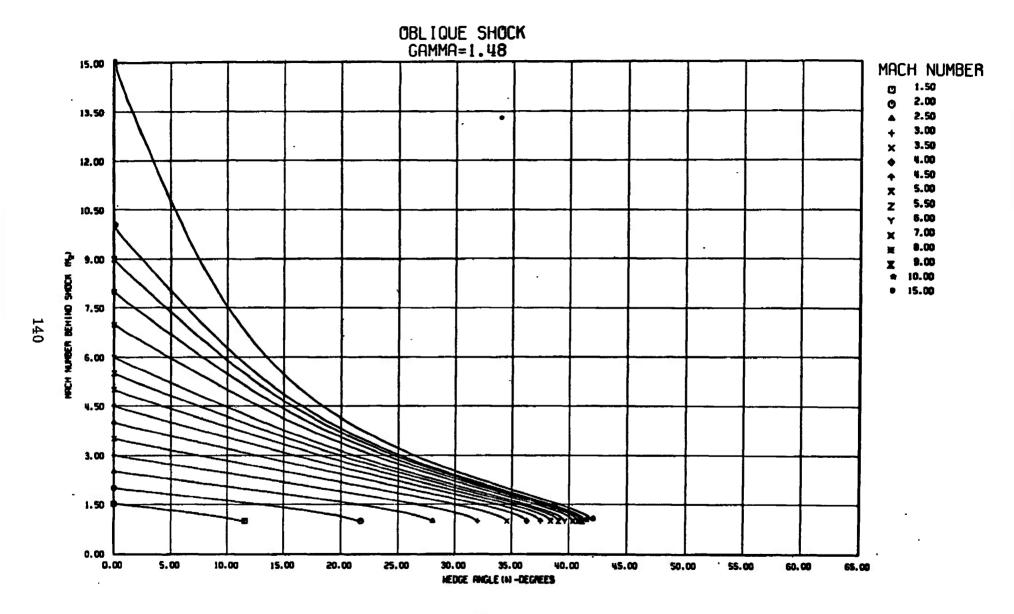


Fig. 21 Continued

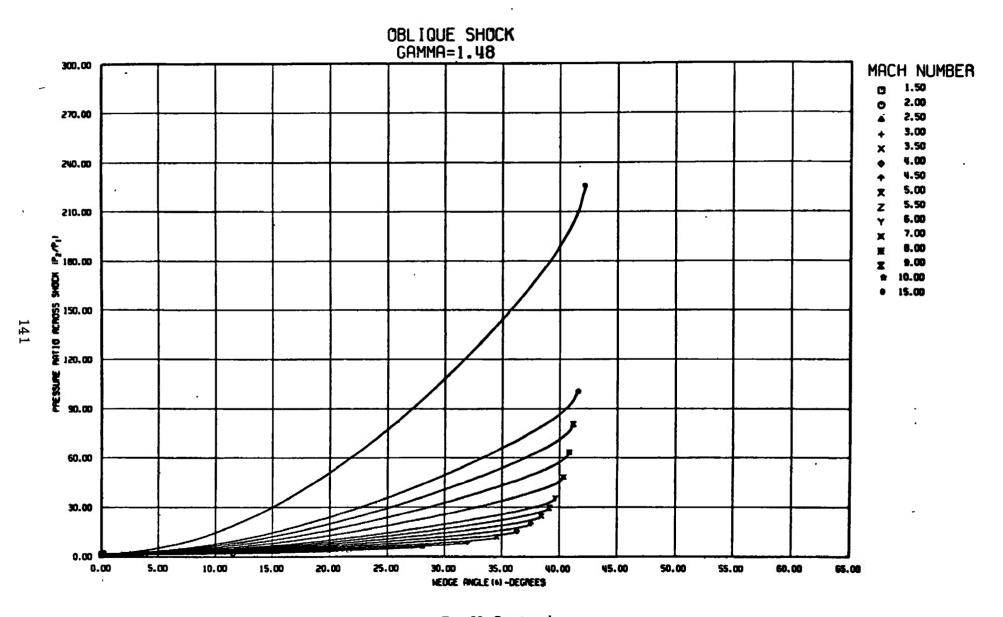


Fig. 21 Continued

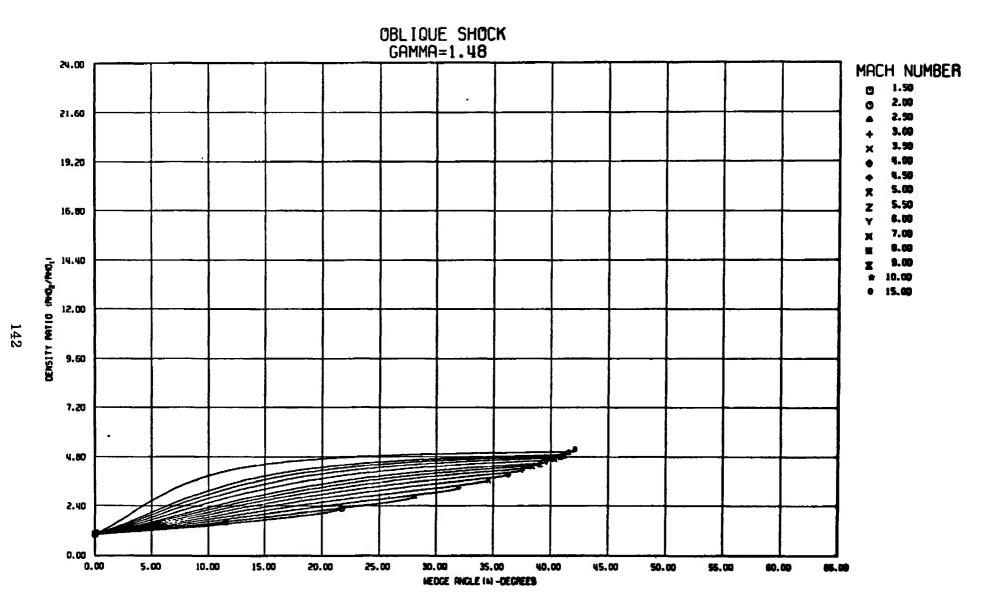


Fig. 21 Continued

Fig. 21 Continued

Fig. 21 Continued

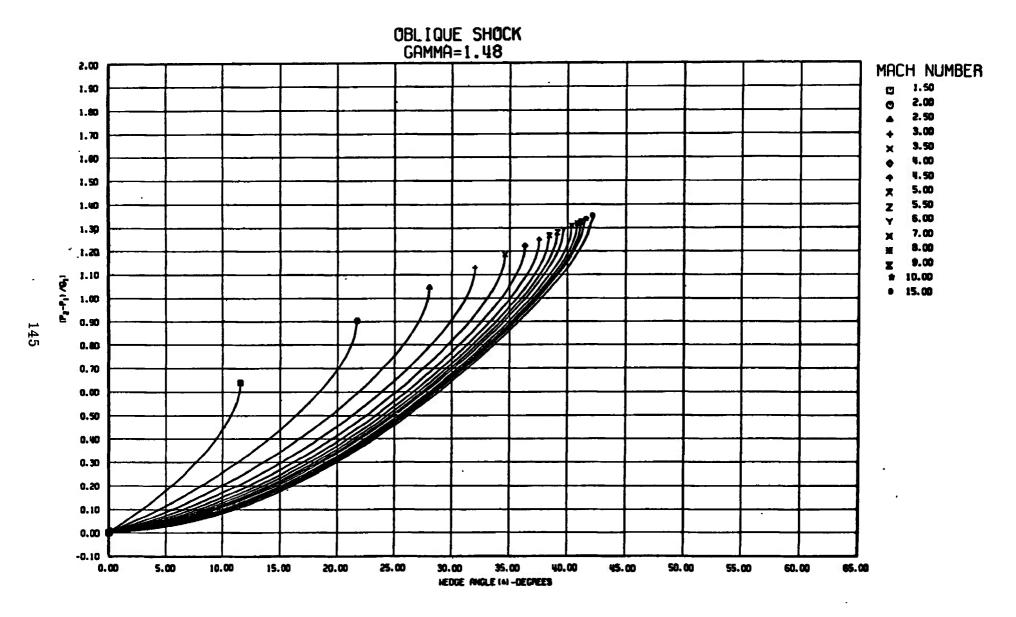


Fig. 21 Concluded

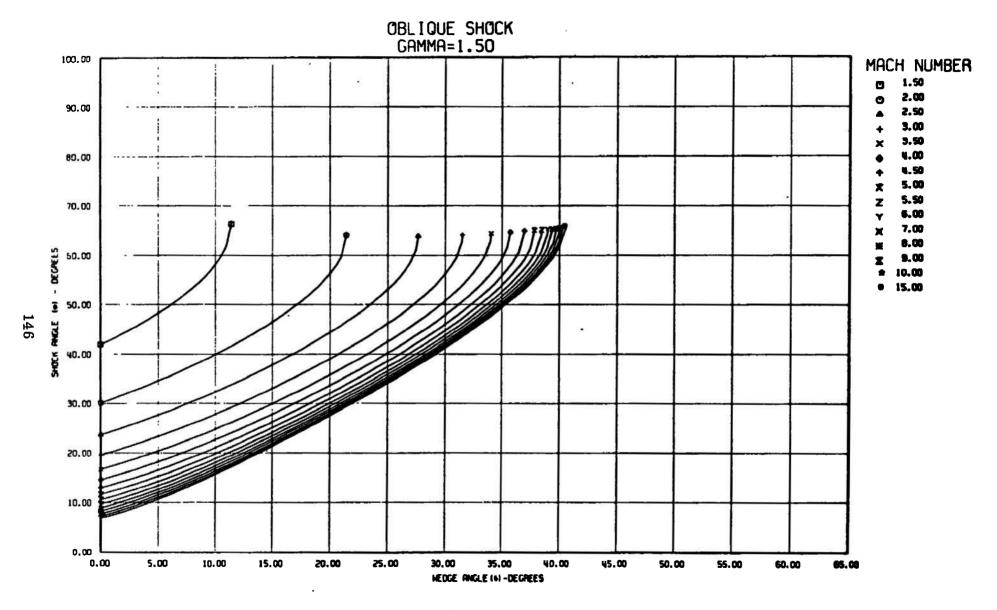


Fig. 22 $\gamma = 1.50$

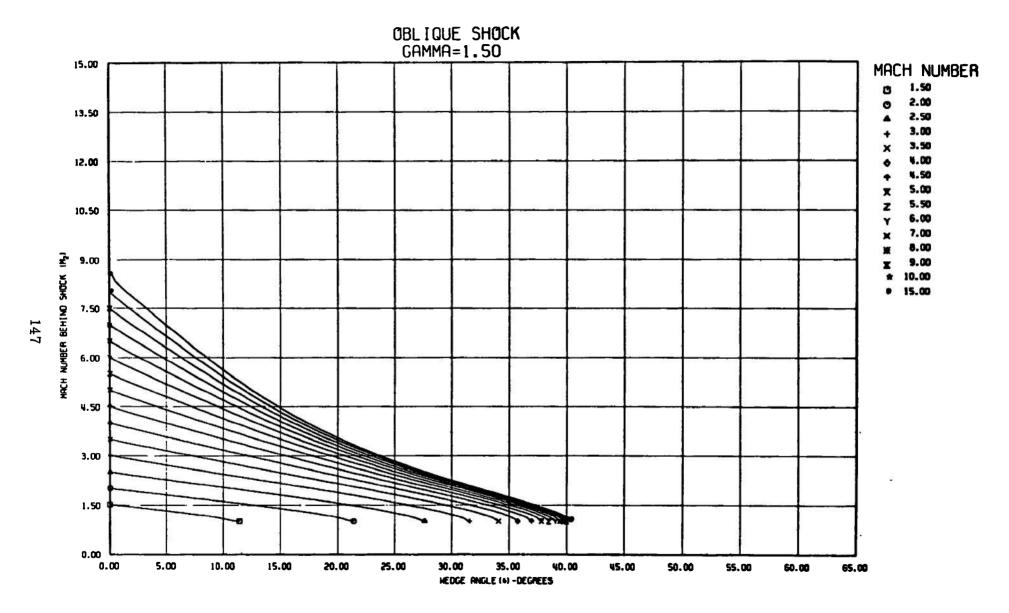


Fig. 22 Continued

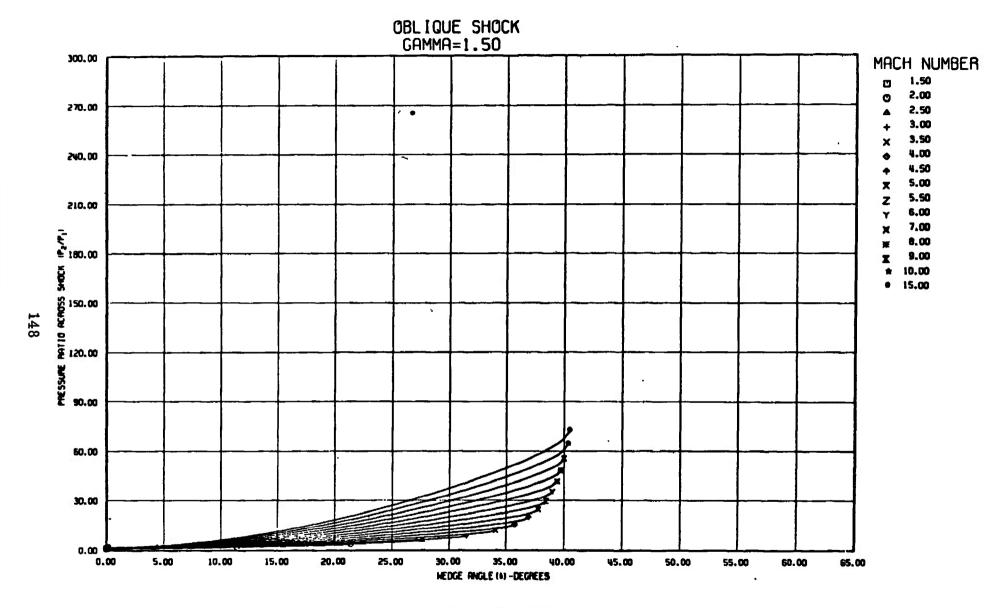


Fig. 22 Continued

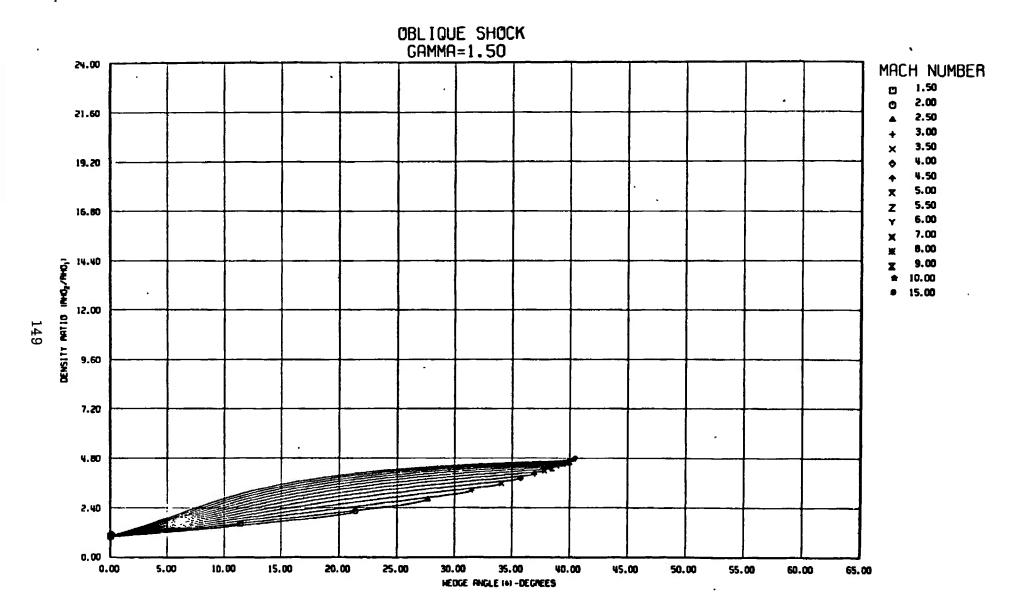


Fig. 22 Continued

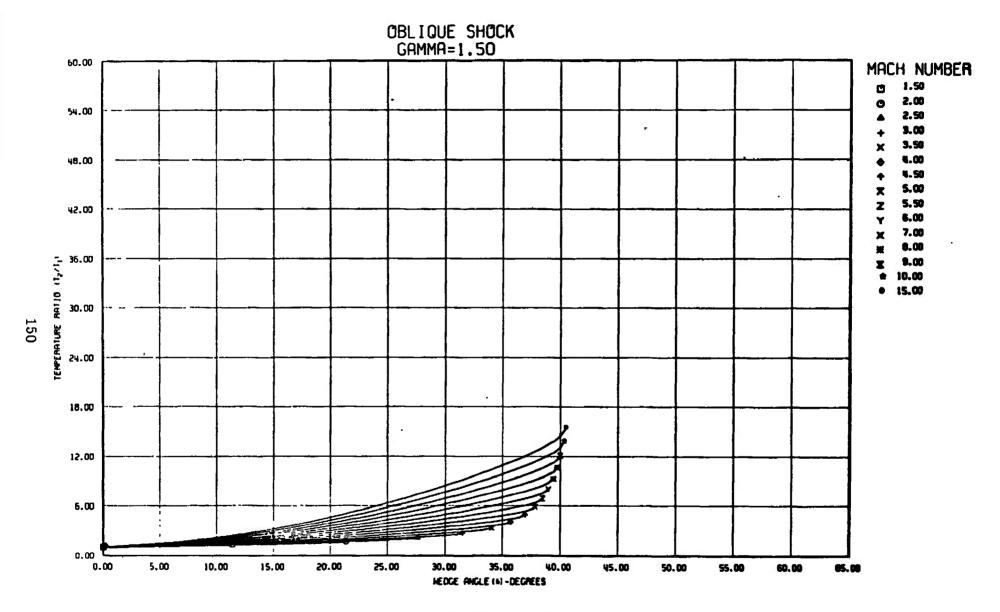


Fig. 22 Continued

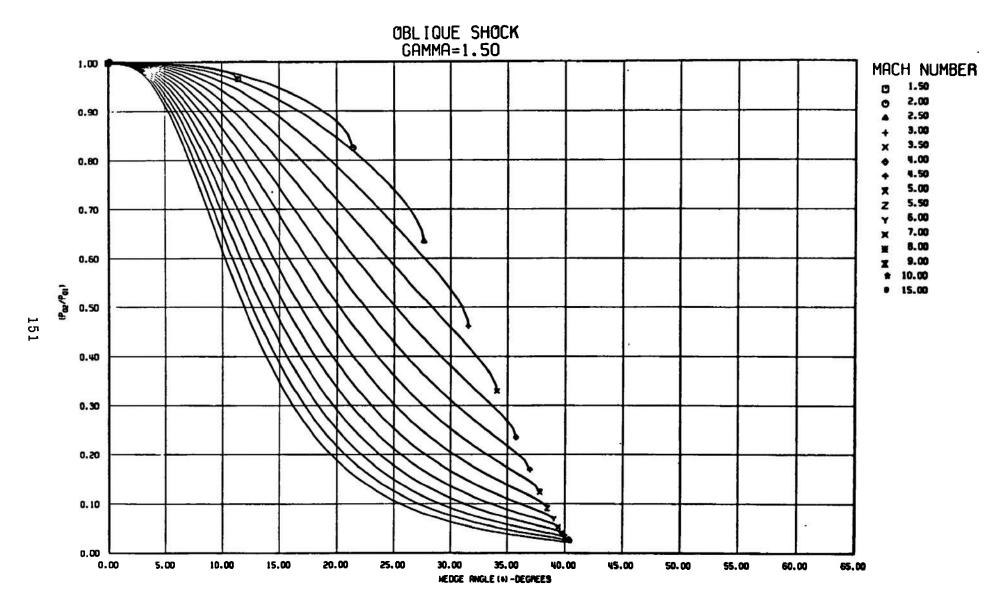


Fig. 22 Continued

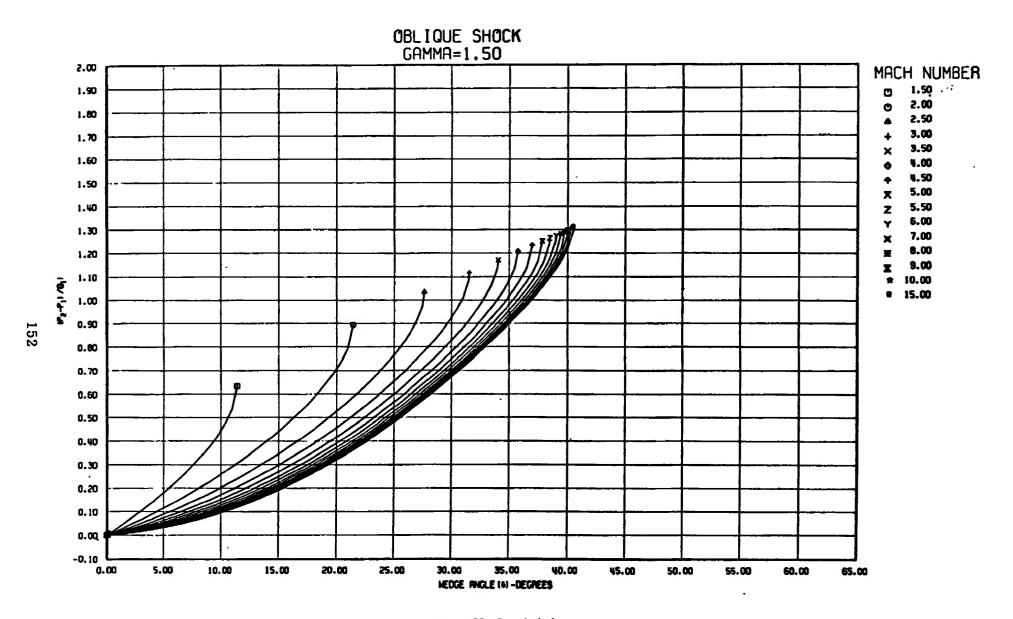


Fig. 22 Concluded

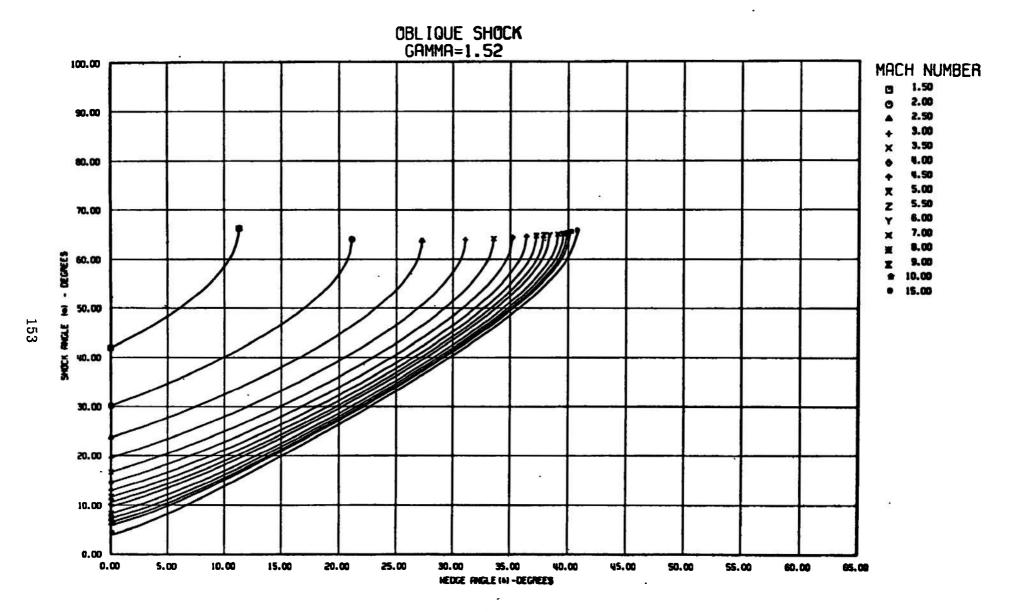


Fig. 23 y = 1.52

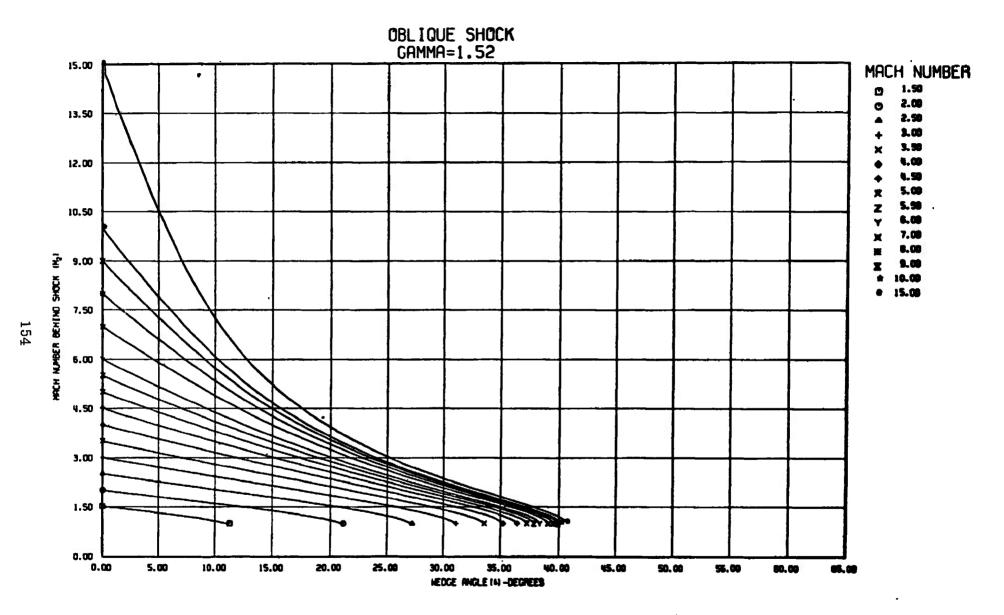


Fig. 23 Continued

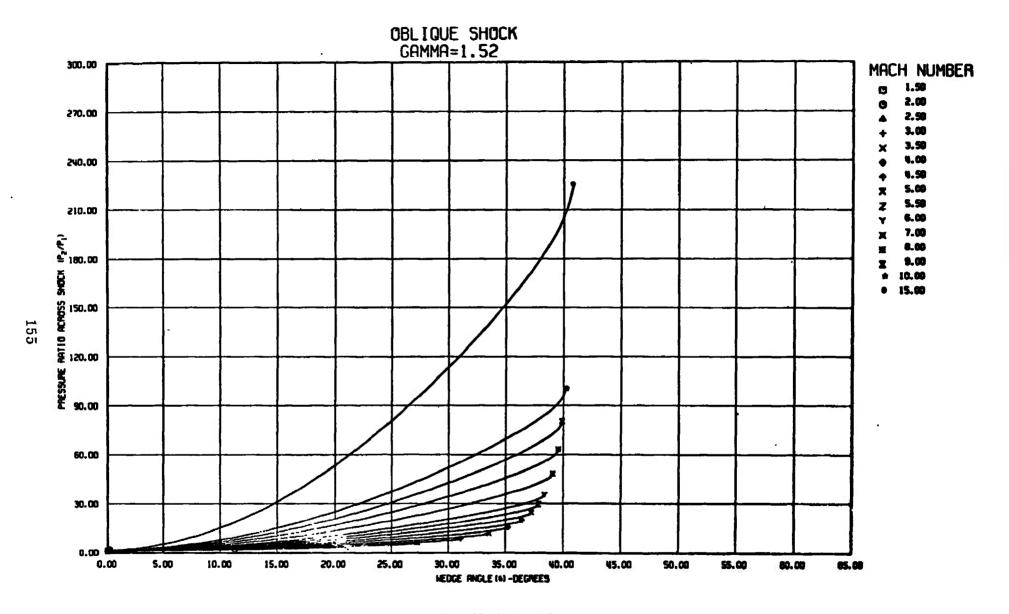


Fig. 23 Continued

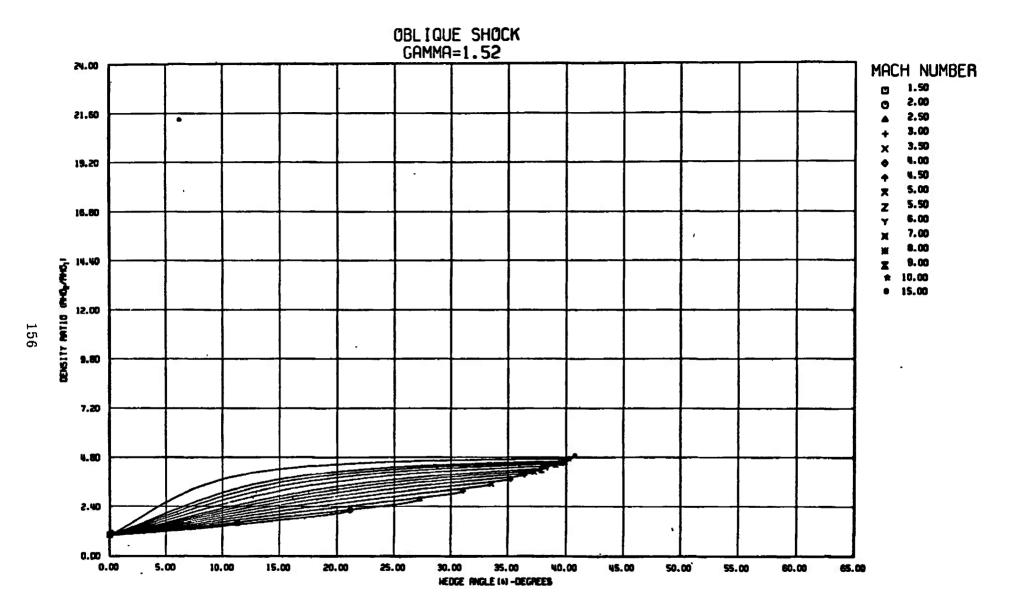


Fig. 23 Continued

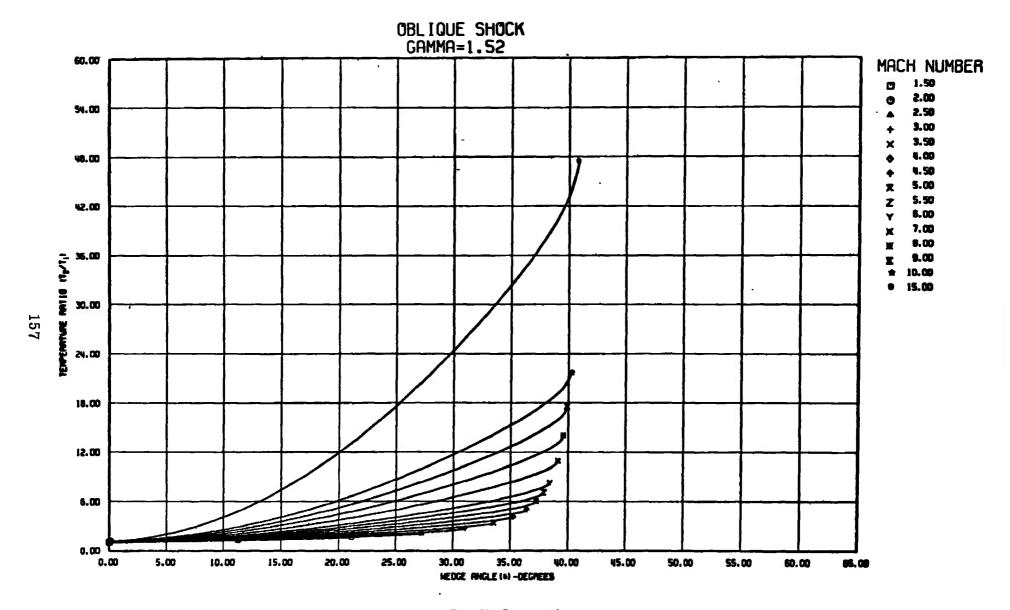


Fig. 23 Continued

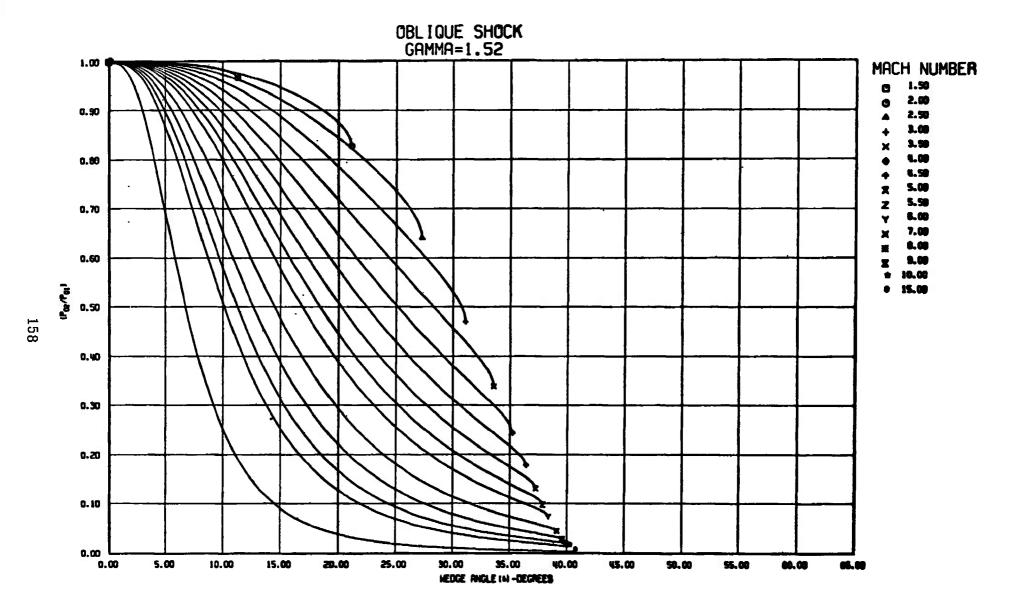


Fig. 23 Continued

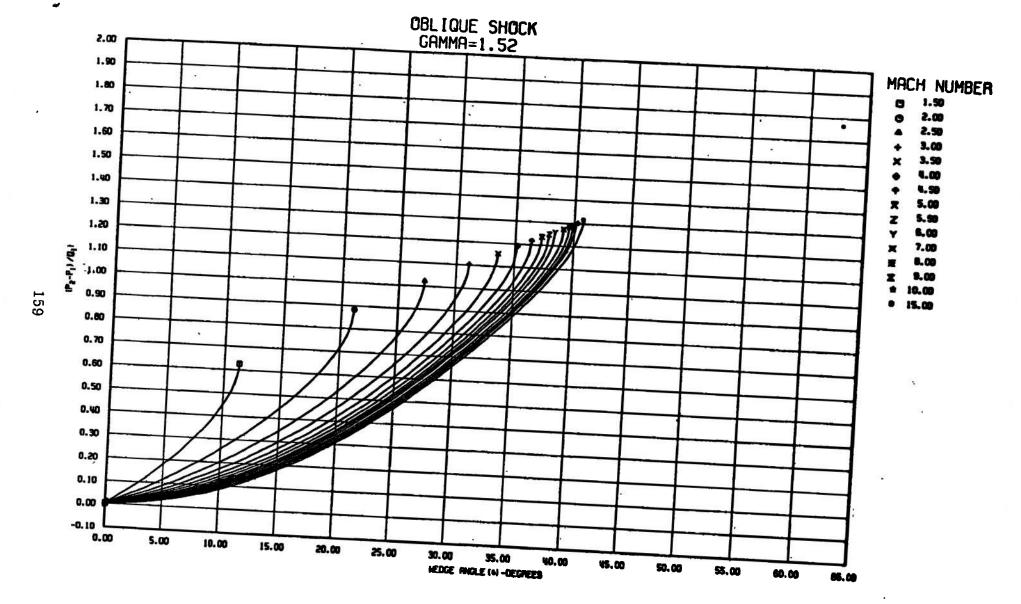


Fig. 23 Concluded

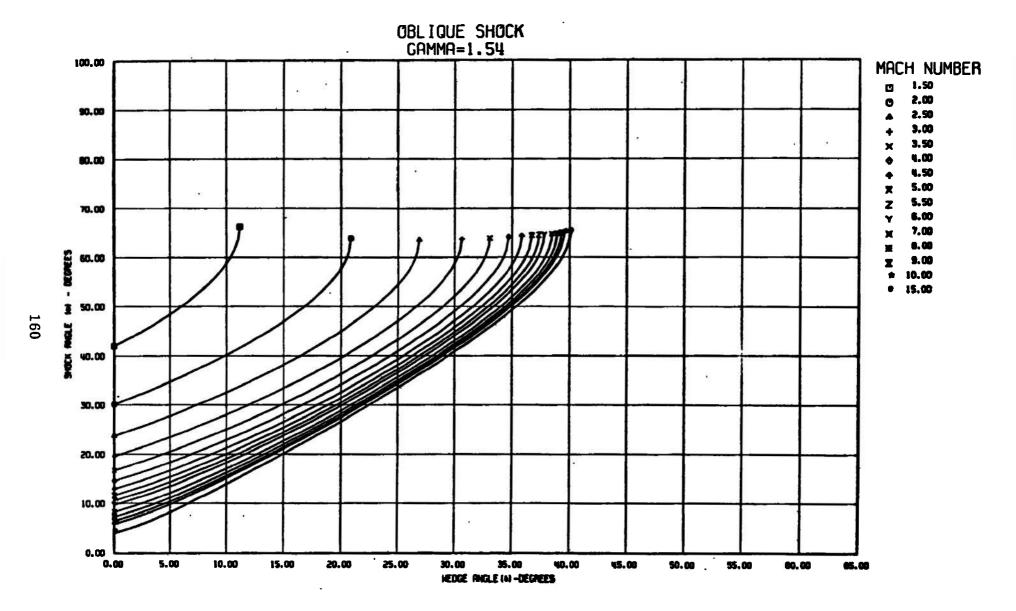


Fig. 24 y = 1.54

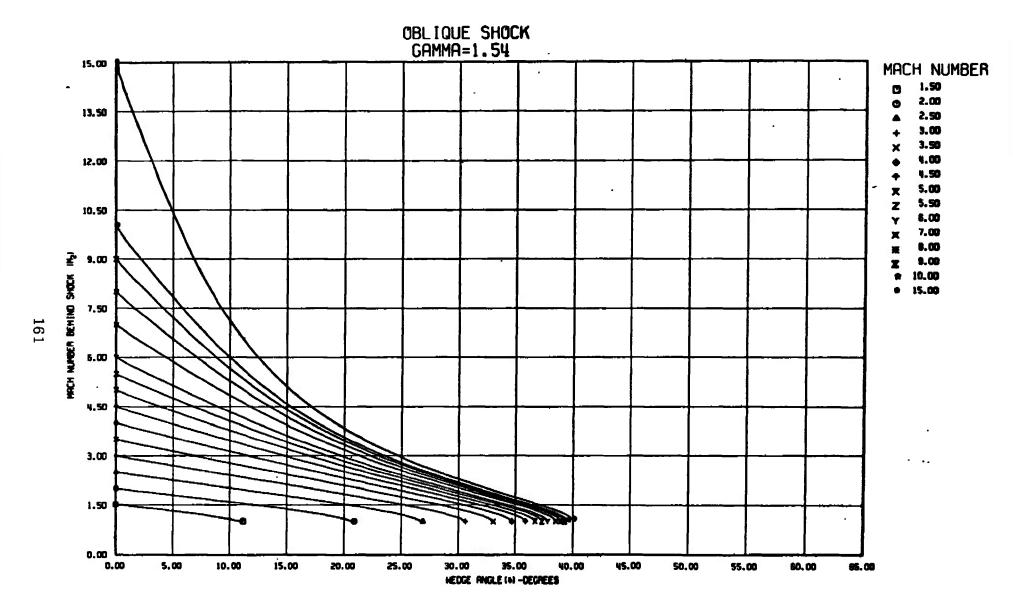


Fig. 24 Continued

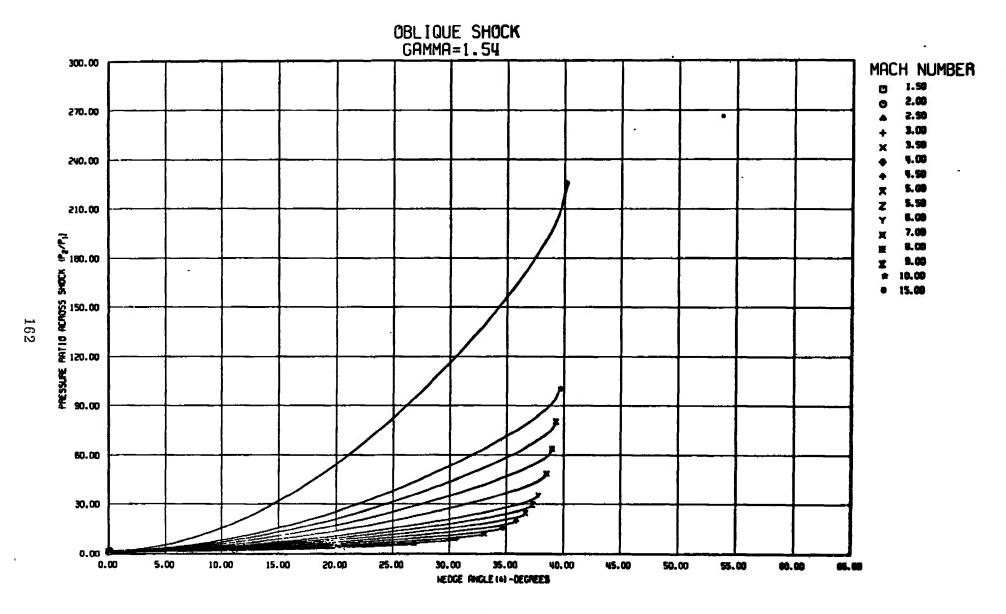


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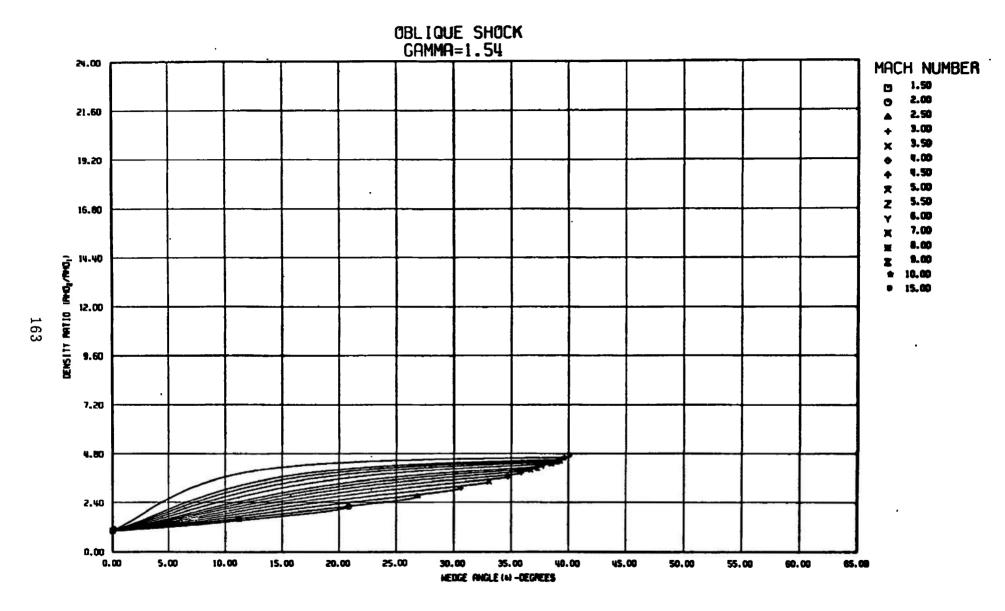


Fig. 24 Continued

Fig. 24 Continued

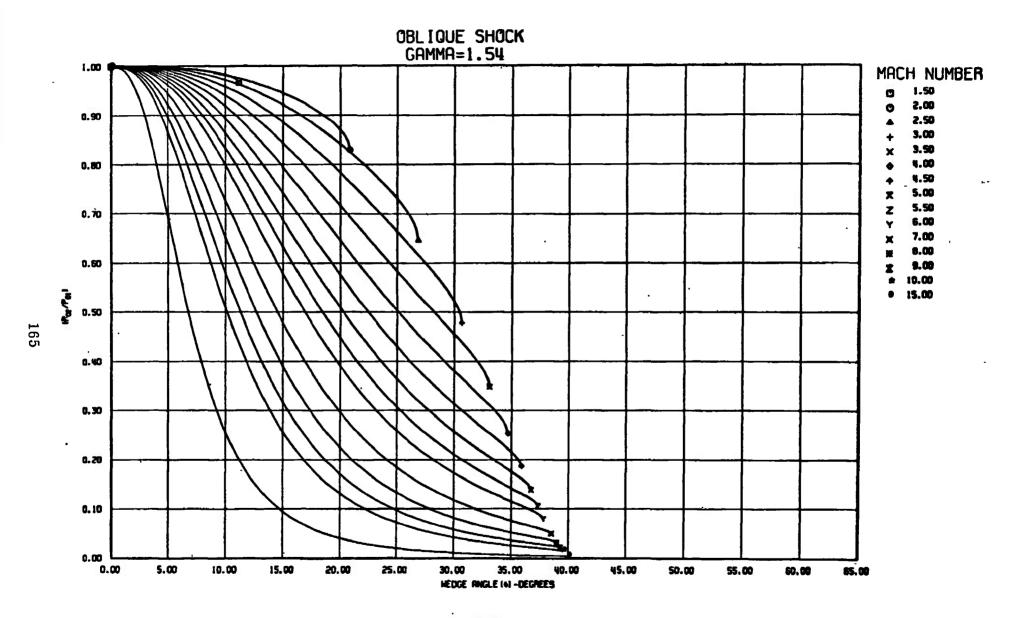


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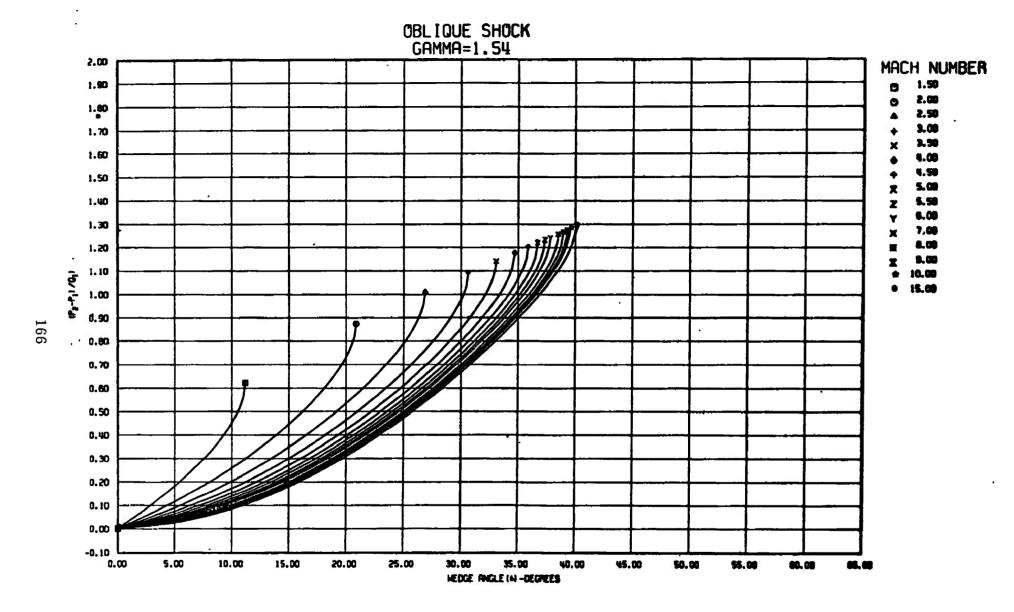


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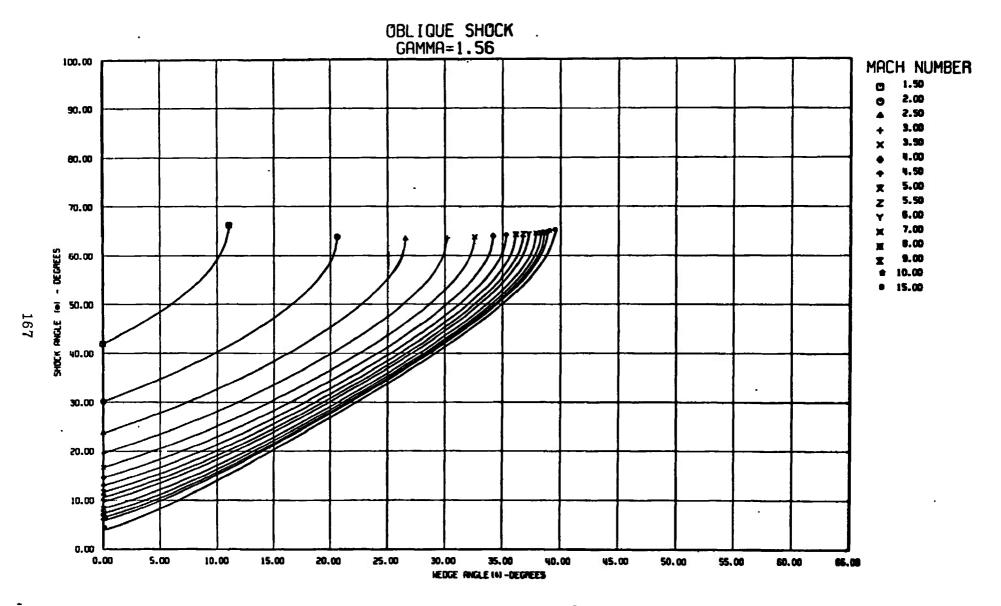


Fig. 25 y = 1.56

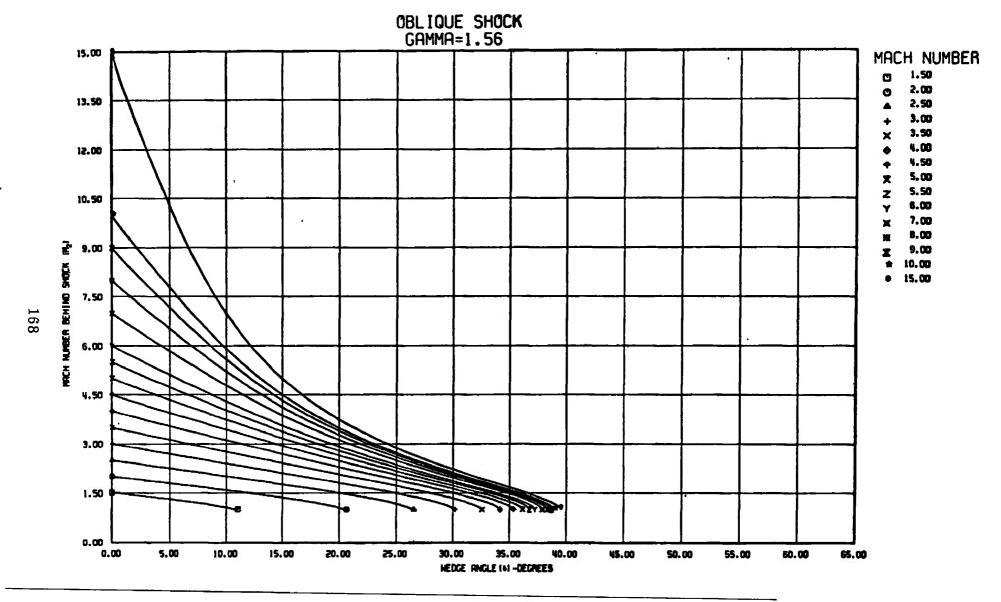


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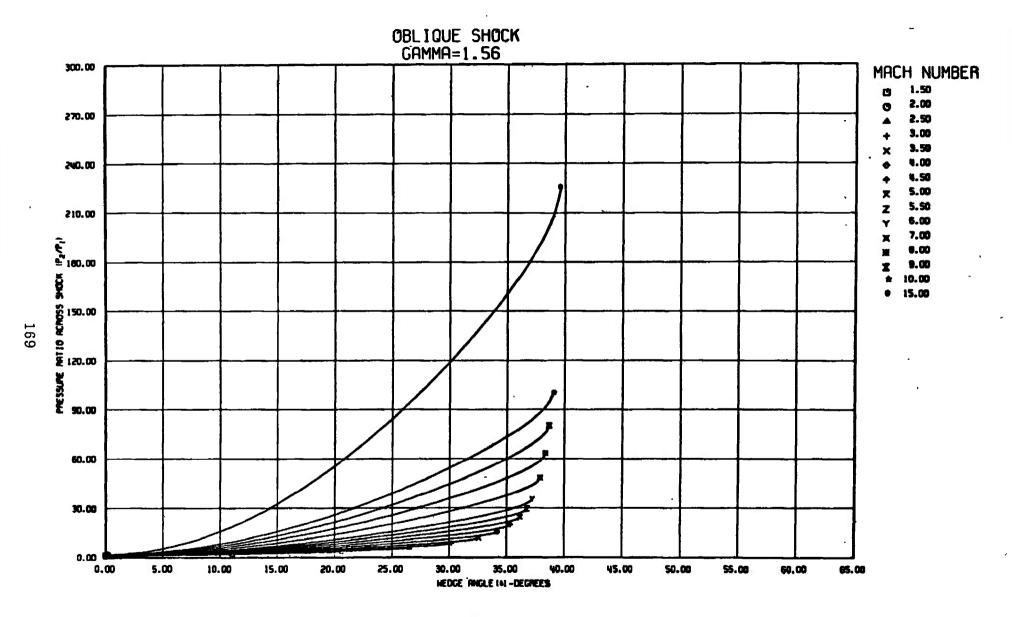


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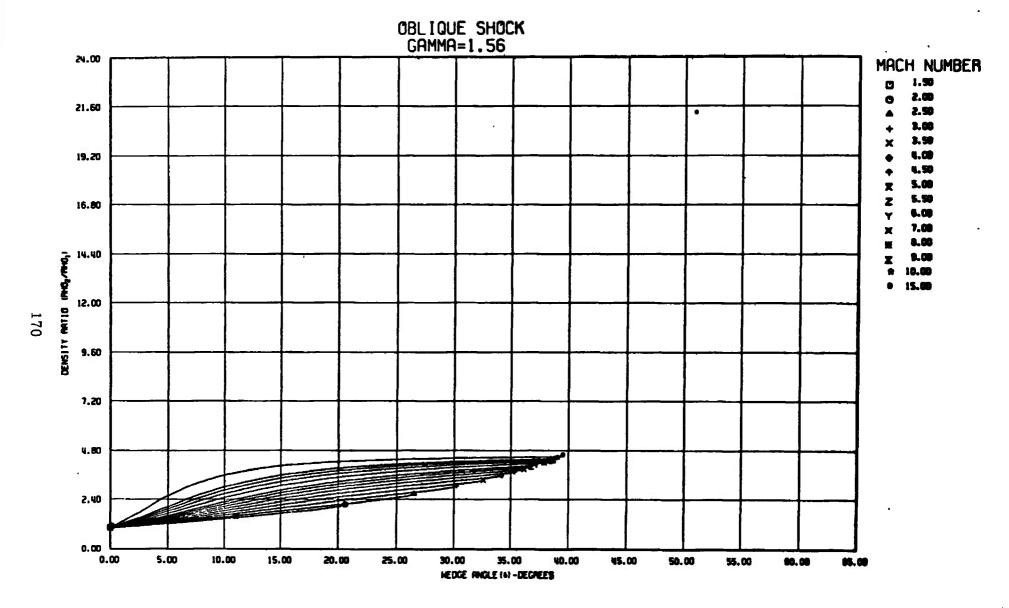


Fig. 25 Continued

Fig. 25 Continued

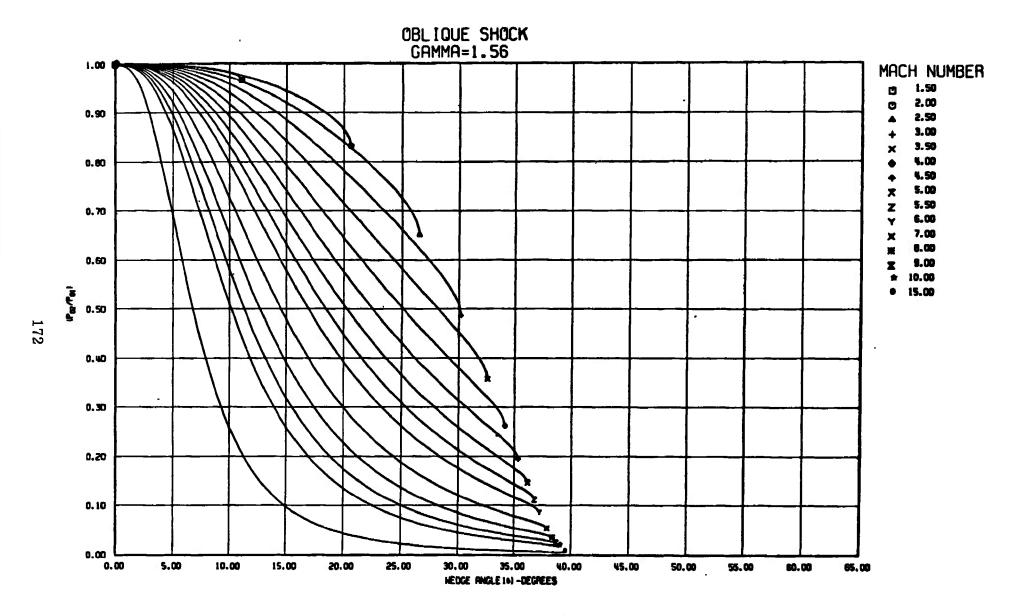


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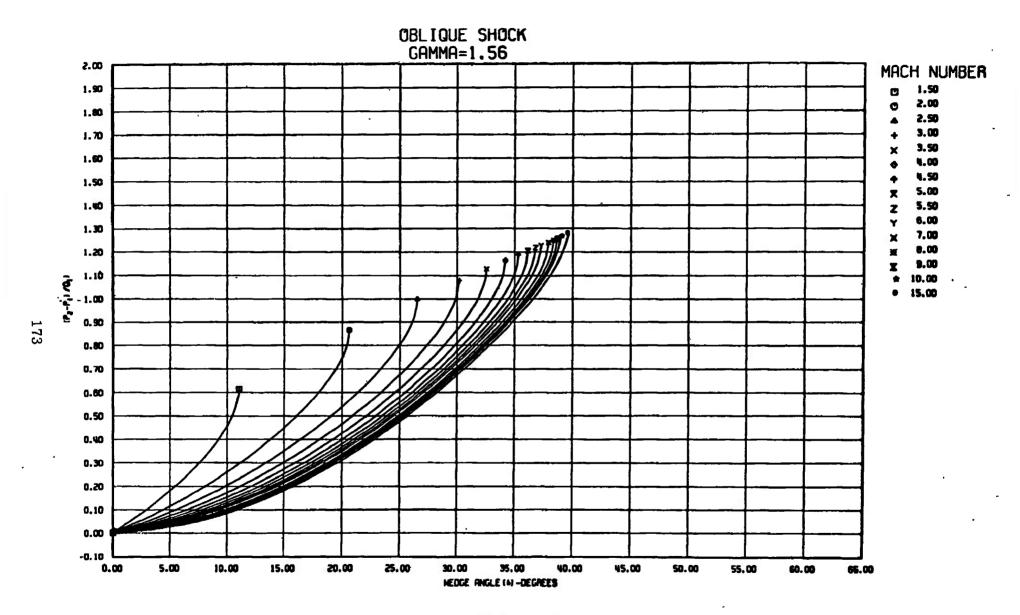


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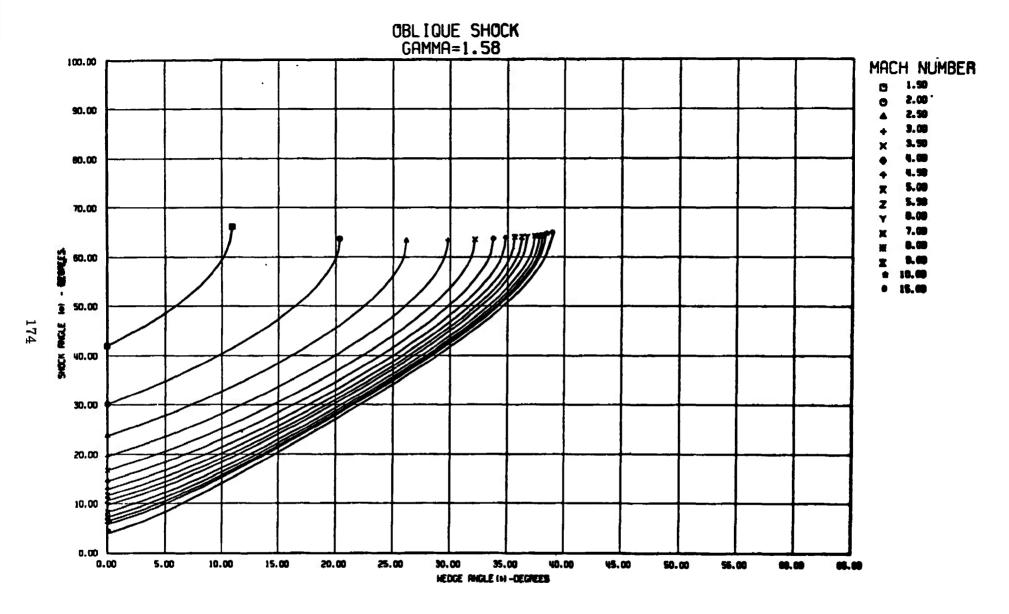


Fig. 26 γ - 1.58

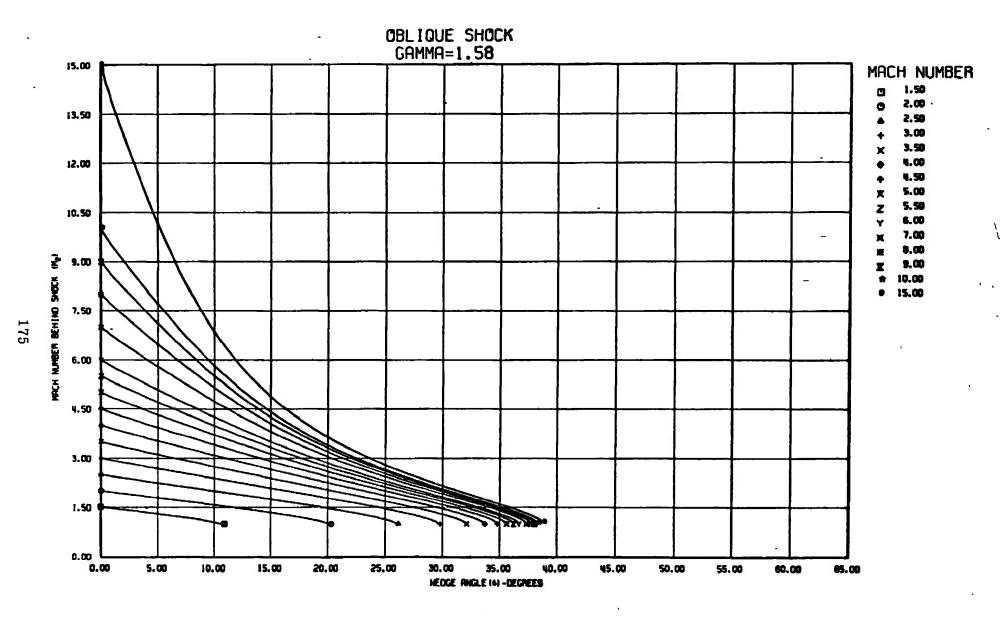


Fig. 26 Continued

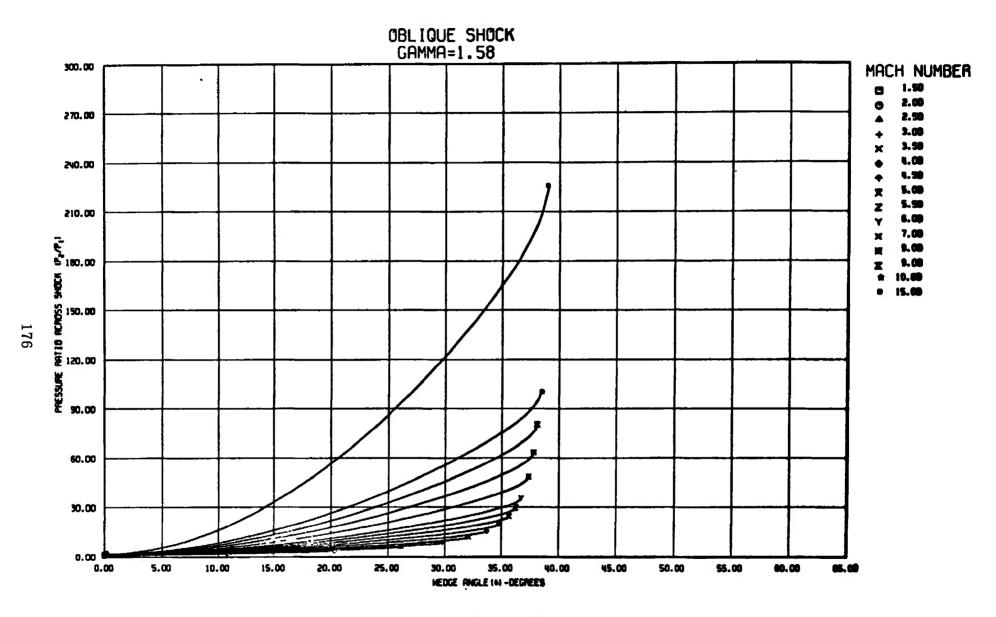


Fig. 26 Continued

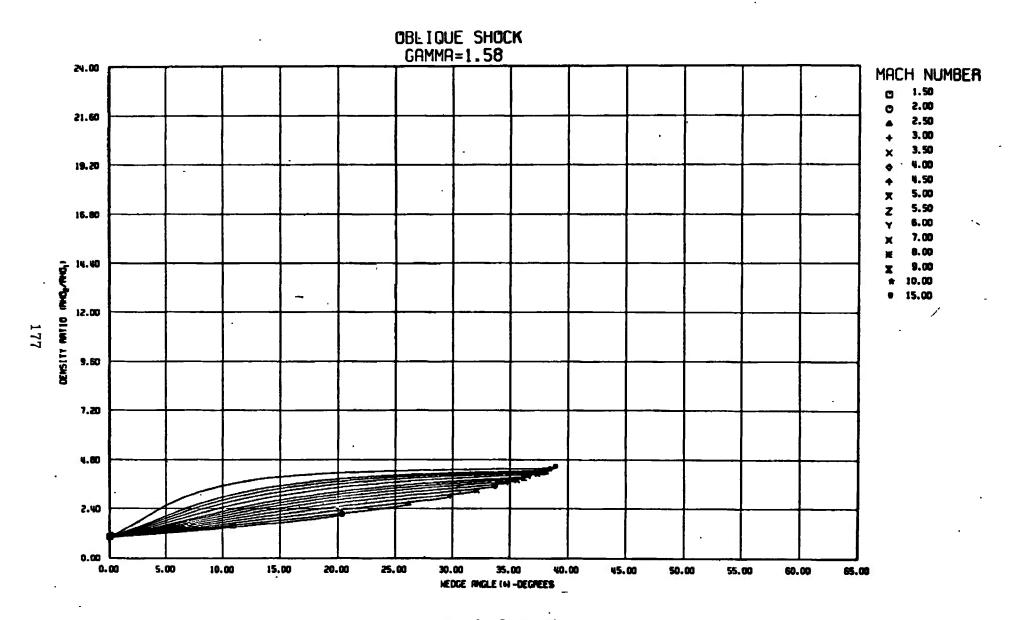


Fig. 26 Continued

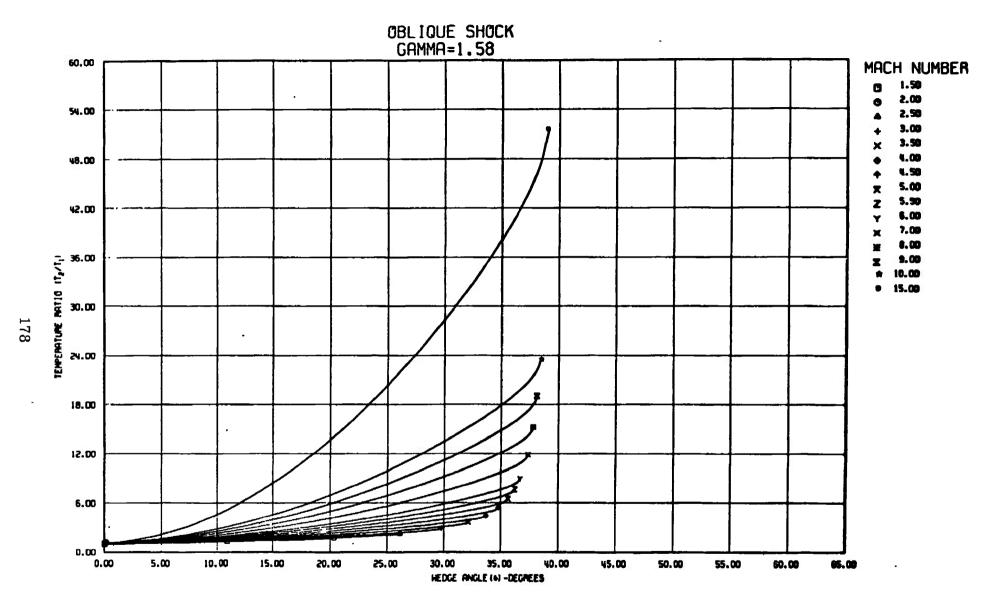


Fig. 26 Continued

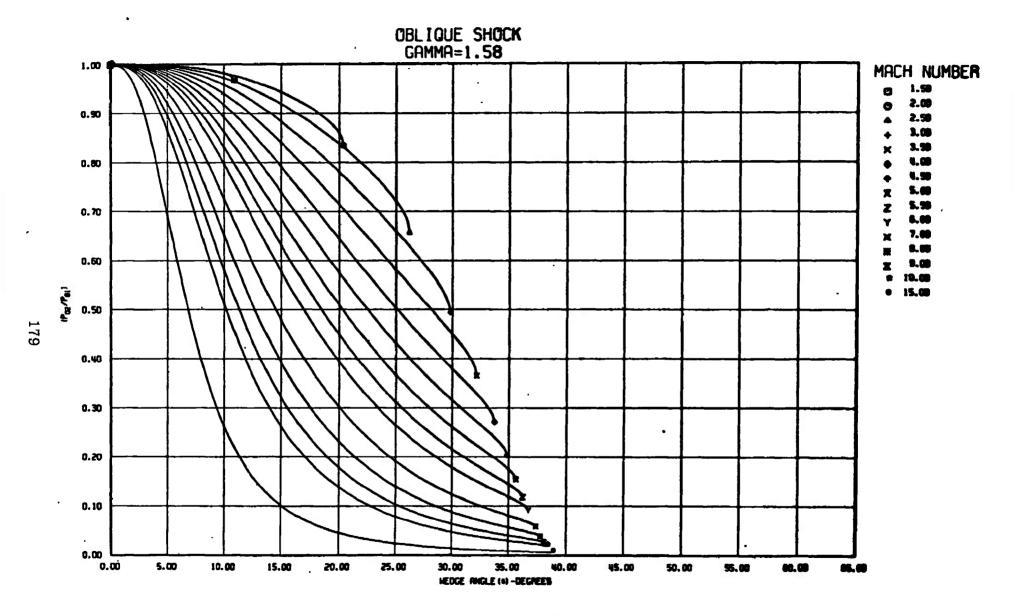


Fig. 26 Continued

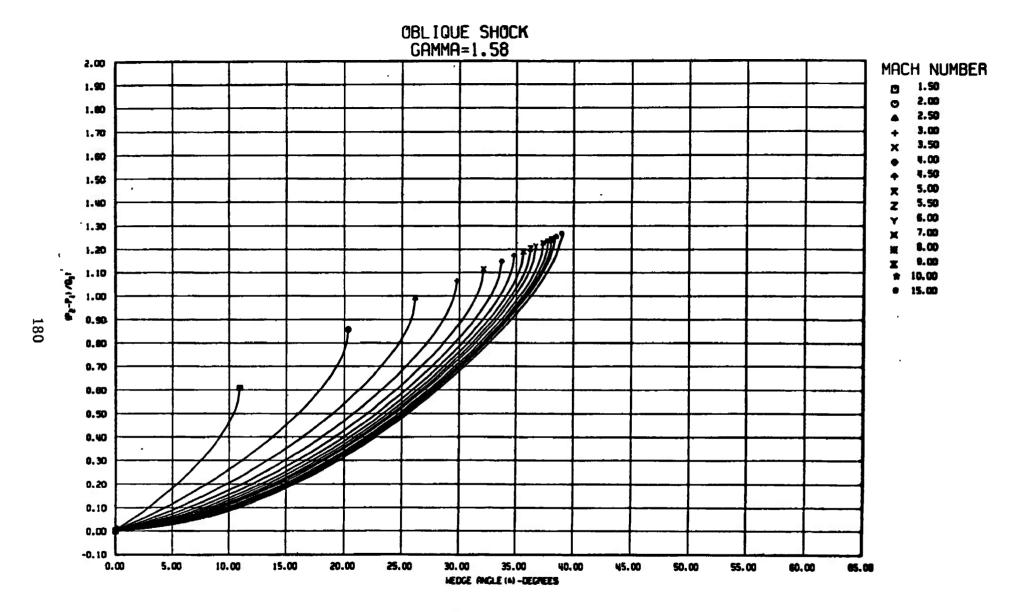


Fig. 26 Concluded

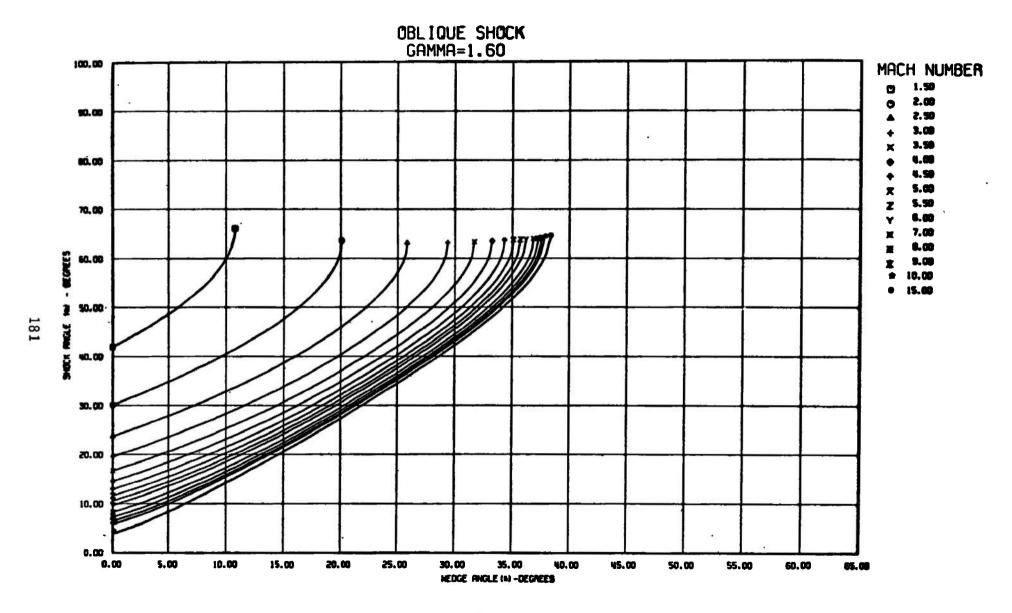


Fig. 27 y = 1.60

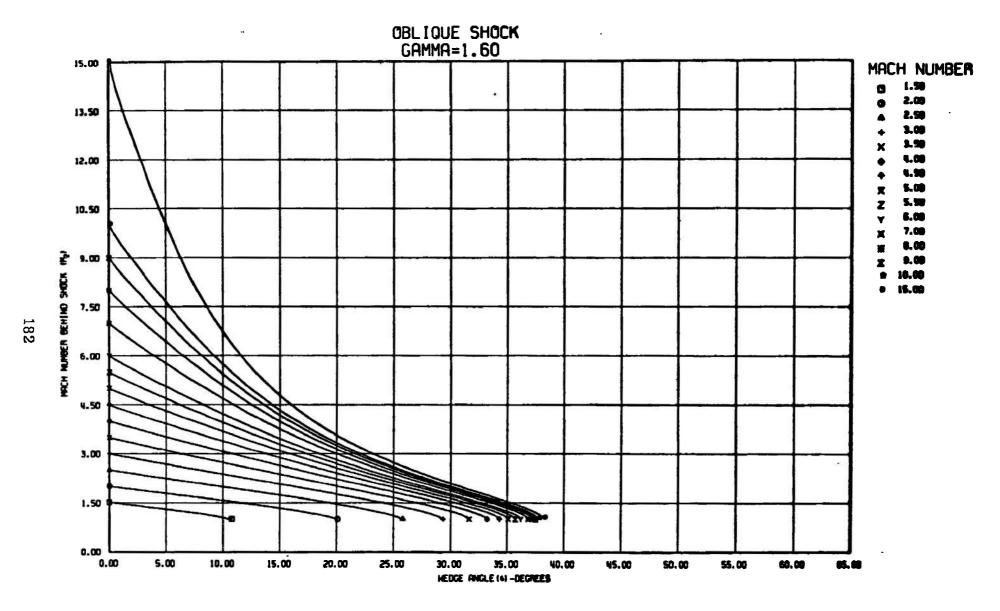


Fig. 27 Continued

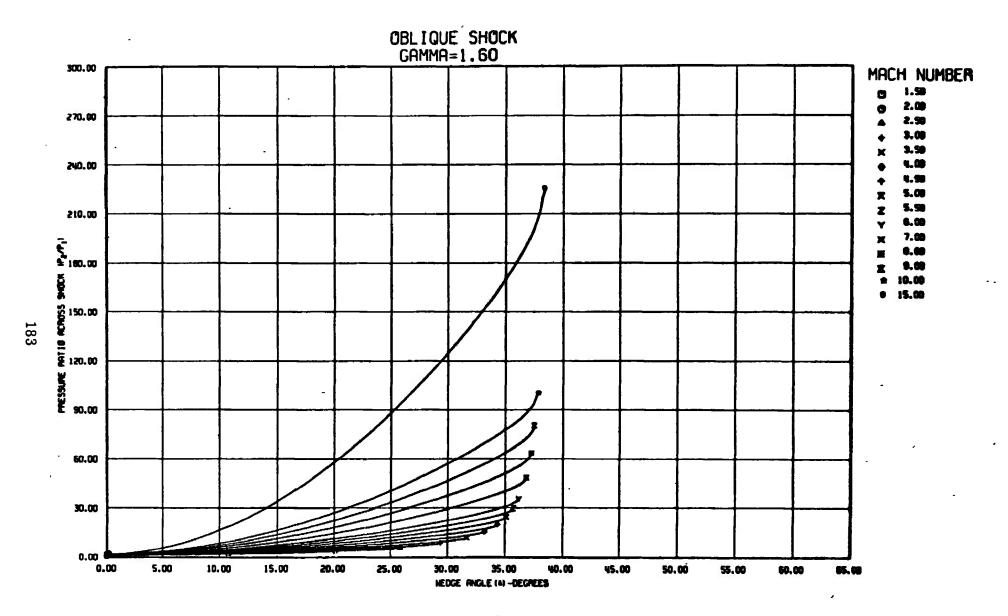


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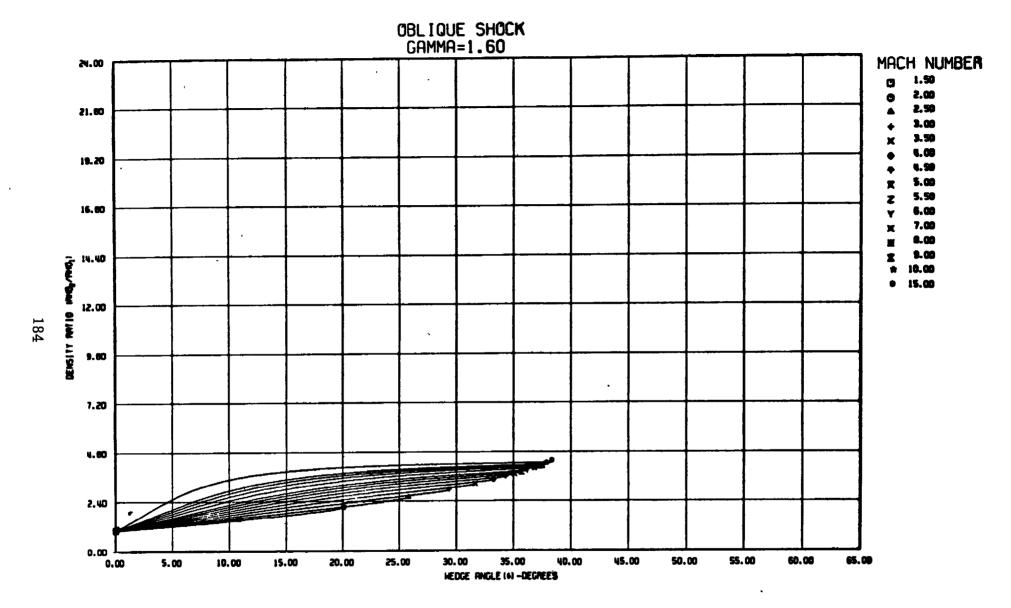


Fig. 27 Continued

Fig. 27 Continued

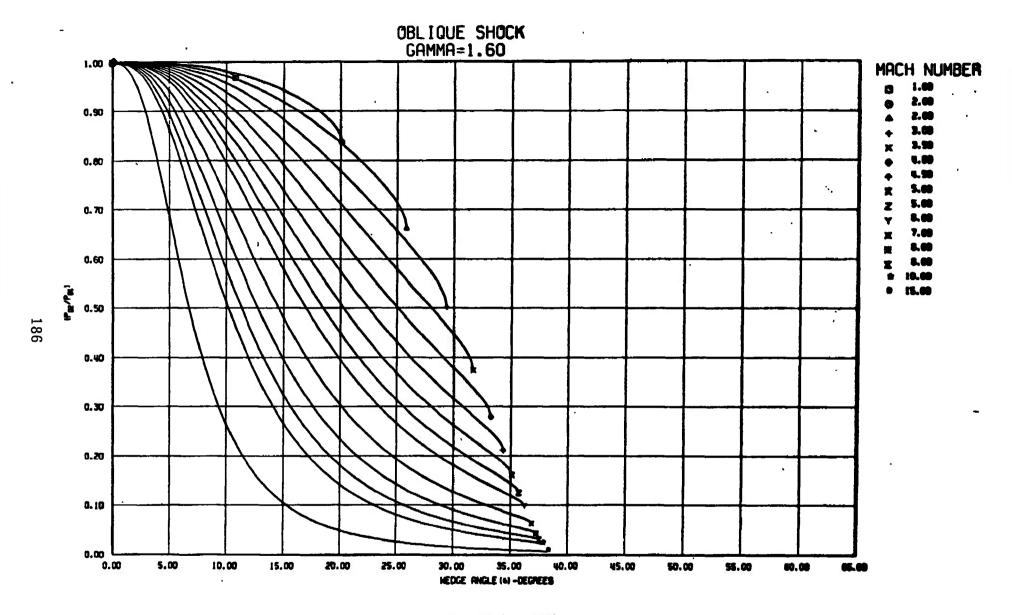


Fig. 27 Continued

Fig. 27 Concluded

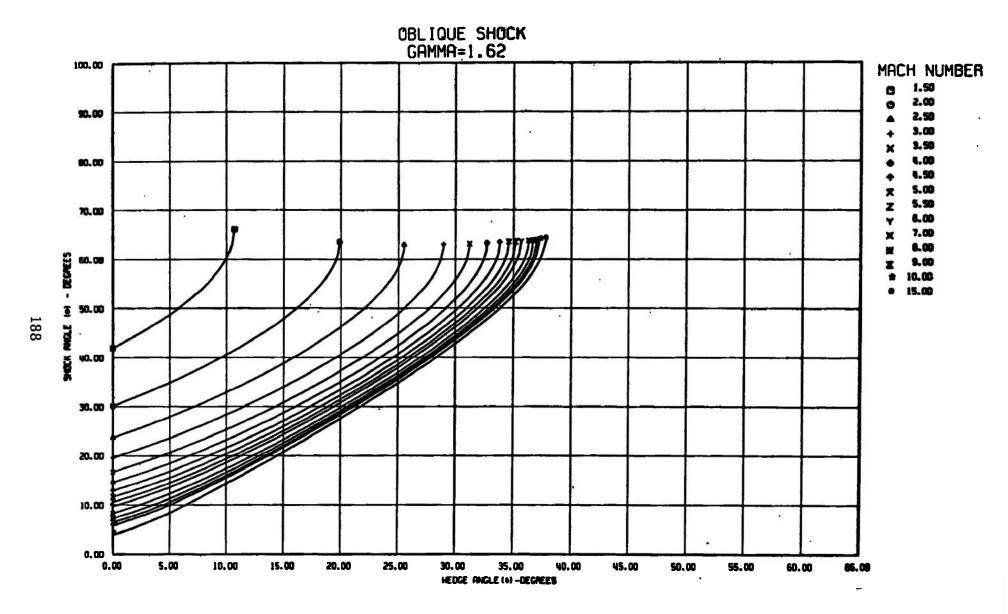


Fig. 28 y = 1.62

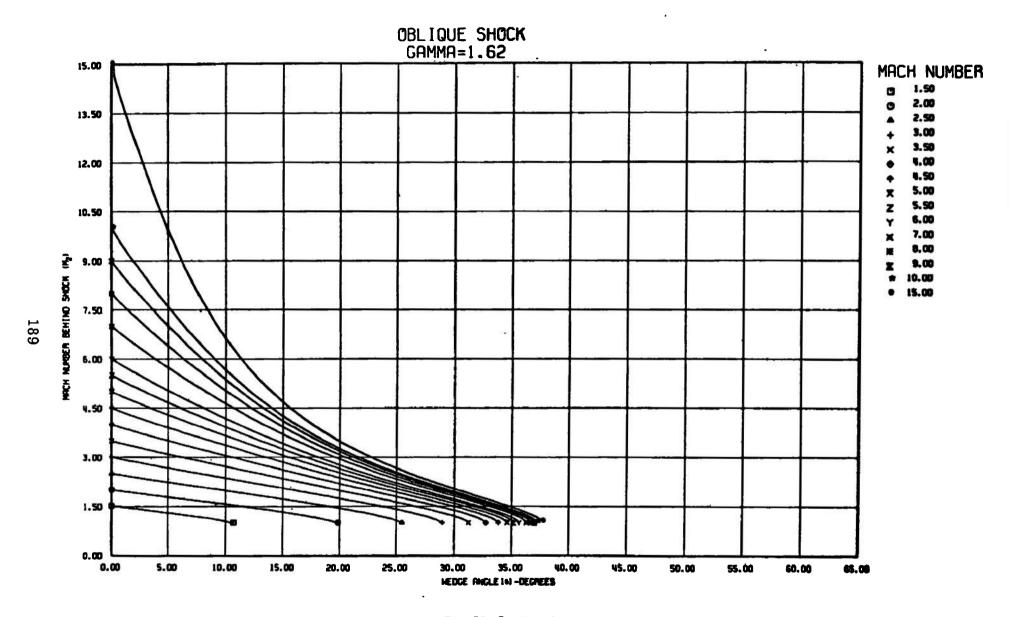


Fig. 28 Continued

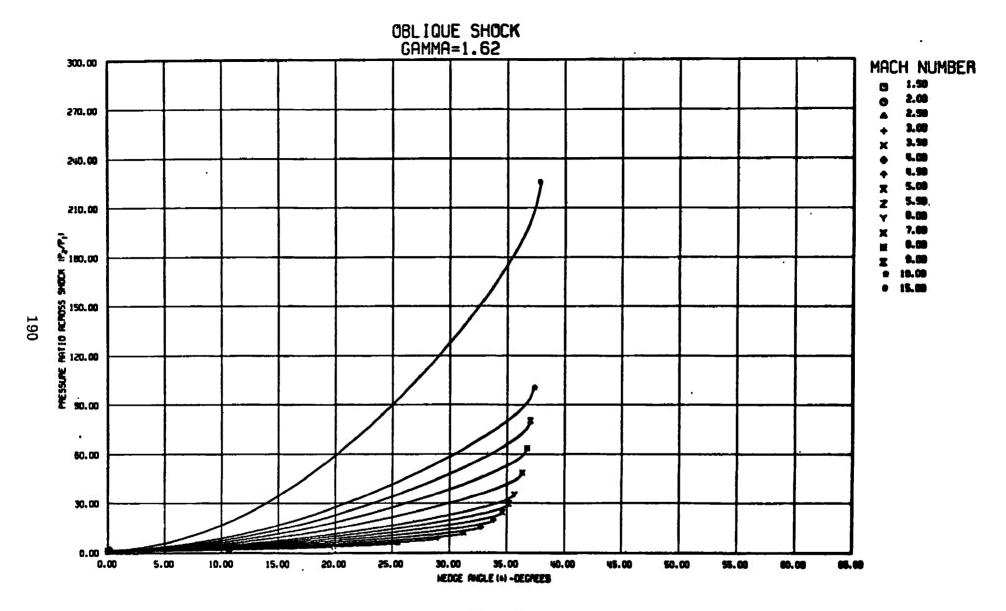


Fig. 28 Continued

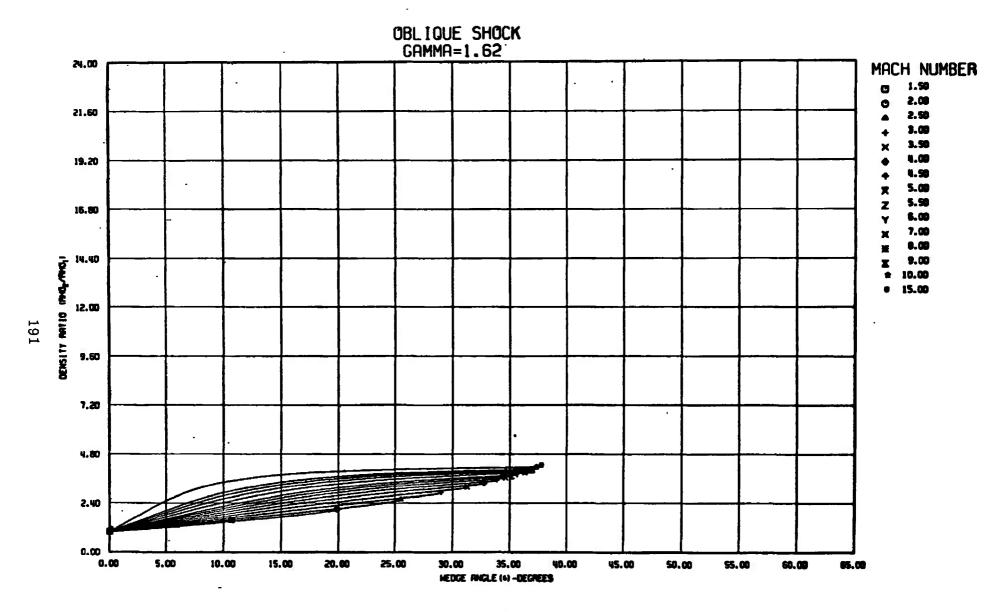


Fig. 28 Continued

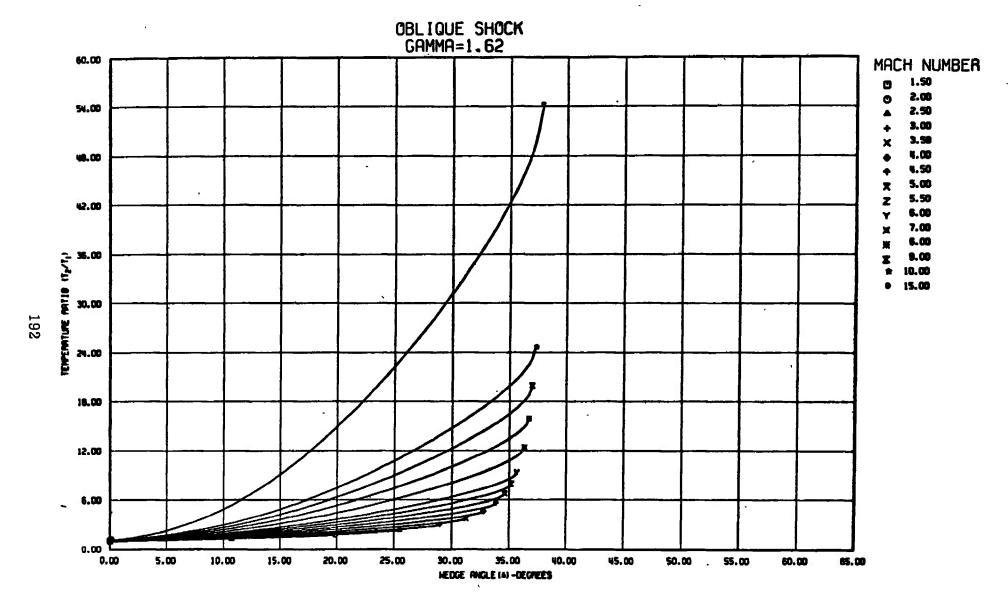


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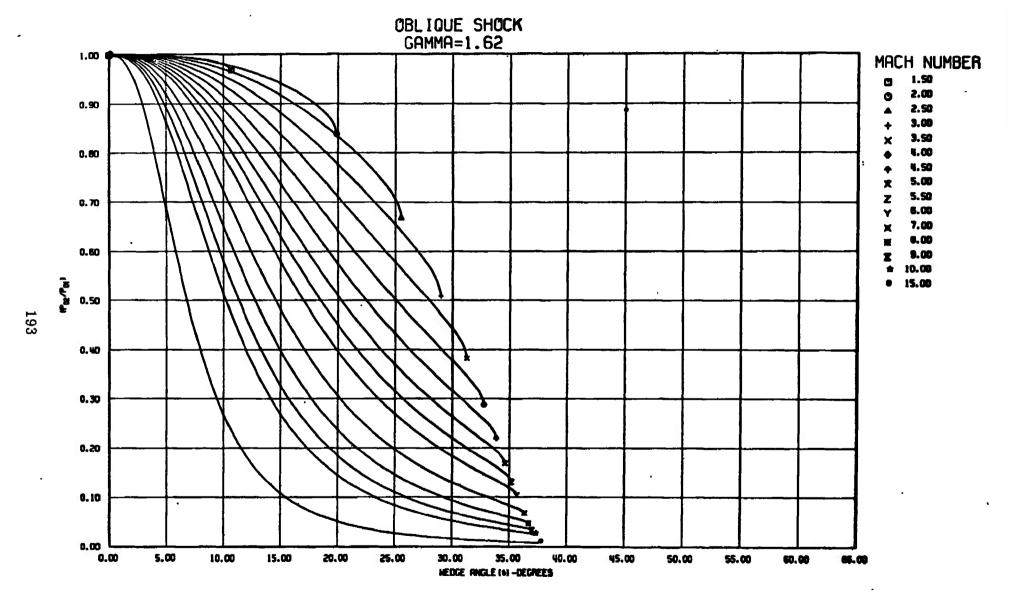


Fig. 28 Continued

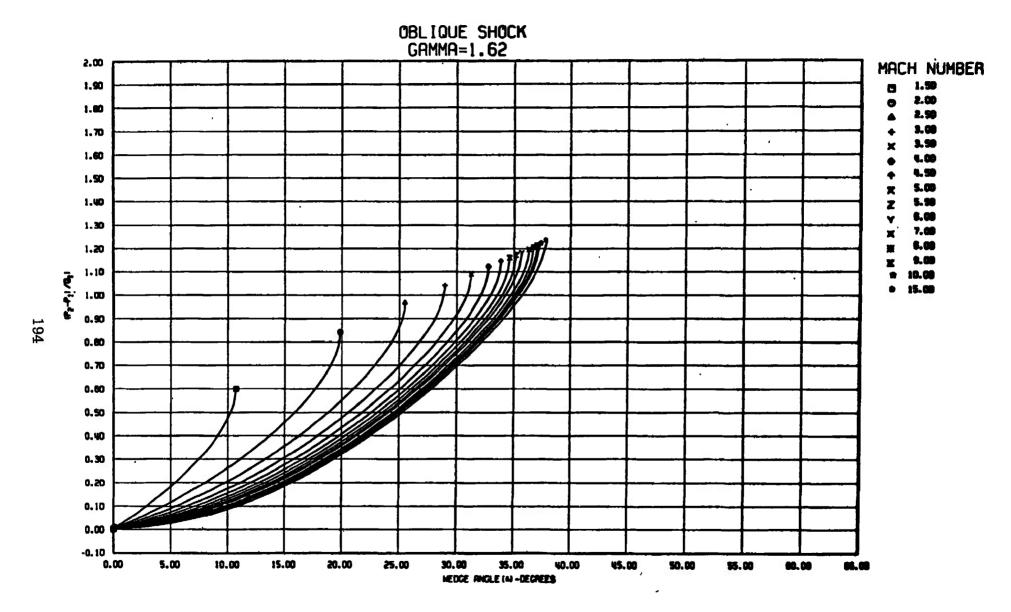


Fig. 28 Concluded

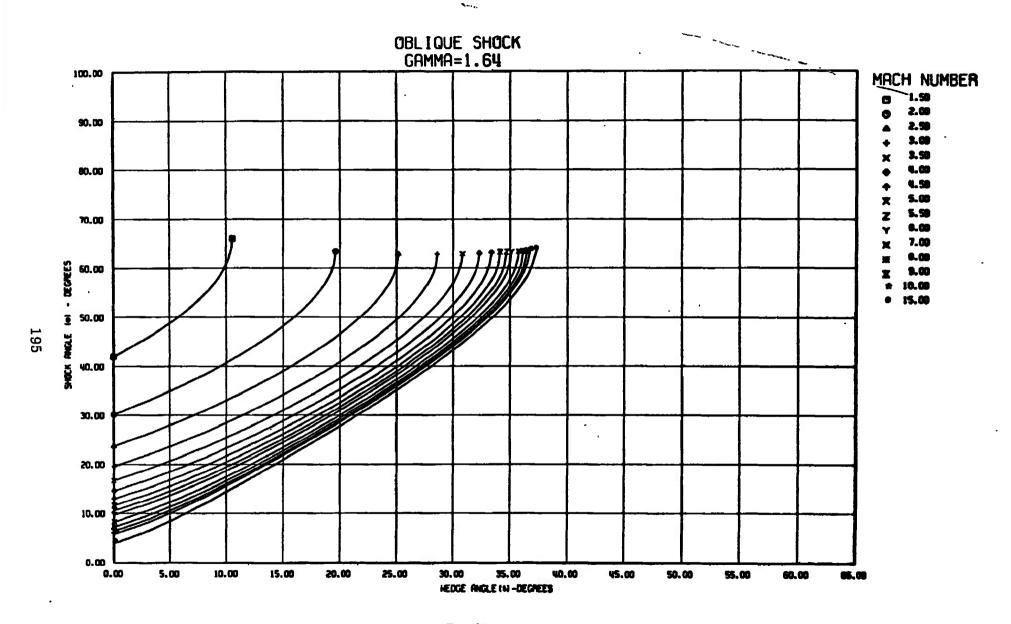


Fig. 29 y = 1.64

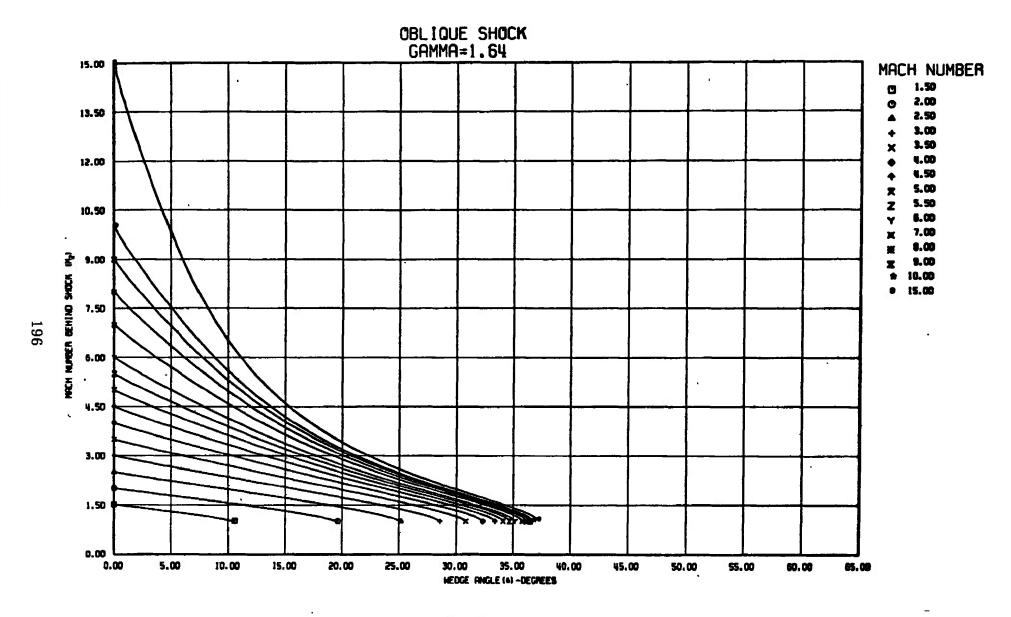


Fig. 29 Continued

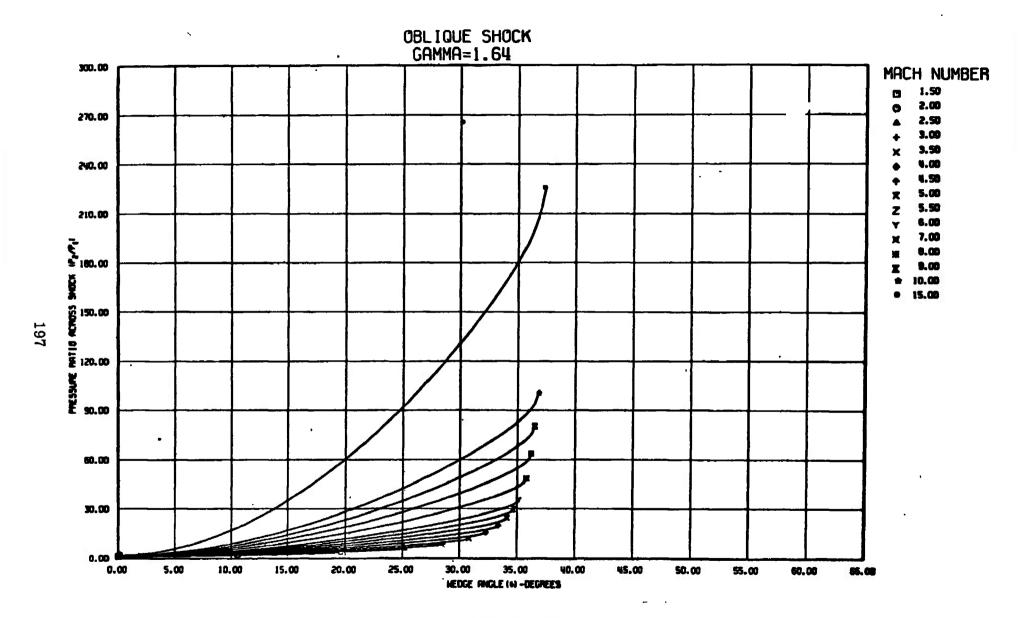


Fig. 29 Continued

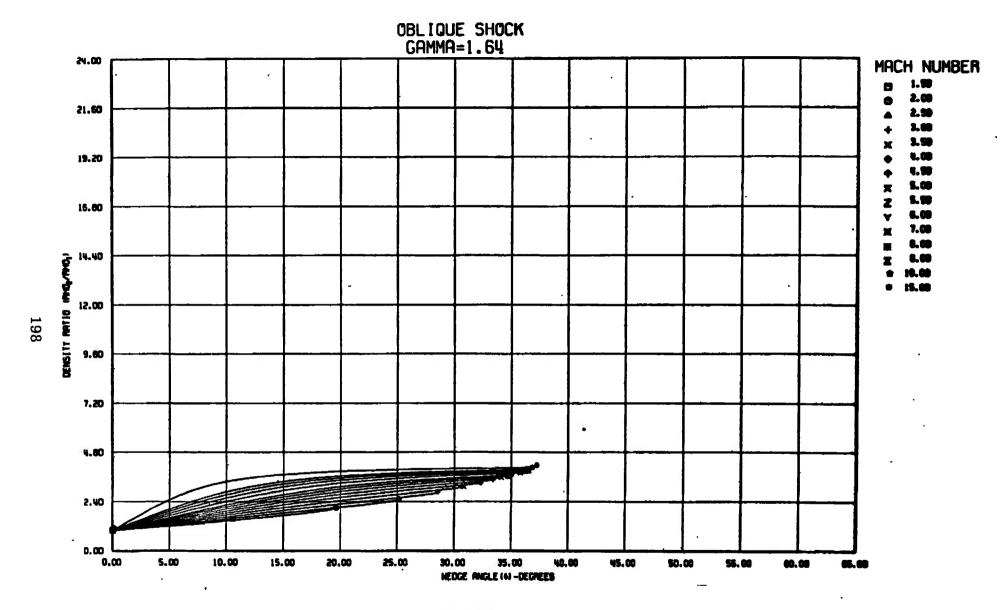


Fig. 29 Continued

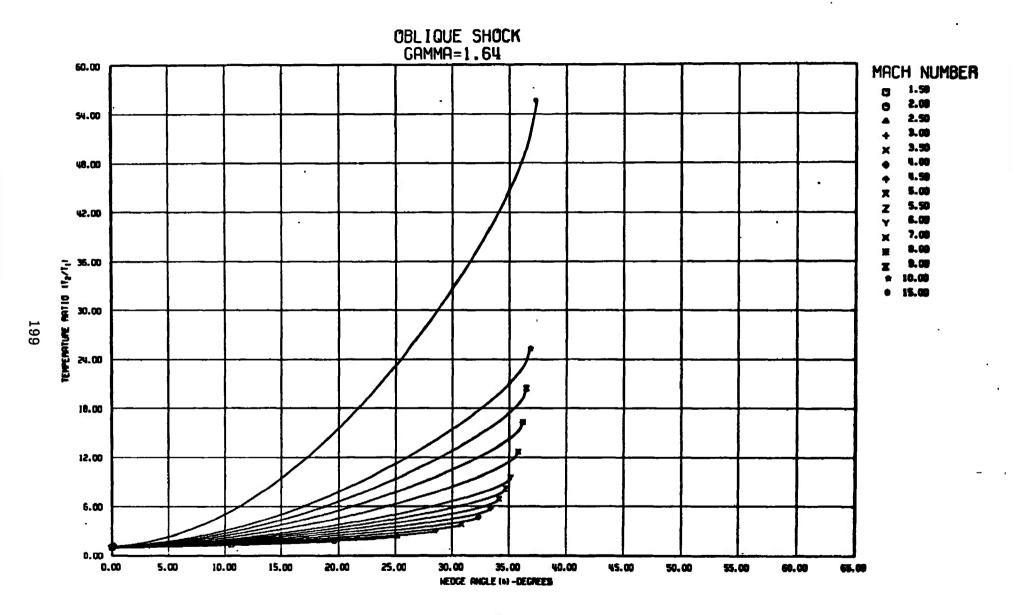


Fig. 29 Continued

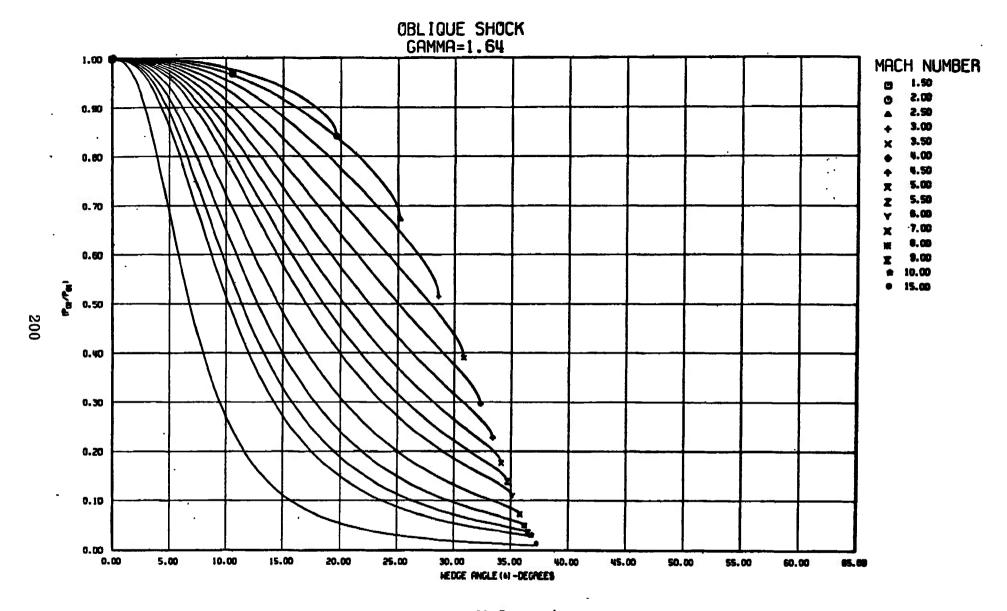


Fig. 29 Continued

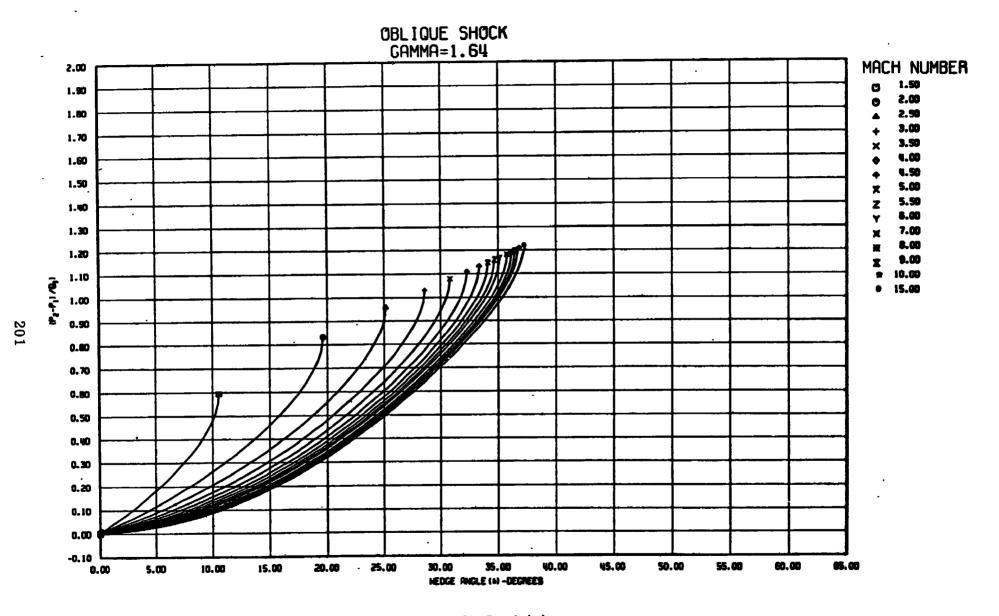


Fig. 29 Concluded

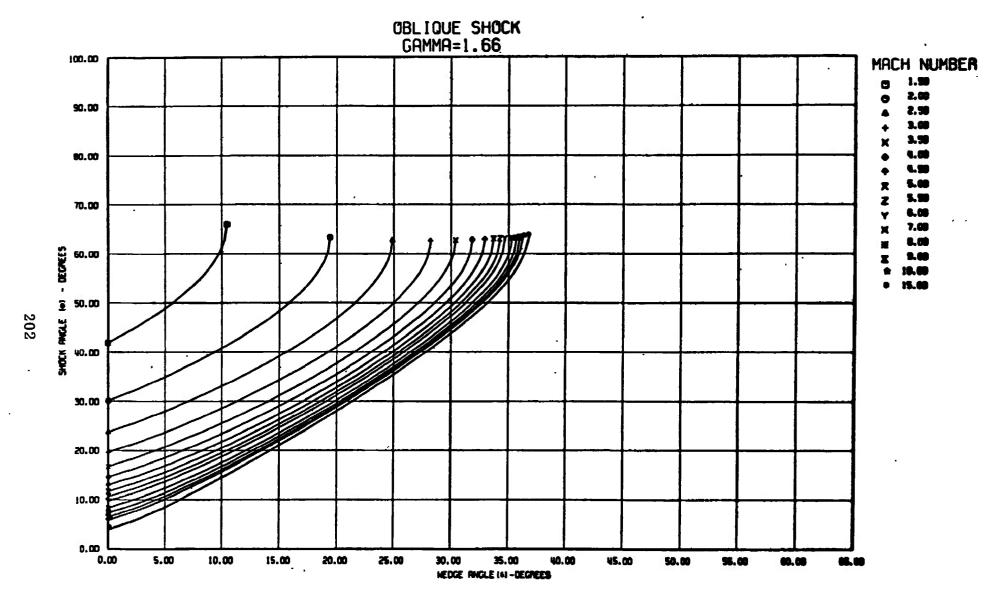


Fig. 30 y = 1.66

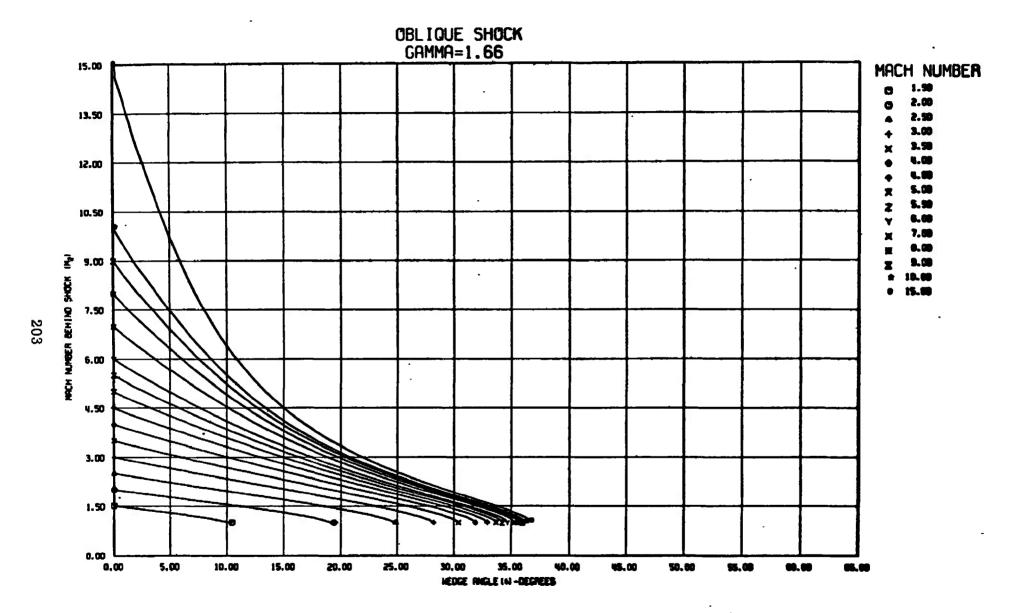


Fig. 30 Continued

Fig. 30 Continued

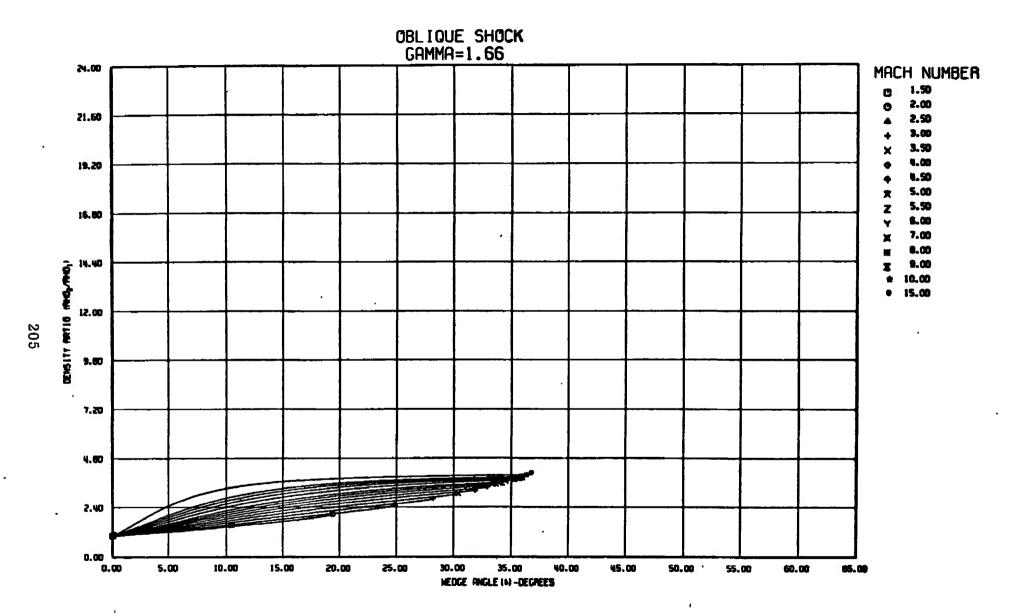


Fig. 30 Continued

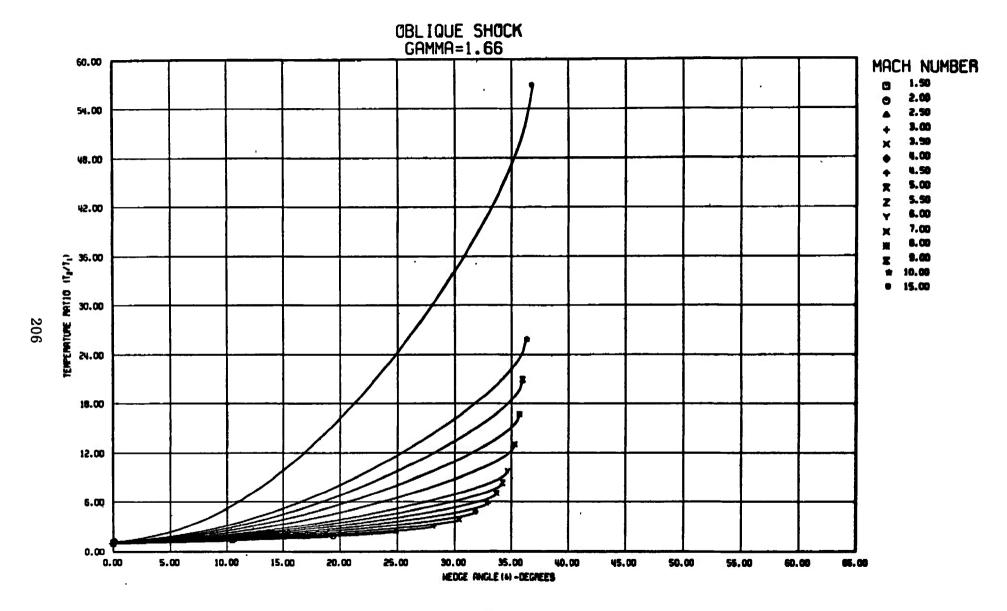


Fig. 30 Continued

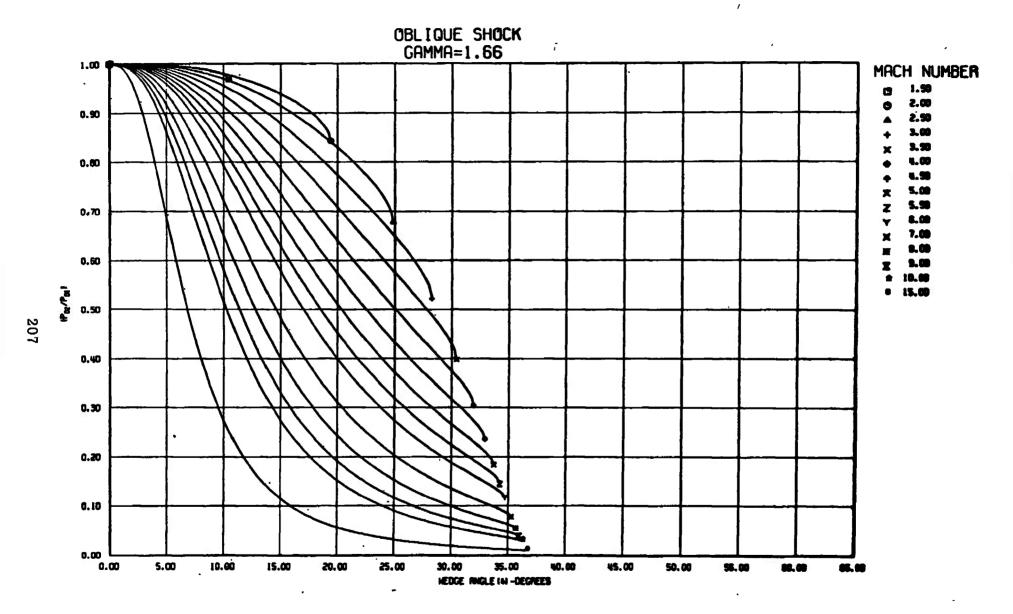


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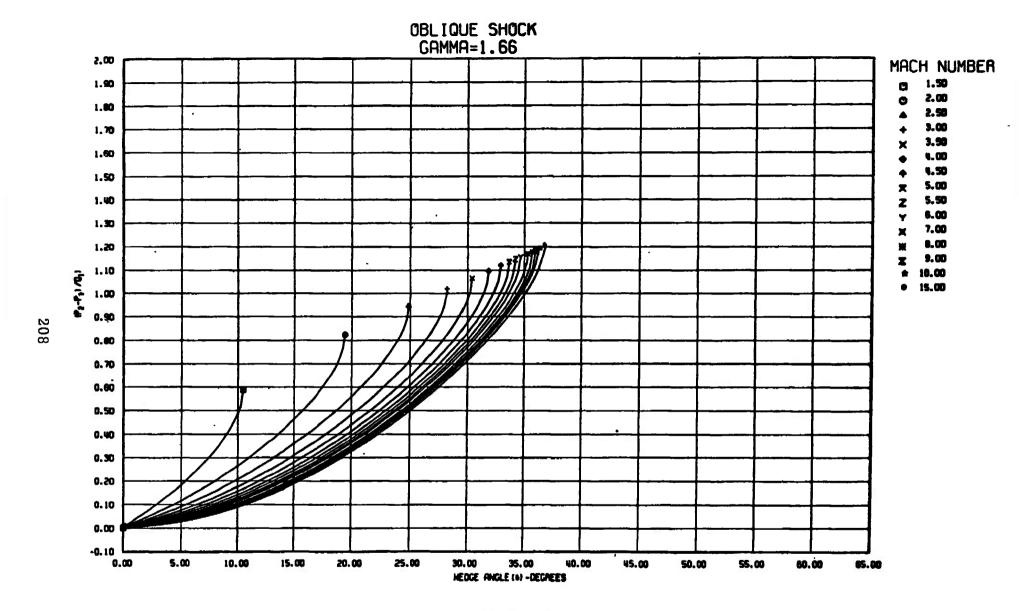


Fig. 30 Concluded

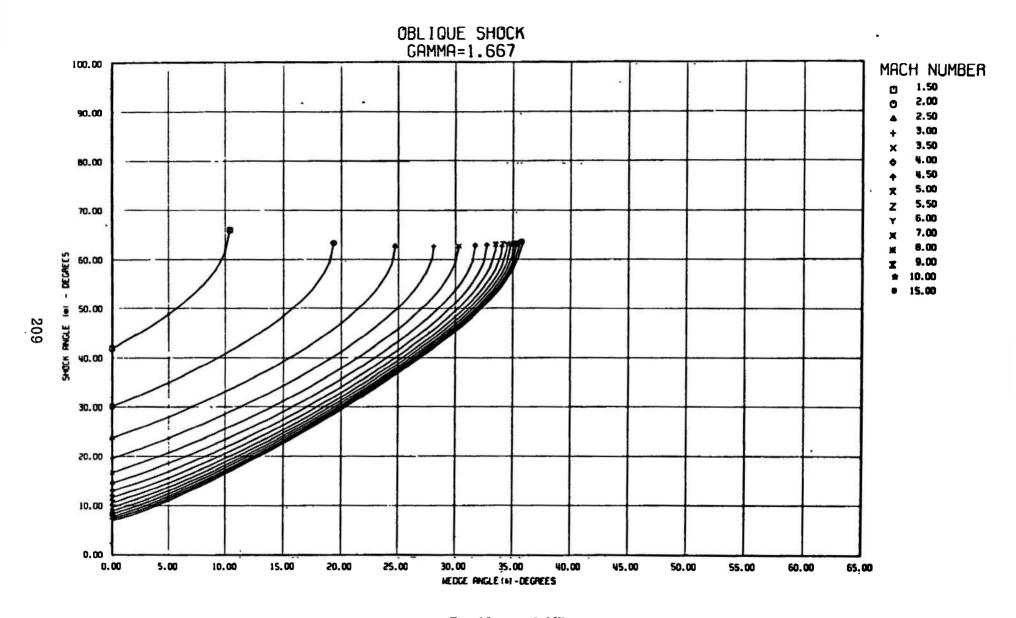


Fig. 31 y = 1.667

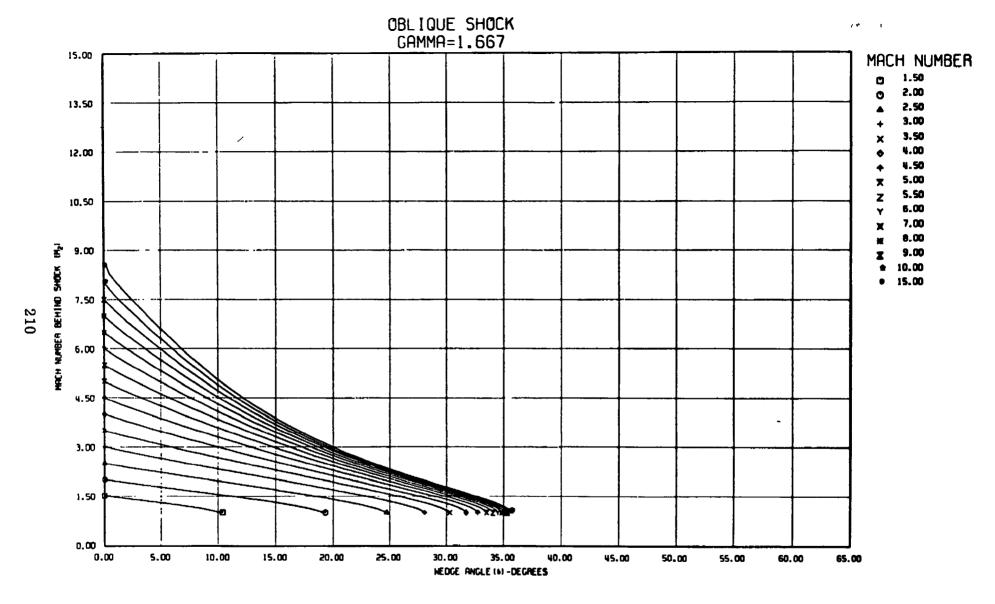


Fig. 31 Continued

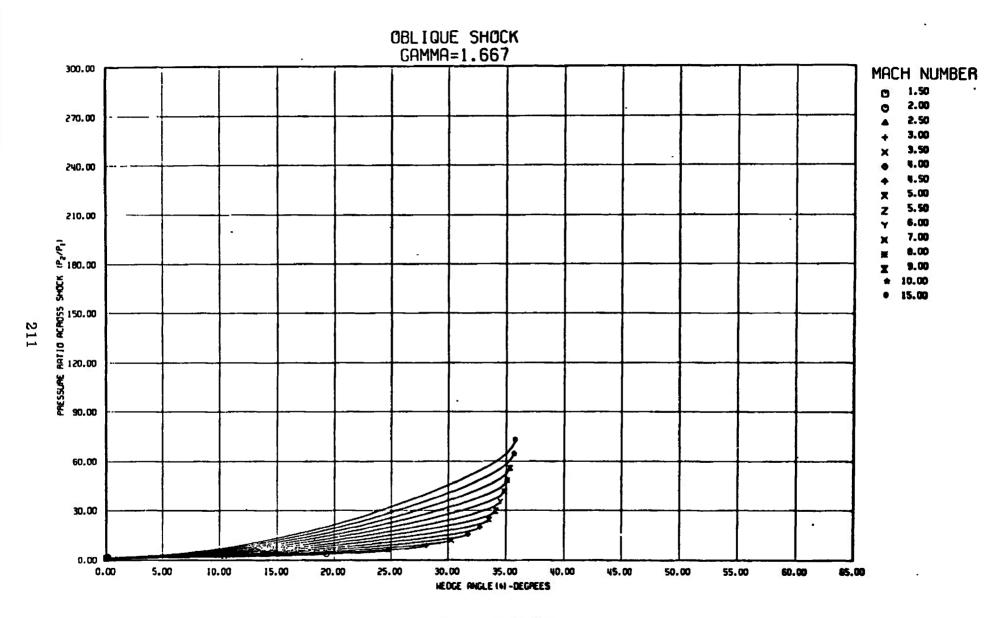


Fig. 31 Continued

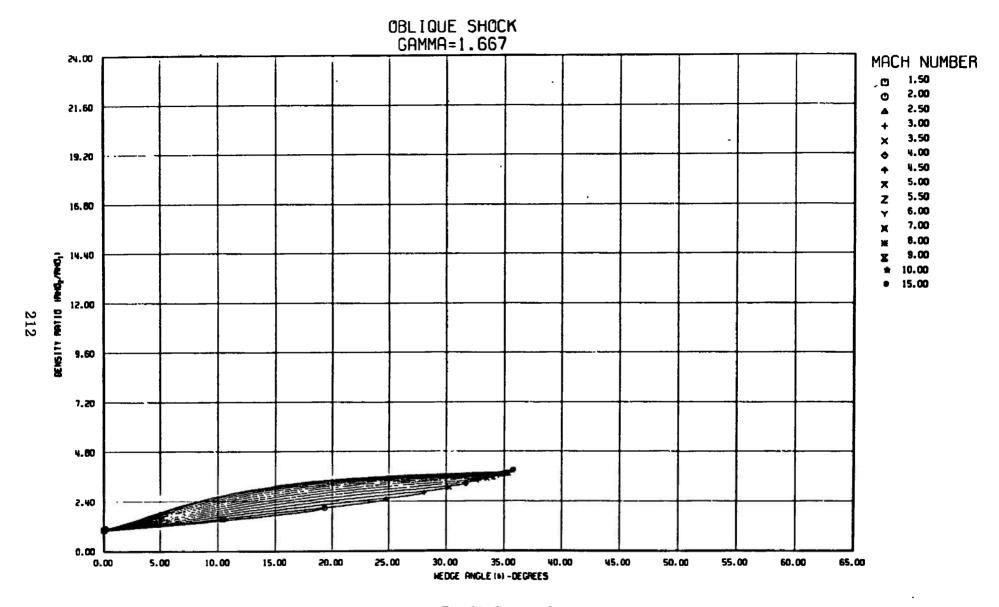


Fig. 31 Continued

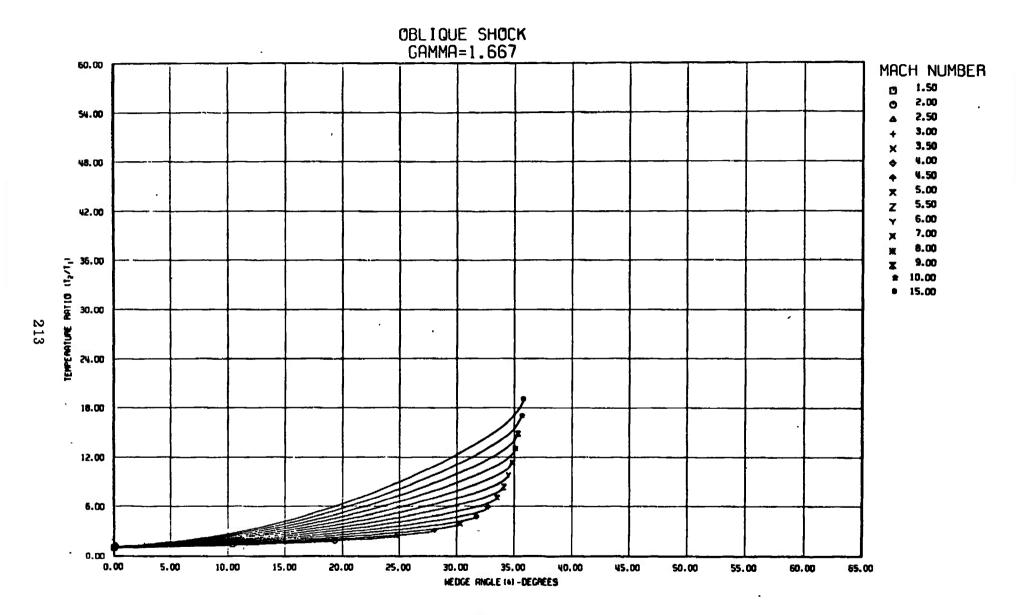


Fig. 31 Continued

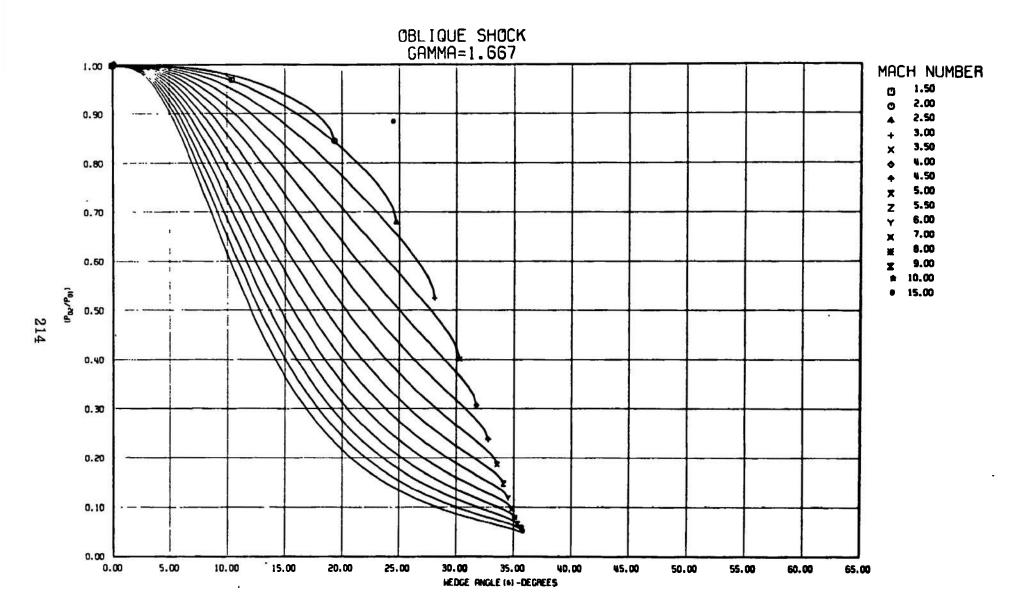


Fig. 31 Continued

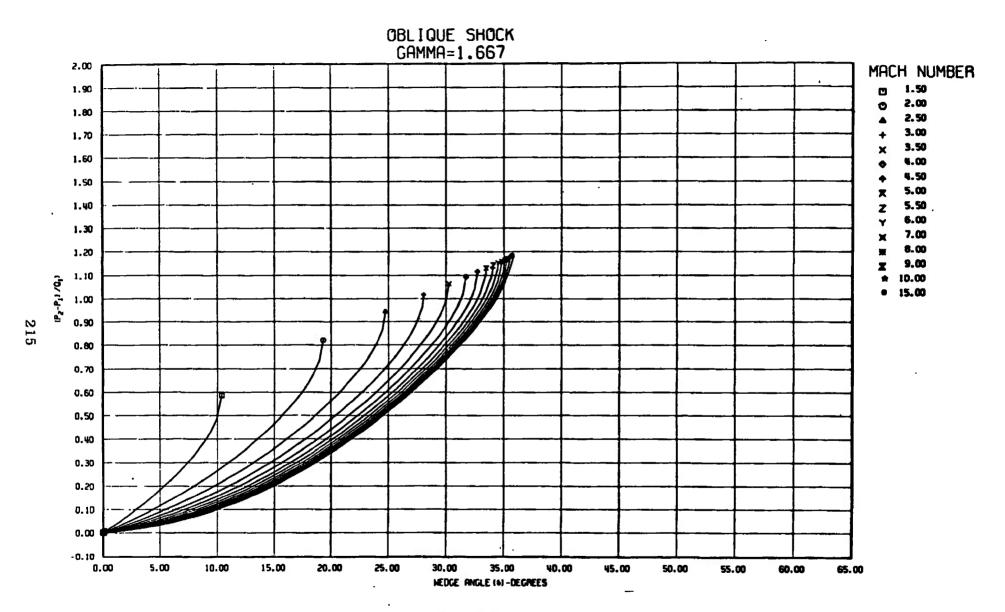


Fig. 31 Concluded

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This report is a presentation of graphs of the oblique-shock relationships for several ratios of specific heats at several Mach numbers. The specific heat ratios vary from 1.10 to 1.66 in increments of 0.02, and a plot is also shown for a specific heat ratio equal to 1.667. The Mach numbers range from 1.5 to 6.0 with increments of 0.5 and from 6.0 to 10.0 with increments of 1.0; Mach number 15.0 is presented separately.

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14. KEY WORDS	LINK A		LINKB		LINK C	
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specific heat						
supersonic flow						
hypersonic flow						
wind tunnels						
shock waves	:					
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